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Numerical prediction of the aerodynamics and acoustics of a tip leakage flow using Large-Eddy Simulation

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Abstract: A Large-Eddy Simulation of the tip leakage flow of a single airfoil is carried out. The configuration consists of a non-rotating, isolated airfoil between two horizontal plates with a gap of 10 mm between the tip of the airfoil and the lower plate. The Mach number of the incoming flow is 0.2 and the Reynolds number based on the chord is 9.3×10^5 . The objective of the present study is to investigate the best way to compute both the aerodynamics and acoustics of the tip leakage flow. In particular, the importance of the inflow conditions on the prediction of the tip leakage vortex and the airfoil loading is underlined. On the other hand, the complex structure of the tip leakage vortex and its convection along the airfoil is recovered thanks to the use of a mesh adaptation based on the dissipation of the kinetic energy. Finally, the ability of the wall law to model the flow in the tip leakage flow region is proven in terms of wall pressure fluctuations and acoustics in the far-field.

Keywords: large-eddy simulation; fan noise; tip leakage flow; tip clearance noise

1. Introduction

Due to strong environmental constraints regarding the noise emitted by aircraft, the bypass ratio of modern turbofan engines tends to increase. It is associated with a reduction of the fan rotation speed, the exhaust jet speed and possibly the nacelle length. When looking at the noise sources of an engine at approach regime, the fan stage is one of the major contributors. In this context, the understanding and prediction of secondary noise sources, such as the tip clearance noise in the fan stage, is required.

In the fan stage of turbofan engines, a gap between the tip of fan blades and the casing wall is present. As a consequence, a highly three dimensional unsteady secondary flow develops. The tip leakage flow goes from the pressure side to the suction side of the blade. When the tip leakage flow leaves the gap, it interacts with the primary flow and rolls up to form the tip leakage vortex. The aerodynamic phenomena are mainly controlled by the blade tip loading, gap height, blade tip thickness, stagger angle, Reynolds and Mach numbers. The consequences of a too strong gap are the drop of the aerodynamic fan performance and the increase of radiated far field noise [1].

This increase of the radiated noise from axial fans was first observed experimentally when the height of the gap increased [2]. Then, source mechanisms responsible for tip clearance noise generation were investigated. Firstly, Kameier and Neise [3] have identified a component of the tip clearance noise called the rotating instability. This mechanism consists of coherent vortical structures coming from the tip clearance which interact with the fan blades, causing periodic fluctuations of the blade loading, and thus inducing tonal noise in the far field. Yet, as these vortices have a range of tangential velocities, broadband humps are observed instead of sharp tonal peaks. This mechanism appears at off design conditions, close to the rotating stall, and the structure of the tip clearance flow region is completely changed. Secondly, Fukano *et al.* [4] studied the tip clearance self noise. The periodic velocity fluctuations generated by the wandering of the tip leakage vortex produce



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tonal noise. Simultaneously, a broadband noise due to the enhancement of stochastic velocity fluctuations in the blade passage is generated. Previous observation was more detailed in the experiment of Jacob *et al.* [5]. Indeed, the authors described the vortical structures generated by the tip leakage flow and observed that they are scattered as sound by the edges of the tip trailing-edge corner, acting like dipole sources. Moreover, they described the jet-like leakage flow as another component of the tip clearance noise, with the characteristic of a quadrupole noise source.

Various numerical studies were performed to investigate the tip clearance noise. An Unsteady Reynolds-Averaged Navier-Stokes (URANS) simulation of a rotor has been achieved by März *et al.* [6] to confirm the experimentally-observed phenomena of rotating instability and to interrogate the physical mechanism behind it. Then, Zhu *et al.* [7] used unsteady aeroacoustic predictions with the Lattice Boltzmann Method (LBM) to shed more light on this noise generation mechanism. Moreover, Boudet *et al.* [8] achieved a Zonal Large-Eddy Simulation (ZLES) of a fan rotor where the region of interest at tip is simulated with full Large Eddy Simulation (LES) and the hub and midspan regions are simulated with Reynolds-Averaged Navier-Stokes (RANS). It allows to identify a tip leakage vortex wandering producing tonal noise. An isolated fixed airfoil with a gap designed to study the tip clearance noise self noise is considered in this paper. ZLES [9], LES [10] and LBM [11] approaches were achieved on this configuration.

The isolated non-rotating airfoil is mounted in an open-jet wind-tunnel facility. This experimental environment is tough to reproduce numerically because of the strong interaction between the jet and the airfoil. Indeed, when testing a lifting airfoil, the main stream is deflected by the equivalent lateral momentum injection, which reduces the effective angle of attack. The flow around an airfoil when installed in a free-jet wind tunnel significantly deviates from that of the same airfoil placed in a uniform stream. A solution to compute the airfoil in an uniform flow is to modify the angle of attack to retrieve the proper airfoil loading. Although the integrated lift can be adjusted in this way, the precise distribution of pressure coefficient is not perfectly recovered. As proposed by Moreau *et al.* [12], one way is to impose a more realistic inlet boundary condition from a precursor RANS calculation. The other way is to take into account the full experiment set-up.

The objective of the present study is to investigate the best way to compute both the aerodynamics and acoustics of the tip leakage flow in order to transfer the methodology to real turbomachinery configurations. To do so, the same experimental set-up will be simulated using two different computational domains, including or modelling the inflow conditions, with a predictive LES approach. The use of a wall model, synthetic-turbulence injection and adaptive mesh refinement is also considered.

The paper starts with a description of the experimental set-up. Then, the numerical set for each configuration is detailed in the second section. In the third section, LES results for the two different computational domain approaches are compared and discussed. Next, the effect of mesh refinement on the prediction of the tip leakage vortex is shown. Finally, the ability of the wall law to model the boundary layer in the gap region is analysed, as well as its impact on the acoustic radiation. Concluding remarks and perspectives are also given in the last section.

2. Experimental set-up

The numerical study is based on the isolated non-rotating airfoil experiment conducted by Jacob *et al.* [13]. Indeed, the advantage is that the tip clearance noise contribution to the far field noise is more easily isolated than in a rotating turbomachinery configuration. A sketch of the experimental set-up is shown in Figure 1. A fixed single airfoil is mounted between two flat plates with a tunable gap between the lower plate and the airfoil tip. Air is coming from a rectangular nozzle. To ensure a uniform flow, the isolated airfoil is placed into the potential core of the rectangular freejet.

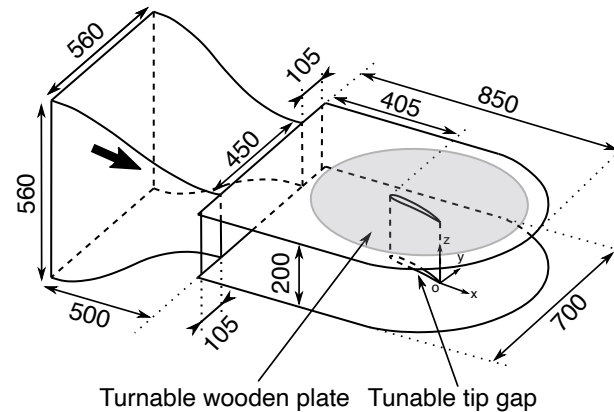


Figure 1. Sketch of the experimental set-up from Jacob *et al.* [13]. Dimensions are in millimeters.

The airfoil is a NACA 5510 of chord $c = 200$ mm. The geometrical angle of attack is $\beta = 16.5^\circ$. The gap height is $s = 10$ mm. The mean flow velocity at the exit nozzle is $U_0 = 70$ m/s, corresponding to a Mach number $Ma = 0.20$ and a Reynolds number based on the chord $Re = U_0 \cdot c / \nu = 9.3 \times 10^5$. One chord upstream of the airfoil, the boundary layer thickness on the plate is 6.2 mm. The experiment is carried out under ambient pressure $p_a = 97\,700$ Pa and ambient temperature $T_a = 290$ K.

The coordinate system $(O, \vec{x}, \vec{y}, \vec{z})$ used in this study is depicted in Figure 1. The origin, defined at the trailing edge-tip corner, is more appropriate to study the tip leakage vortex. The \vec{x} axis is in the streamwise direction. The \vec{y} axis is in the cross-stream direction, from pressure side to suction side. The \vec{z} axis is in the spanwise direction, from the lower to the upper plate.

3. Numerical settings

The simulations performed in this study are based on the LES methodology developed at CERFACS [14,15]. LES are performed using *AVBP*, an explicit, unstructured, massively parallel solver [16] which solves the compressible Navier–Stokes equations. The package *pyhip* [17] to handle unstructured computational grids and their associated datasets is used in combination with the *antares* [18] pre-postprocessing library. In this paper, each LES is performed using the same following set-up. The convective fluxes are computed using the Two-Step Taylor-Galerkin C (TTGC) finite element scheme [19]. It is third-order accurate in time and space. The viscous fluxes are computed using the 2Δ diffusion operator from Colin [20]. Finally, the closure of the LES equations is done using the SIGMA subgrid scale model from Nicoud *et al.* [21]. Regarding the boundary condition, each simulation shared the wall modelling approach and the outlet boundary modelling: a wall law [22] is applied on each wall and a characteristic boundary condition (NSCBC) based on static pressure is applied at outlet [23]. Inlet boundary conditions are detailed below.

In order to define the best approach to predict correctly both airfoil flow-field, tip-leakage vortex, and associated acoustics, it has been chosen to compare the full experimental set-up including the convergent of the open-jet (see Figure 1) with a case where the inlet condition is imposed from a RANS simulation that included a convergent. The computational domains and the boundary conditions are summed up in Figure 2a. The simulation including the convergent is referred to as ‘LES CONV’. In this case, total pressure and temperature are imposed at the inlet of the convergent using dedicated NSCBC [24]. On either side of the nozzle, a colinear flow of 1% of the jet velocity U_0 (7 m/s) is imposed. No synthetic turbulence is injected in this case at inlet.

In the second LES (referred to as ‘LES NO CONV’), in order to save CPU time, the inlet is placed one chord upstream the airfoil leading edge (the blue line in Figure 2a). Mean velocity field and static temperature are specified from a RANS computation [25]. A

fully non-reflecting inlet boundary condition is used to inject three-dimensional turbulence while still being non-reflecting for outgoing acoustic waves [26]. The injected synthetic turbulence which is required to trigger the mixing layers is based on Kraichan's method [27]. The turbulence spectrum has a Passot-Pouquet expression [28]. The Root-Mean-Square (RMS) velocity of the injected turbulent field is the one from the RANS simulation and its most energetic turbulent length scale L_e is 6.3 mm. The latter is computed using a property of the Passot-Pouquet spectrum ($L_e = \sqrt{2\pi}L_t$) and the measured integral length scale L_t (2.5 mm).

In each case, edge size of the mesh around the airfoil is unchanged as depicted with close-ups in Figure 2b. The mesh sizes at the wall of the lower plate and the airfoil are $\Delta x^+ = \Delta y^+ = \Delta z^+ < 100$ in wall units. 20 elements are used to discretise the gap. The total number of tetrahedrons of is 229×10^6 for the case without convergent, whereas it is 252×10^6 with it. The fixed time-step is $3.5 \times 10^{-5} c/U_0$ corresponding to a CFL number of 0.82. In each case, a computational time of $T_{ini} = 7c/U_0$ is required to leave the transient state. Convergence is monitored with pressure probes in the incoming flow, in the tip leakage vortex and on the airfoil. A total of 4096 processors during 70 hours were used to acquire statistics over $T_{sim} = 14c/U_0$. For the same simulated time, the computational cost is increased by 20% when adding the convergent. All calculations were performed on the Joliot-Curie supercomputer in production in CEA's Very Large Computing Centre (TGCC).

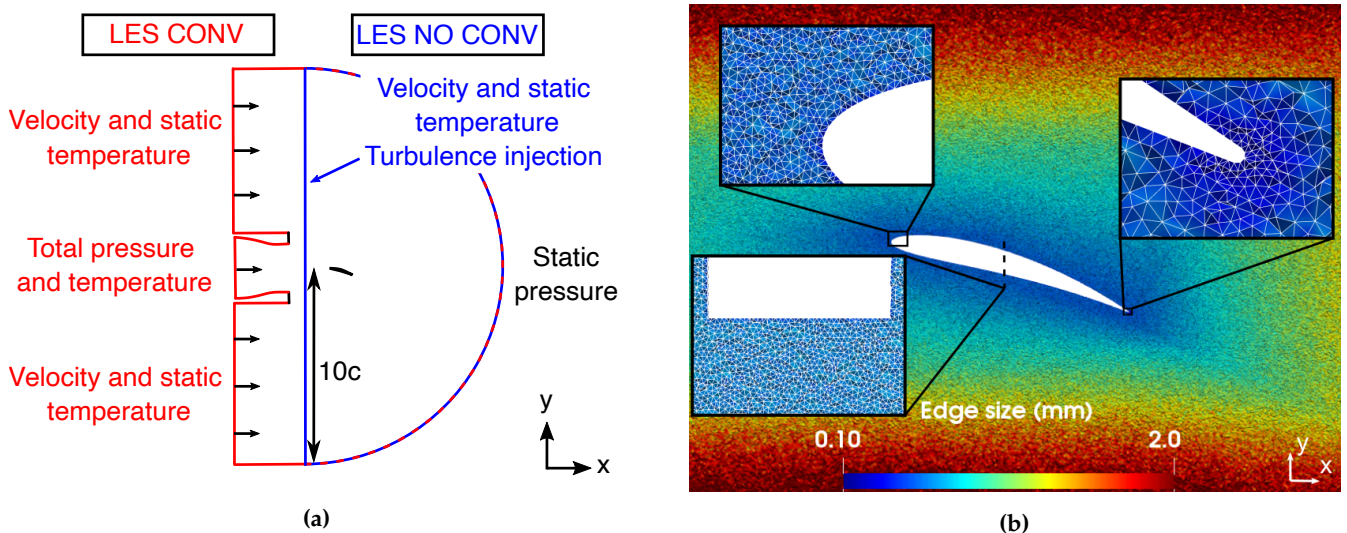


Figure 2. (a) Sketch of the computational domains and the boundary conditions. (b) Edge size of the mesh around the airfoil at $z/c = 0.1$ and close-ups at the airfoil leading and trailing edges and in the gap.

- 1 Table 1 summarizes simulation parameters as well as simulation time and cost. In
- 2 the following, probe data are sampled at 0.01 ms leading to a LES cut-off frequency of 50
- 3 kHz. Welch's method is used to compute Power Spectral Density (PSD) using 10 Hanning
- 4 windows with an overlap of 50%. Instantaneous quantities on the airfoil surfaces are
- 5 dumped every 0.025 ms leading to a cut-off frequency of 20 kHz.

Table 1. Simulation parameters, time and cost of the two computational domain approaches.

	LES NO CONV	LES CONV
Mesh size	229×10^6	252×10^6
Wall resolution $\Delta x^+ = \Delta y^+ = \Delta z^+$	100	100
Wall model	yes	yes
Convective scheme	TTGC	TTGC
Subgrid scale model	SIGMA	SIGMA
$\Delta t c/U_0$	3.5×10^{-5}	3.5×10^{-5}
$T_{sim} c/U_0$	14	14
CPU time	70h	84h

6 4. Effects of inflow conditions

7 4.1. Instantaneous flow

8 In order to have a global view of the flow field in the zone of interest, Figure 3 shows
9 instantaneous iso-surfaces of Q criterion ($Q = 3.0 \times 10^2 (U_0/c)^2$) colored by the velocity
10 magnitude in the tip leakage flow region for LES CONV case. It should be underlined that
11 as the instantaneous flow looks very similar in the LES NO CONV case, it is not shown here.
12 The airfoil is seen from the suction side. Three vortices are identified. The tip separation
13 vortex in the gap is generated by the separation of the tip leakage flow from the airfoil tip.
14 The tip leakage vortex developing from the airfoil leading edge is the major one. Next to it,
15 an induced vortex is generated by the important circulation of the tip leakage vortex. The
16 last two vortices are contra-rotating to each other.

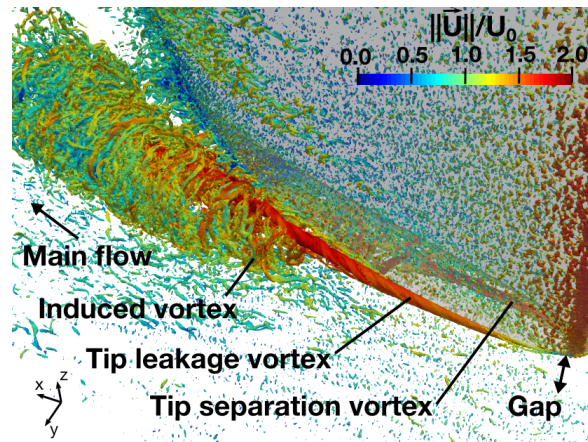


Figure 3. LES instantaneous iso-surfaces of Q criterion ($Q = 3.0 \times 10^2 (U_0/c)^2$) colored by the velocity magnitude in the tip leakage flow region.

17 Figure 4 shows instantaneous vorticity and dilatation fields in both cases at $z/c =$
18 0.1. Large differences are observed between the two approaches. Firstly, in the vorticity
19 field, whereas the tip leakage vortex ($x/c > -0.5, 0 < y/c < 0.5$) and the airfoil wake are
20 similar between the two cases, mixing layers starting from $y/c = -1.0$ and 1.3 are really
21 different. Indeed, considering the full experimental setup with the convergent seems to
22 lead to a more natural growth of the jet mixing layers (Figure 4a) than with the 'LES NO
23 CONV' case in Figure 4b. Secondly, when considering the acoustic field represented by
24 the dilatation field, the case without the nozzle is polluted by a strong numerical spurious
25 noise coming from the inlet. The two sources seem to be located on the jet mixing layers
26 and generated by the interaction of the injected turbulence and the non-constant inflow
27 condition imposed (mean and turbulent velocity fields from RANS).

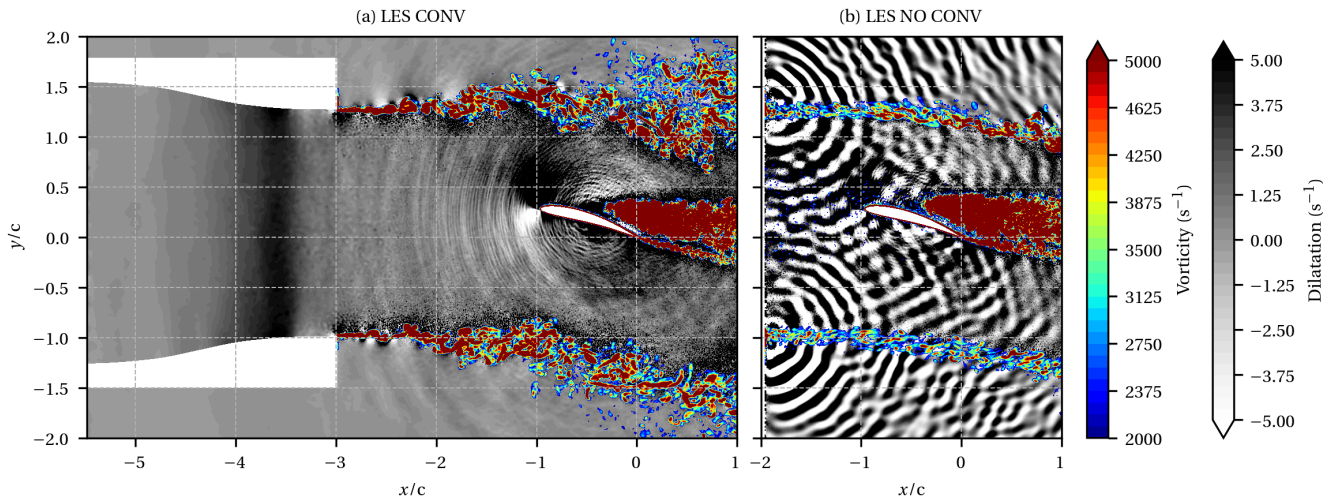


Figure 4. Instantaneous vorticity and dilatation fields with (a) and without (b) the convergent at $z/c = 0.1$.

28 4.2. Mean flow

29 The mean velocity magnitude is presented in Figure 5. The two mixing layers de-
 30 veloped from the convergent exit section of LES CONV (Figure 5a) and from the inlet
 31 of LES NO CONV (Figure 5b) are observed as well as their deviation. Indeed, when the
 32 rectangular jet reaches the airfoil leading edge at $x/c = -1$, it is deflected of about 8° by the
 33 circulation generated by the airfoil. Lobes of velocity around the airfoil interact with the
 34 mixing layers at $x/c = -0.5$, $y/c = \pm 1$ in both cases. However, mixing layer development
 35 and tip leakage vortex signature differ. Indeed, as mentioned before, the mixing layers
 36 exhibit a larger growth with the add of the convergent (Figure 5a) whereas their thickness
 37 in the LES NO CONV case (Figure 5b) seems to remain constant. Regarding the tip leakage
 38 vortex flow region, a deficit of velocity magnitude is observed at $y/c = 0.25$, from $x/c = 0$ in
 39 both cases. It corresponds to the trajectory of the tip leakage vortex. Whereas the structure
 40 of the tip leakage vortex looks similar with and without convergent, the deficit of velocity
 41 magnitude is amplified without the convergent.

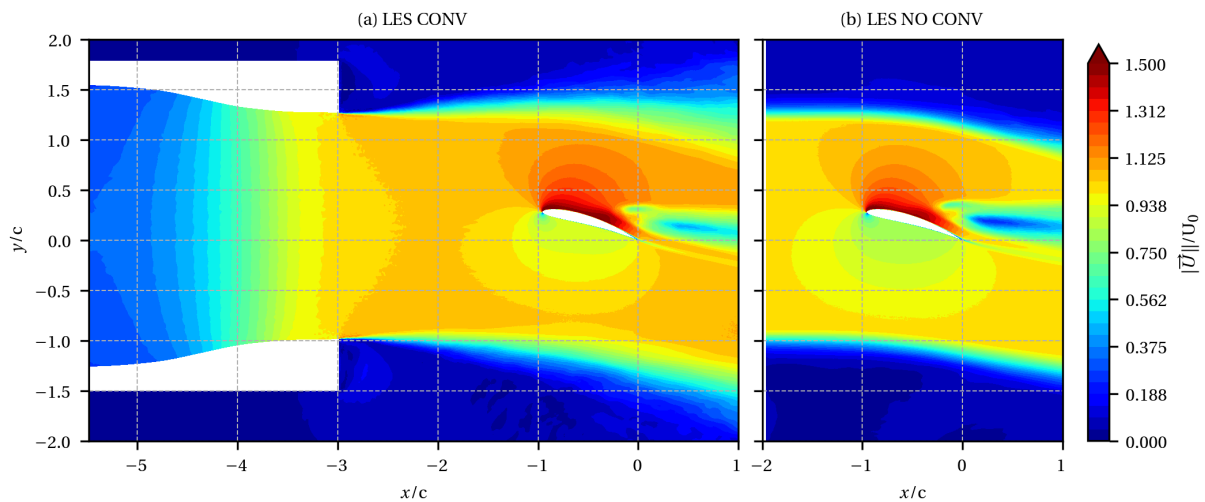


Figure 5. Mean velocity magnitude fields with (a) and without (b) the convergent at $z/c = 0.1$.

42 However, the pressure coefficients on the airfoil presented in Figure 6 are similar in
 43 both cases to the measured ones (in circles) at midspan (Figure 6a) and at the tip (Figure 6b).

44 The LES without the convergent is in blue solid line whereas the LES with the convergent
 45 is in red. The following definition of pressure coefficient is used:

$$C_p = \frac{p - p_0}{0.5\rho_0 U_0^2} \quad (1)$$

46 At midspan (Figure 6a), the LES without the convergent globally matches the ex-
 47 periment. The operating point of the simulation is then validated. A slight deviation is
 48 observed on the suction side, close to the leading edge (upper line for $x/(c.\cos(\beta)) < -0.8$).
 49 The prediction in this region is improved by adding the convergent while the rest of the
 50 airfoil surface exhibits the same level of pressure between the two cases.

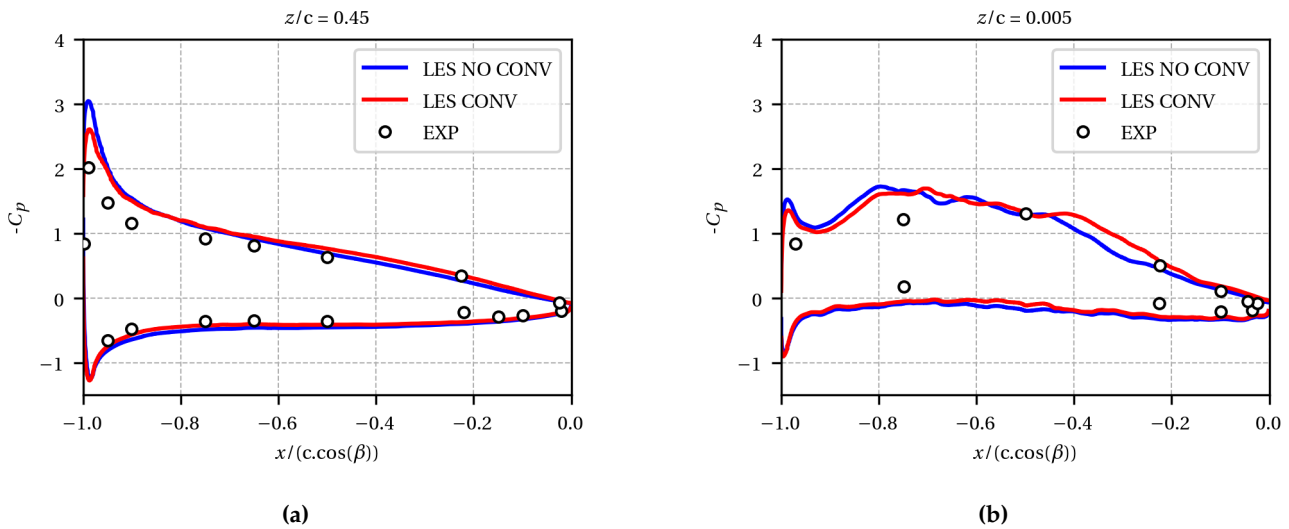


Figure 6. Mean pressure coefficients on the airfoil at midspan, $z/c = 0.45$ (a) and at the tip, $z/c = 0.005$ (b).

51 At the tip (Figure 6b), the measured airfoil loading is globally reduced compared to
 52 the one at midspan. Indeed, the tip leakage flow from the pressure side to the suction
 53 side partially balances the pressure difference. Again, the two computed cases are able to
 54 properly predict the pressure distribution at tip which is a key point of validation. Indeed,
 55 the airfoil tip loading is one of the main parameters which control the tip leakage flow.
 56 For $-0.5 < x/(c.\cos(\beta)) < -0.2$ on the suction side, a difference in the level of pressure is
 57 observed between the two LES. Unfortunately, no measurement has been performed in
 58 this area. Further explanations will be given to understand the difference.

59 5. Tip leakage vortex trajectory

60 Figures 7 shows the streamwise U , horizontal V and vertical W mean velocity compo-
 61 nents of the tip leakage vortex at the airfoil trailing edge ($x/c = 0.01$), from top to bottom,
 62 respectively. LES with and without convergent are compared with 3D Particle Image
 63 Velocimetry (PIV) performed by Jacob *et al.* [13]. Since the tip leakage vortex is roughly
 64 aligned with the x axis, the considered plane is almost perpendicular to the trajectory of
 65 the tip leakage vortex. The flow is viewed from downstream. The velocity components are
 66 normalised by the reference mean velocity U_0 . The airfoil trailing edge is plotted in black
 67 solid line at $y/c = 0$. The white rectangle ($0.0 < y/c < 0.1$) in Figures 7d and 7g defines the
 68 airfoil projected surface as seen from the camera but it has no physical meaning in terms of
 69 velocity since the signal in this region is disrupted by light reflections [13].

70 When looking at the mean axial velocity component U of the tip leakage vortex from
 71 the PIV data (Figure 7a), two distinct regions are identified. Firstly, a strong acceleration
 72 region with a maximum of $1.4U_0$ is measured at $y/c = 0.22$ and $z/c = 0.04$. This position
 73 corresponds to the center of the tip leakage vortex. Secondly, a low velocity region sur-
 74 rounding the zone of acceleration extends from the plate until $z/c = 0.15$. The latter is

75 generated by the detachment of the plate boundary layer by the tip leakage flow. In both
 76 cases, LES predicts a topology that is different from the experiment, but tends to recover the
 77 two regions. It is observed that the LES with convergent captures better the acceleration,
 78 meaning that the incoming flow is more realistic. Nevertheless, velocity magnitudes are
 79 lower than the measured ones. Indeed, in the LES NO CONV, the longitudinal velocity
 80 component at the center of the tip leakage vortex is underestimated of 50% compared with
 81 experiment. When adding the convergent the difference is about 21%. This underprediction
 82 is attributed to the mesh resolution and will be discussed latter.

83 Looking at the PIV measurements in Figures 7d and 7g, a region of positive V is
 84 observed for $z/c < 0.05$ whereas a region of negative V is shown for $z/c > 0.05$. For the
 85 vertical mean component W , two regions are also identified: positive W for $y/c > 0.2$ and
 86 negative W for $y/c < 0.2$. This clearly shows the roll up of the tip leakage vortex. The same
 87 kind of flow topology is noticeable around $y/c = 0.35$ but with a smaller spatial extension
 88 and opposite signs compared to the tip leakage vortex. This flow topology indicates an
 89 induced vortex. In addition, for the horizontal component V , the extension of the region in
 90 red in the gap ($z/c < 0$) brings out the tip leakage flow that feeds the vortex.

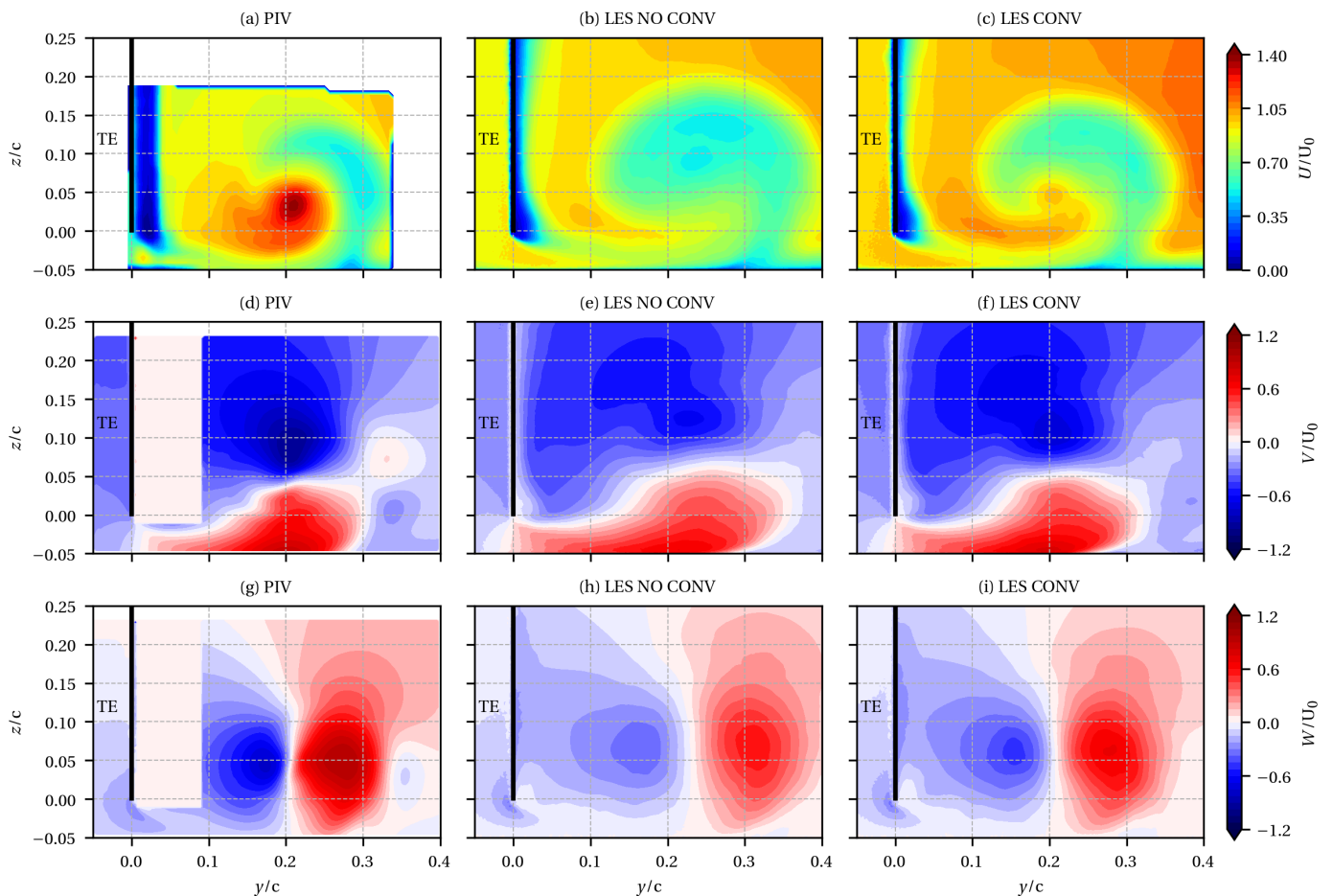


Figure 7. Streamwise U , horizontal V and vertical W mean velocity components of the tip leakage vortex at the airfoil trailing edge ($x/c = 0.01$).

91 The LES without the convergent, in Figures 7e and 7h, correctly reproduces the
 92 topology of the tip leakage flow region but diffusion is noted. Indeed, lower velocity
 93 magnitude is observed and the tip leakage vortex is much more spatially spread out
 94 compared to the PIV. This is even more pronounced for the vertical component W . The LES
 95 with convergent in Figures 7f and 7i also reproduces the topology of the tip leakage vortex
 96 with an improvement on the position of the vortex. Indeed, on the PIV data, the y position

of the tip leakage vortex which is identified by the sudden change of sign on W , is $y/c = 0.2$. Whereas the LES without the convergent predicts the vortex at $y/c = 0.23$, adding the convergent allows to obtain the correct y position of the vortex. A slight improvement is also observed on the z position.

In order to quantify more precisely the tip leakage vortex trajectory, a vortex identification method developed by Graftieux *et al.* [29] is applied. This method is based on the function Γ_1 derived from the velocity field. This function is able to characterise the locations of the large-scale vortex centers, by considering only the topology of the velocity field and not its magnitude.

The function Γ_1 is defined as

$$\Gamma_1(P) = \frac{1}{S} \int_{M \in S} \frac{(\mathbf{PM} \wedge \mathbf{U}_M) \cdot \mathbf{n}}{\|\mathbf{PM}\| \cdot \|\mathbf{U}_M\|} dS \quad (2)$$

where S is a surface surrounding P , M lies in S and \mathbf{n} is the unit vector normal to S . \mathbf{U}_M is the velocity vector at M and \mathbf{PM} is the distance vector between P and M . Γ_1 is dimensionless and $\Gamma_1 \in [-1, 1]$. Γ_1 may be interpreted as the normalized angular momentum of the velocity field. The sign of Γ_1 defines the rotation sign of the vortex. $\Gamma_1 > 1$ is for clockwise rotation whereas $\Gamma_1 < 1$ is for counterclockwise rotation. The center of the vortex is defined as the maximum of $|\Gamma_1|$ with a pragmatic threshold value at 0.9 for validity. The integration over the surface S plays a role of a spatial filter.

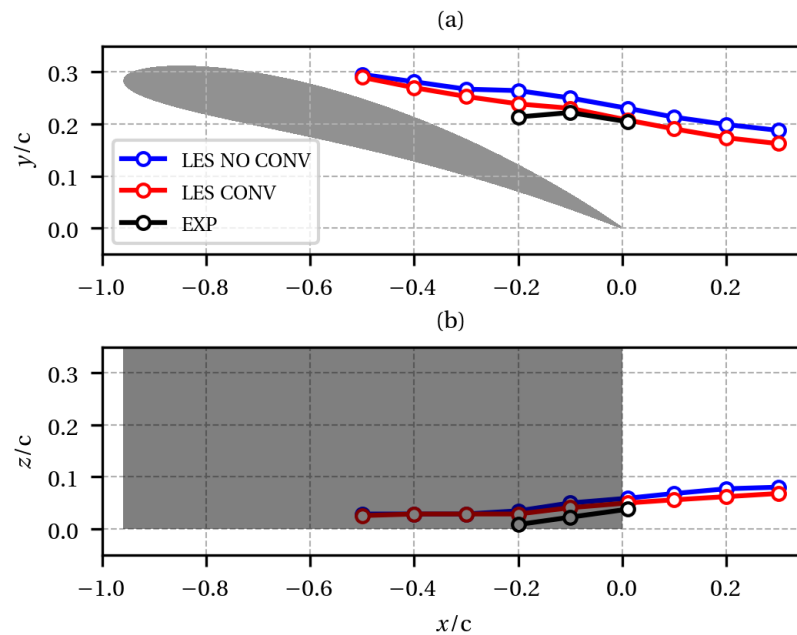


Figure 8. Projected mean trajectory of the tip leakage vortex on planes xOy (a) and xOz (b)

Using the previous algorithm at different spatial positions in the streamwise direction on yOz planes allows to identify the vortex center. The resulting trajectory projected on planes xOy (Figure 8a) and xOz (Figure 8b) is displayed in Figure 8 for the experiment and each LES. The airfoil is in grey shapes. It is observed that if the correct inflow conditions are taken into account as in the LES CONV, the experimental trajectory is well retrieved. As explained by Storer *et al.* [30], the vortices at tip have an influence on the pressure on the airfoil surface. The modification of the trajectory of the tip leakage vortex observed in Figure 8 explains the difference on the pressure coefficient in Figure 6b. With the convergent, the tip leakage vortex is closer to the airfoil, as shown in Figure 8a. Therefore, the pressure on the airfoil surface is lower compared to the case without the convergent.

123 6. Tip leakage vortex convection

124 In the previous section, the mean trajectory of the tip leakage vortex has been improved
 125 in the LES with the convergent (Figure 8). However, the longitudinal velocity acceleration
 126 of the tip leakage vortex, that is to say the convection of the vortex, remains an issue (Figure
 127 7).

128 To improve the prediction of the LES, a mesh adaptation based on the dissipation of
 129 the kinetic energy is performed. Following the approach sets up by Daviller *et al.* [31], a
 130 static h-refinement strategy is used to refine precisely the tip leakage vortex region. From
 131 the previous LES CONV simulation, the time-average dissipation field $\overline{\Phi}$ is used to build a
 132 metric. The quantity of interest $\overline{\Phi}$ is defined as:

$$\overline{\Phi} = \overline{(\mu + \mu_t) \left(\frac{\partial \tilde{u}_i}{\partial x_j} + \frac{\partial \tilde{u}_j}{\partial x_i} \right)^2} \quad (3)$$

with μ the kinematic viscosity and μ_t the local turbulent viscosity computed by the LES
 subgrid scale model. Operators $\tilde{\cdot}$ and $\overline{\cdot}$ represent the LES filtered variables and the time-
 average, respectively. A normalization is first done thanks to the minimum and maximum
 values of $\overline{\Phi}$:

$$\Phi^* = \left[1 - \left(\frac{\overline{\Phi} - \overline{\Phi}_{\min}}{\overline{\Phi}_{\max} - \overline{\Phi}_{\min}} \right) \right]^\alpha \quad (4)$$

Then the metric range is defined using the ϵ parameter:

$$\text{metric} = \Phi^*(1 - \epsilon) + \epsilon \quad (5)$$

133 Using the *pyhip* [17] tool, 38×10^6 tetrahedrons are added to the initial mesh and the
 134 minimal edge size is divided by a factor of 1.12. The magnification factor is set to $\alpha = 100$
 135 and the minimum of the metric field to $\epsilon = 0.7$. The spatial extension of the adaptation is
 136 limited to $z_{\max}/c = 0.5$ spanwise and to $x_{\max}/c = 1.25$ streamwise.

137 The adapted mesh at $z/c = 0.01$ is shown in Figure 9. The mesh has been refined in
 138 the zones of interest, that is to say the tip leakage vortex, the wake and around the airfoil
 139 surface. For the same simulated time, the computational cost is increased by 25%. The
 140 edge size of the mesh before and after adaptation at the airfoil trailing edge is respectively
 141 presented in Figures 10a and 10b.

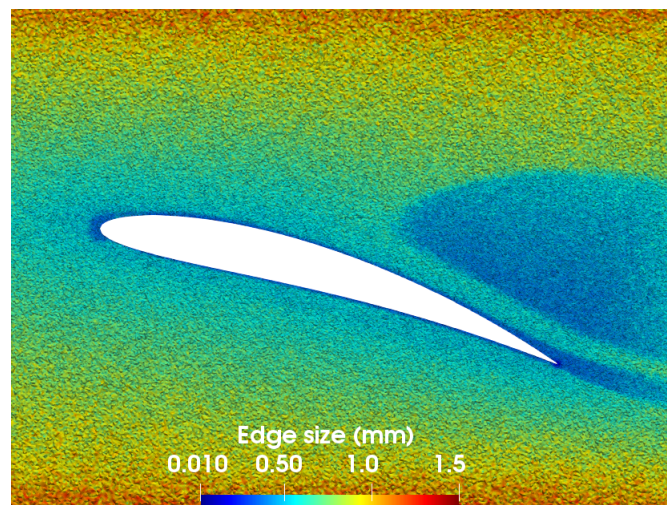


Figure 9. Adapted mesh at $z/c = 0.1$.

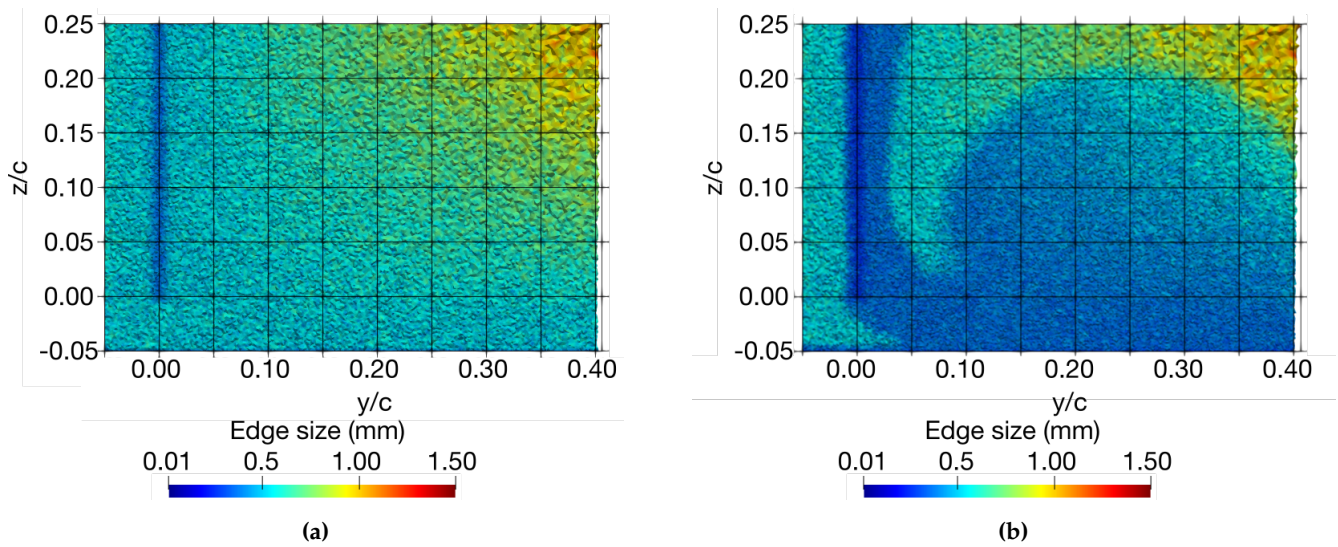


Figure 10. Mesh cuts at the airfoil leading edge at $x/c = 0.01$ before (a) and after (b) adaptation.

142 Figure 11 compares the mean axial velocity U between PIV, LES CONV and LES
 143 ADAPT of the tip leakage vortex at the airfoil trailing edge ($x/c = 0.01$). With the proper
 144 mesh refinement, LES ADAPT is able to better retrieve the topology measured by the PIV.
 145 Indeed, the two velocity regions and even the position of the maximum of U are captured
 146 with less than 15% of error compared to PIV.

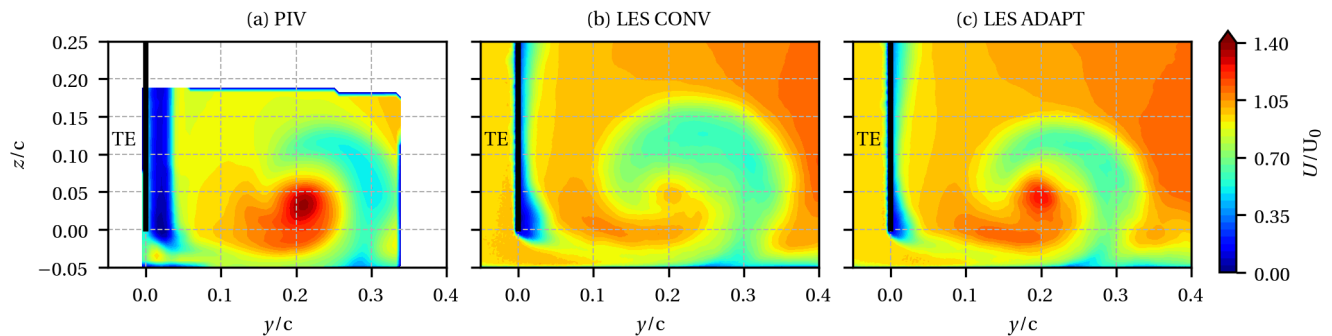


Figure 11. Longitudinal velocity component U of the tip leakage vortex at the airfoil trailing edge ($x/c = 0.01$).

147 To deepen the analysis, 1D velocity profiles are plotted at $z/c = 0.05$ in Figure 12. Using
 148 the mesh adaptation, the predicted velocity profile is clearly improved. Indeed, whereas
 149 the deficit of velocity caused by the airfoil wake is retrieved by both LES around $y/c = 0$
 150 with the correct amplitude, some discrepancies are observed in the tip leakage vortex zone,
 151 which extends from $y/c = 0.17$ to 0.35 . Indeed, the LES with mesh adaptation in green is
 152 able to recover the amplitude of the maximum U at $y/c = 0.2$. Mesh adaptation allows to
 153 recover the complex structure of the tip leakage vortex and especially the acceleration of
 154 the longitudinal velocity component.

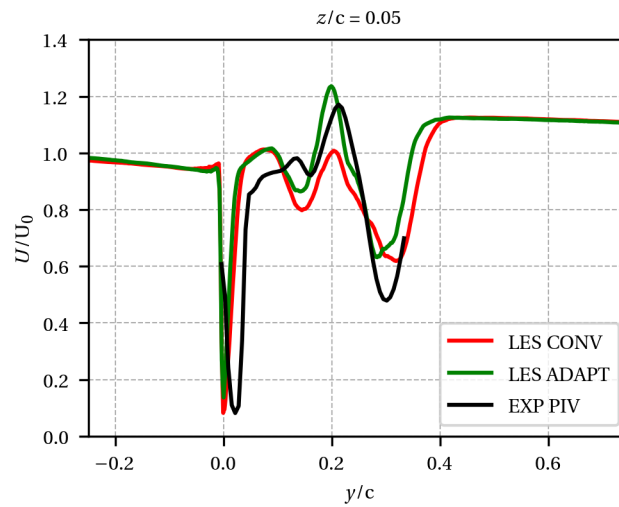


Figure 12. Mean velocity profile of U at $x/c = 0.01$ and $z/c = 0.05$.

155 7. Spectral signature of the tip leakage flow

156 Regarding the high Reynolds number which characterises the flow in a real turboma-
 157 chinery configuration, a wall law is required for computational cost issue. Therefore, the
 158 capability of the wall law to predict the aerodynamics and acoustics of the tip leakage flow
 159 of the isolated airfoil is studied in this section. The wall-modeled LES performed in this
 160 paper is compared to two previous wall-resolved LES from Boudet *et al.* [9] and Koch *et*
 161 *al.* [10]. Please note that these two LES are achieved at an angle of attack of 15° whereas the
 162 current LES is at 16.5° . Previously, Figure 6 showed that the wall law is able to reproduce
 163 the mean pressure distribution on the airfoil surface, especially in the tip region.

164 Figure 13 presents the PSD of the wall pressure fluctuations on the airfoil surface.
 165 For clarity, only results from LES ADAPT are shown. Two positions at 77.5% of chord are
 166 considered. Probe 21 (Figure 13a) is located on the airfoil suction side, 1.5 mm away from
 167 the tip whereas probe B (Figure 13b) is on the airfoil tip, on the camber line. Since wall
 168 pressure spectra of the LES from Koch *et al.* are not available at 77.5%, the spectra at 75%
 169 are used in Figure 13. The three LES are compared to the measurements extracted from
 170 Jacob *et al.* [5]. The experimental cut-off frequency is 22 kHz but data are only available till
 171 10 kHz.

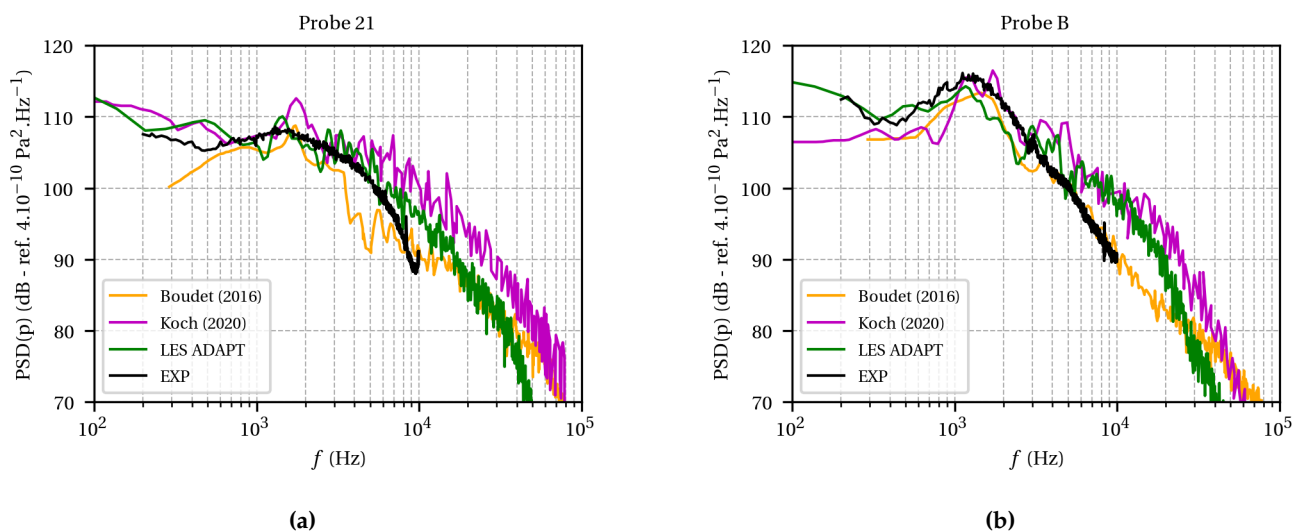


Figure 13. PSD of wall pressure on the airfoil suction side (a) and on the airfoil tip (b) at 77.5% of chord.

172 On probe 21, the LES exhibits a good agreement with the experiment in both shape
 173 and level. The spectrum is even in better agreement than the two wall-resolved cases. The
 174 LES from Koch *et al.* (magenta) that was also performed with *AVBP* exhibits the same
 175 shape than the current LES with a shift in frequency. On probe B, the hump around 1.3
 176 kHz characterises the pressure fluctuations induced by the detachment of the tip leakage
 177 flow on the airfoil pressure side-tip corner. A broadband hump is observed instead of a
 178 tonal peak because of the intermittency of the phenomenon [32]. The LES is able to well
 179 retrieve the hump at 1.3 kHz. For frequencies higher than 6 kHz, a slight overprediction is
 180 observed from the experiment. The LES from Koch *et al.* is again showing the same trend.
 181 The ZLES from Boudet *et al.* remarkably predicts the wall pressure fluctuations even at
 182 high frequencies. Figure 13 shows the capacity of the wall law to predict the wall pressure
 183 fluctuations on the airfoil surface in the tip region.

184 Figure 14 presents the PSD of acoustic pressure in the far-field. The microphone is
 185 placed 2m away from the airfoil suction side, forming an angle of 90° with the airfoil chord.
 186 The acoustic propagation in the far-field is ensured using the solid Ffowcs-Williams and
 187 Hawkings' analogy (FWH). This means that only the dipole sources are taken into account
 188 to estimate the sound; the aforementioned quadrupoles associated with the tip-gap jet are
 189 ignored. The python library *antares* [18] is used following the advanced time formulation
 190 of Casalino [33].

191 The microphone recorded the noise emitted by the airfoil in no-gap (grey) and 10-
 192 mm-gap (black) configurations. It allows to identify a frequency range of the tip clearance
 193 noise from 0.7 to 7 kHz. The wall-resolved LES in orange and magenta are able to retrieve
 194 the noise level in this range. The wall-modeled LES in green is able to predict the noise
 195 level on an even wider range of frequencies. Whereas the acoustic spectra from the two
 196 wall-resolved LES drop for frequencies higher than 7 kHz, the LES presented in this paper
 197 manages to predict the proper noise level. It may be explained by the size of the LES
 198 domain. Indeed, Boudet *et al.* performed a ZLES with a LES zone reduced to the tip region
 199 and Koch *et al.* achieved a LES on a modified geometry with a reduced span. In both cases,
 200 the pressure fluctuations on the airfoil surface are not computed over the full span. This
 201 comparison demonstrates also the capacity of the wall law to model the tip leakage flow
 202 for the purpose of acoustic prediction.

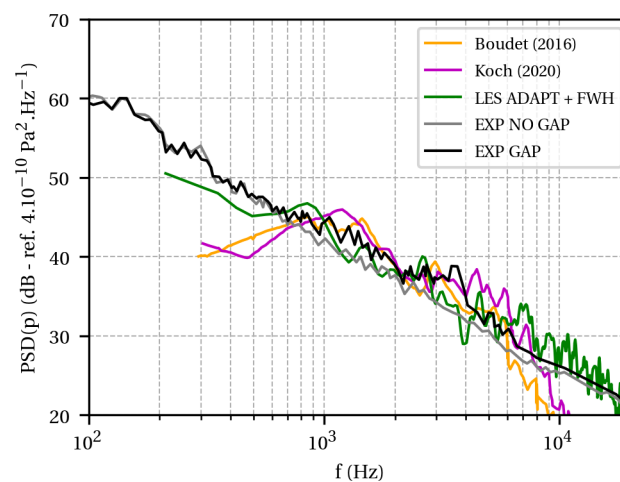


Figure 14. PSD of acoustic pressure 2m away from the airfoil suction side, forming an angle of 90° with the airfoil chord.

203 8. Conclusions

204 With the aim of improving existing prediction models or to model new noise sources
 205 features of the tip clearance noise, a LES of an isolated airfoil with a gap has been performed.
 206 Two computational domains with the same experimental set-up were considered, including

207 or modelling the inflow conditions. It is observed that the LES with a modelling of the
 208 inflow conditions (*i.e.*, without the convergent of the open-jet wind-tunnel facility) allows
 209 to obtain correct results in terms of airfoil loading and mean tip leakage vortex. However,
 210 some deviations were observed when compared to measurements. In particular, the mean
 211 axial velocity of the tip leakage vortex was underestimated and its mean trajectory farther
 212 away from the airfoil. On the other hand, taking into account the full experimental set-up
 213 in the computational domain allowed to correct these differences and better matched the
 214 experiment. This improvement is explained by a more realistic development of the jet
 215 which has a non-negligible interaction with the flow around the airfoil. Moreover, it has
 216 been shown that the use of a mesh adaptation is necessary in order to recover the complex
 217 structure of the tip leakage vortex and especially the acceleration of the longitudinal
 218 velocity component. Finally, the present wall-modelled LES methodology allows to predict
 219 accurately the wall pressure fluctuations on the airfoil surface and the acoustic spectrum
 220 in the far-field. In particular, the frequency range of the tip clearance noise was correctly
 221 captured.

222 Resorting to the LES is essential for the intended future acoustic applications such
 223 as Ultra-High Bypass Ratio turbofan engine, details of which are beyond the scope of
 224 the present paper. Indeed, explicit wall-pressure statistics requiring the simulation of the
 225 turbulence are generally used as input data in the sound prediction models. The wall-
 226 modelled LES strategy developed in this paper was designed to address this issue on more
 227 realistic rotating configurations.

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 230 curation, D.L.; writing—original draft preparation, D.L. and G.D.; writing—review and editing, D.L.
 231 and G.D.; visualization, D.L.; supervision, G.D., M.R. and H.P.; project administration, D.L., G.D.,
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243 Symbols

244	β	Angle of attack ($^{\circ}$)
245	Γ_1	Vortex identification function
246	$\bar{\Phi}$	Time-average dissipation
247	C_p	Pressure coefficient
248	f	Frequency (Hz)
249	p	Static pressure (Pa)
250	T	Static temperature (K)
251	U, V, W	Mean velocity components ($\text{m}\cdot\text{s}^{-1}$)
252	c	Chord (m)
253	Ma	Mach number (-)
254	Q	Q criterion (s^{-2})

255	Re	Reynolds number (-)
256	s	Gap height (m)
257	Acronyms	
258	CFL	Courant–Friedrichs–Lewy.
259	LBM	Lattice Boltzmann Method.
260	LES	Large Eddy Simulation.
261	PIV	Particle Image Velocimetry.
262	PSD	Power Spectral Density.
263	RANS	Reynolds-Averaged Navier-Stokes.
264	RMS	Root-Mean-Square.
265	URANS	Unsteady Reynolds-Averaged Navier-Stokes.
266	ZLES	Zonal Large-Eddy Simulation.

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