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Stabilization and tracking control algorithms for VTOL aircraft: Theoretical and practical overview

J. Betancourt · P. Castillo · R. Lozano

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Abstract Control theory applied to multirotor aerial systems (MAS) has gained attention with the recent increase on the power computation for embedded systems. These systems are now able to perform the calculations needed for a variety of control techniques, with lower cost of sensors and actuators. These types of control algorithms are applied to the position and the attitude of MAS. In this paper, a brief overview evaluation of popular control algorithms for multirotor aerial systems, especially for VTOL - Vertical Take-Off and Landing aircraft, is presented. The main objective is to provide a unified and accessible analysis, placing the classical model of the VTOL vehicle and the studied control methods into a proper context. And therefore, to provide the basis for beginner users working in aerial vehicles. In addition, this work contributes in presenting a comprehensive analysis of the implementation for the Nonlinear and Linear Backstepping, Nested Saturation and the Hyperbolic Bounded Controllers. These techniques are selected and compared to evaluate the performance of the aircraft, by simulations and experimental studies.

Keywords Multirotor Aerial Systems · Classical approach · Virtual approach · Nonlinear/Linear Backstepping · Nested Saturation · Hyperbolic Bounded Control

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1 Introduction

Unmanned Aerial Vehicles systems (UAVs) have been studied extensively in recent years [23,40], due to their high mobility and capacity to perform tasks with complete autonomy. Besides, these vehicles present an ideal test-bed for innovative theoretical approaches to the problem of controlling mechanical systems. They are characterized by their under-actuation, low computational power, high working frequency, and low autonomy [22]. Out of all UAVs, multirotors stand out for their good manoeuvrability, stability and payload. The first studies of these multirotors vehicles were based on stabilizing their attitude because it represents the main challenge before flying the vehicle. VTOL vehicles, specially the PTVOL - Vertical Take-Off and Landing - or the Quadrotor configurations, are taken as particular cases of study and several control methodologies have been used to stabilize them, we can cite for example; Backstepping [54,47,17], Sliding Mode Control [51,27,37], PID [1,19,32,41,33], optimal control [26,28,10,45], robust control [8,53,12], learning-based control [35], and others [3,55,31].

1.1 Related work

In control systems for multirotor aerial vehicles there are two different types of control, depending on the loop to which the controller is applied; the position controller and the attitude controller. In fact, there is no direct actuation control per se. Attitude control is the concept of pointing a craft in the desired direction. More specifically, it involves controlling the orientation of the robot on its three Euler angular velocities (roll rate $\dot{\theta}$, pitch rate $\dot{\phi}$ and yaw rate $\dot{\psi}$) and controlling the robots thrust in the Z-axis of the robot frame of reference. This is performed due to the underactuation characteristics of the robot. Then, the outputs of this control are sent to a mixing-of-motors algorithm to generate the reference signals for the actuators of the MAVs. In these cases, the attitude control algorithm must take into account the number of motors and their respective control signal. In contrast, position control is the controller that gives the Euler angles as references to the attitude controller in order for the MAV to move in a desired trajectory [25].

For the last 10 years, the two dynamics attitude and position were popularly controlled by monitoring the altitude (z -axis) using feedback linearization followed by a linear/nonlinear controller to achieve the desired altitude. The values of the control torque can be founded by replacing the controlled altitude in the atti-

tude expression. This method for designing the control torque will be denoted *classical* approach in this paper. In addition, adaptive control has been studied to solve the problems of stabilization and tracking of the dynamics. Hossein et al. proposed in [6] a novel trajectory tracking scheme by combining two inner feedback linearization loops to stabilize the nominal quadrotor. Moreover, Wenchao presented in [24] a robust adaptive tracking control scheme based on a self-tuning regulator. This tracking method is implemented in the inner loop and a classical proportional integrator (PI) controller is to be employed in the outer loop. In this way, Wenchao verified that the robustness of the whole system is guaranteed. Similarly, Tomashevich introduced the altitude and attitude control laws for the quadrotor based on the passivity method [44]; The simulation results demonstrate high efficiency and robustness with respect to the plant parametric uncertainty.

Furthermore, a time-varying backstepping technique is presented by Santiaguillo-Salinas et al. [39], where a 3D control tracking law is studied and applied experimentally on a quadrotor. This control law is implemented by assuming that only the position and orientation parameters of the vehicles are known; whereas the linear and angular velocities are estimated using Luenberger observers. Rolsado-Serrano in [36] improved the work developed by Santiaguillo-Salinas by considering a non-linear time-varying version of the backstepping technique. This work takes all the non-linearities of the dynamics into account; resulting in a higher stability validated theoretically with numerical simulations.

Moreover, dynamic inversion is a new control technique developed in [4,9] for the stabilization and dynamics tracking of a quadrotor vehicle. The tracking controller is composed of two loops; The dynamic inversion is implemented in the inner loop, where an internal dynamics stabilization is applied to the outer loop. The control objectives are enclosed in the form of constraint differential equation, and the resultant control law is obtained by inverting the constraint dynamics using Moore-Penrose Generalized Inverse (MPGI). Computer simulations were performed to exhibit the performance of the offered control technique in nominal and perturbed conditions. In addition to the dynamic inversion controllers, Antonio-Toledo et al. proposed in [5] an integral backstepping controller with sliding mode control approach for a quadrotor. In this approach the control of the UAV is performed in the altitude and position control. The altitude control produces translational force, which is used to calculate virtual controls for the X and Y movements and desired angles for pitch and roll. The proposed controller was tested in a Qball-

X4 prototype. In the same sense, J. Escareño and S. Salazar addressed a hierarchical control in [11], based on sliding-mode and adaptive control techniques, to deal with slow and fast time-varying wind conditions respectively. A backstepping technique is used to stabilize the inner-loop heading dynamics. Simulations results show the validity of the proposed control strategy while tracking a time-parametrized straight-line and sinusoidal trajectory. In addition, Ibarra-Jimenez and Castillo designed in [18] a controller that ensures the quick convergence in the dynamics of the quadrotor in order to catch a ball before it touches the ground. The algorithm is based on the sliding mode approach to assure robustness and finite time convergence. Simulations results verifies the performance of the controller to catch the ball even in presence of unknown and external disturbances.

Recently, a new fashion to design the torque control of the vehicle is considering the vehicle model fully actuated by means of a virtual input. In this paper, we will call the aforementioned as *virtual* approach. In this context, there are some works exploiting the differential flatness of the quadrotor model and then, with a trajectory generation algorithm, compute high-performance flight trajectories that are capable of moving a quadrotor from a large class of initial states to a given target point that will be reached at rest [16,13,29]. The algorithm can compute a feasible trajectory within tens of microseconds; the remaining computation time is used to iterative improve the trajectory. For example, Riu Wang proposed in [48] a control scheme based on a virtual control input using a backstepping approach with Nussbaum function, a priori-bounded control torque for the rotational subsystem was designed to track the desired orientations generated by the translational subsystem. In the same sense, Xu et al. addressed in [50] a robust cost controller derived by Lyapunov stability theorem. The designed robust guaranteed cost controller ensures the closed-loop system asymptotically stable and the robustness with respect to all admissible parameter uncertainties. Besides, Warier et al. solved in [49] the tracking control problem in two-step approach. First, a translational control scheme that tracks the desired position trajectory is constructed assuming the translational dynamics to be fully actuated. The magnitude of the translational control input is used as the magnitude of the control thrust. Second, the unit vector representing the direction of the translational control input used as the desired thrust direction.

This paper gives an overview focused on the two existing methods to design attitude and positions controllers to solve the stabilization and tracking problems

that have been proposed in the time span from the first works in MAV control in 2002 to date. Our research has identified more than two hundred works, most of which are conference papers. We have selected for this document only journal papers and the most prominent recent conferences, so that the reader can understand the evolution of the state-of-the-art in multirotor control. Therefore, four of these control algorithms have been chosen to be implemented and compared. These control algorithms are described in detail and applied to a simulation model and validated through a series of experimental tests. To the best of the authors knowledge, this is the first time that the *virtual* control algorithm based on hyperbolic bounded functions is implemented in real time experiments.

The remainder of the paper is structured as follows: The simplified dynamic model of a PVTOL vehicle is given in section 2. We use this configuration because most multirotors with parallel motors can be seen as a generalization of the PVTOL vehicle. The control algorithms are introduced in section 3, details in their conception are given using the Lyapunov theory. Section 4 reports the numerical and experimental validation and Section 5 a discussion of the control schemes and their performances are given. Finally, section 6 details the conclusions of this work.

2 The PVTOL aircraft

As previously explained, the PVTOL aircraft is the most simple VTOL configuration, it represents the longitudinal model of the helicopter, the simplified model of a plane moving in the horizontal plane and the quadcopter can be seen as two PVTOL interconnected, as shown in Fig 8. It is an underactuated system and is composed by two independent motors that produce a force and a torque on the plane. Therefore, this configuration will be used to better illustrate the stability analysis of the algorithms. Moreover, the results can be expanded for other VTOL configurations.

The dynamical model is presented following these assumptions

- The vehicle structure is rigid and symmetrical,
- Its center of mass and O_B (body frame) coincides,
- The pitch moment creates a force that is perpendicular to z axis,

Let consider $\mathcal{I} = \{O_I, x_I, z_I\}$ an inertial frame and $\mathcal{B} = \{O_B, x_B, z_B\}$ a rigid frame attached to center of mass of the vehicle.

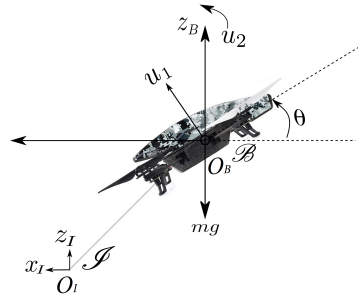


Fig. 1: Schematic model of the PVTOL aircraft.

The dynamics of a rigid body under external forces using the Newton-Euler formalism can be expressed as

$$\begin{aligned} \dot{\xi}(t) &= v(t), & m\dot{v}(t) &= R(t)F(t), \\ \dot{R}(t) &= R(t)\hat{\Omega}(t), & J\dot{\Omega} &= -\Omega(t) \times J\Omega(t) + \tau(t) \end{aligned} \quad (1)$$

where $\xi(t)$ denotes the position of the vehicle with respect to the frame \mathcal{I} , $v(t) \in \mathcal{I}$ describes the linear velocity, $\Omega(t)$ represents the angular velocity of the body defined in \mathcal{B} , and m is the total mass of the vehicle. The constant moment of inertia is denoted by J expressed in \mathcal{B} , $\tau(t)$ expresses the torques applied in the rigid body, $\hat{\Omega}(t)$ introduces the skew-symmetric matrix of $\Omega(t)$, $R(t)$ means the rotation matrix from \mathcal{B} to \mathcal{I} , and $F(t)$ are the forces applied to the rigid body.

From (1) and Figure 8, the nonlinear dynamic model of a PVTOL aircraft can be represented as

$$m\ddot{x} = -u_1 \sin \theta + \epsilon u_2 \cos \theta \quad (2)$$

$$m\ddot{z} = u_1 \cos \theta + \epsilon u_2 \sin \theta - mg \quad (3)$$

$$J\ddot{\theta} = u_2 \quad (4)$$

where x and z are respectively the lateral and vertical coordinates of the aircraft center of mass, θ denotes the pitch angle with respect to the horizon, u_1 describe the main thrust composed by the forces produced by each motor f_i , u_2 represent the pitching moment and g corresponds to the acceleration due to gravity. ϵ represents the coupling value between the torque and the lateral force, in general, it is not well known and in several cases neglected. Thus, the simplified nonlinear model for the PVTOL with $\epsilon = 0$ can be described in the following form

$$m\ddot{x} = -u_1 \sin \theta \quad (5)$$

$$m\ddot{z} = u_1 \cos \theta - mg \quad (6)$$

$$J\ddot{\theta} = u_2 \quad (7)$$

Note that in [34], it was proved that by an appropriate change of coordinates, (2)-(4) can be represented as (5)-(7) without neglecting the term ϵ . From the above, the

input vector can be defined as

$$\mathbf{u} = \begin{bmatrix} u_1(t) \\ u_2(t) \end{bmatrix} = \begin{bmatrix} 1 & 1 \\ l & -l \end{bmatrix} \begin{bmatrix} f_1(t) \\ f_2(t) \end{bmatrix}. \quad (8)$$

where l denotes the distance between one rotor to the center of mass of the vehicle and f_1 and f_2 are the forces produced by motor 1 and 2 respectively.

3 Control architectures

Typically in the literature when controlling a VTOL vehicle is assumed the z -axis can be controlled by some linear or nonlinear controller using specifically u_1 . Usually, this controller is based on a feedback linearization technique as presented in [30, 14, 15, 42]. This *classical* procedure is often used because the idea is to guarantee the vehicle keeps at hover, and later control their displacement in the plane x or y making an under-actuated subsystem.

Following these ideas, the controller for (6) is very often proposed as

$$u_1 = \frac{m}{\cos \theta} (g + r_1) \quad (9)$$

where r_1 is always designed to achieve $z \rightarrow z_d$, for instance $r_1 = -k_1(z - z_d) - k_2\dot{z}$, where k_1, k_2 represent positive constants and z_d denotes the desired altitude. Introducing (9) into (5)–(6), it follows

$$m\ddot{x} = -(-k_2\dot{z} - k_1(z - z_d) + g)m \tan \theta \quad (10)$$

$$\ddot{z} = -k_2\dot{z} - k_1(z - z_d) \quad (11)$$

Note from (11), \exists a time T large enough such that for $t > T$, z and \dot{z} are arbitrarily small such that (10) can be expressed as

$$\ddot{x} \approx -g \tan \theta \quad (12)$$

For simplifying further analysis, consider $J = 1$. Then rewriting (12) and (7), it yields

$$\begin{aligned} \dot{x}_1 &= x_2 \\ \dot{x}_2 &= -g \tan \theta_1 \\ \dot{\theta}_1 &= \theta_2 \\ \dot{\theta}_2 &= u_2 \end{aligned} \quad (13)$$

where $x_1 = x$, $x_2 = \dot{x}$, $\theta_1 = \theta$, $\theta_2 = \dot{\theta}$.

Notice that (13) describes an under-actuated subsystem representing the longitudinal dynamics of a VTOL vehicle. These dynamics will be used for conceiving three control algorithms using two different techniques; the backstepping and the saturation functions. These methodologies will be referred in the following as classical control inputs.

3.1 Nonlinear Backstepping algorithm

Define the error e_1 as $e_1 = x_1 - x_1^d$, where x_1^d is the reference. Let us propose the following positive function $V_1 = \frac{k_1}{2}e_1^2$, with $k_1 > 0$ denoting a constant, then taking the derivative with respect to time and proposing $x_2^v = x_2^d - e_1$ as a virtual control input, it follows that

$$\dot{V}_1 = k_1 e_1 (x_2 - x_2^v - e_1) = -k_1 e_1^2 + k_1 e_1 (x_2 - x_2^v)$$

where $\dot{x}_1^d = x_2^d$. Define the error $e_2 = x_2 - x_2^v$ thus, the previous yields $\dot{V}_1 = -k_1 e_1^2 + k_1 e_1 e_2$. Propose the second positive function $V_2 = \frac{k_2}{2}e_2^2$ with $k_2 > 0$ constant. Taking its derivative with respect to time and proposing $\delta_1^v = -\dot{x}_2^v + \frac{k_1}{k_2}e_1 + e_2$ representing a second virtual control input, implies that

$$\dot{V}_2 = -k_2 e_2^2 - k_1 e_1 e_2 + k_2 e_2 (\delta_1^v - g \tan \theta_1)$$

Defining the error $e_3 = \delta_1^v - g \tan \theta_1$, \dot{V}_2 yields

$$\dot{V}_2 = -k_2 e_2^2 - k_1 e_1 e_2 + k_2 e_2 e_3$$

Propose the third positive function $V_3 = \frac{k_3}{2}e_3^2$, with $k_3 > 0$ constant. Differentiating V_3 and defining $\delta_2^v = \dot{\delta}_1^v + \frac{k_2}{k_3}e_2 + e_3$, it follows that

$$\dot{V}_3 = k_3 e_3^2 - k_2 e_3 e_2 + k_3 e_3 (\delta_2^v - g(1 + \tan^2 \theta_1)\theta_2)$$

Defining the error $e_4 = \delta_2^v - g(1 + \tan^2 \theta_1)\theta_2$, then the above yields

$$\dot{V}_3 = -k_3 e_3^2 - k_2 e_3 e_2 + k_3 e_3 e_4$$

Proposing $V_4 = \frac{k_4}{2}e_4^2$ as positive function, with $k_4 > 0$ defining a constant, and taking its derivative, it appears that

$$\dot{V}_4 = k_4 e_4 (\delta_2^v - g(1 + \tan^2 \theta_1)(u_2 + 2\theta_2^2 \tan \theta_1))$$

Proposing a control law given by

$$u_2 = \frac{1}{g(1 + \tan^2 \theta_1)} (\delta_2^v + \frac{k_3}{k_4}e_3 + e_4) - 2\theta_2^2 \tan \theta_1 \quad (14)$$

implies that $\dot{V}_4 = -k_4 e_4^2 - k_3 e_4 e_3$.

Finally, define the following candidate Lyapunov function with the form $V_T = V_1 + V_2 + V_3 + V_4$. Therefore

$$\dot{V}_T = -k_1 e_1^2 - k_2 e_2^2 - k_3 e_3^2 - k_4 e_4^2 \leq 0.$$

Expressing (14) with respect to the state variables, it follows that

$$\begin{aligned} u_2 &= \frac{1}{g(1 + \tan^2 \theta_1)} (\ddot{x}_2^d - 4\dot{x}_2^d - \bar{k}_1 \dot{x}_2^d + \bar{k}_2 (x_2 - x_2^d) \\ &\quad + \bar{k}_3 (x_1 - x_1^d) - \bar{k}_4 g \tan \theta_1 \\ &\quad - 4g(1 + \tan^2 \theta_1)\theta_2) - 2\theta_2^2 \theta_1 \end{aligned}$$

where

$$\begin{aligned} \bar{k}_1 = \bar{k}_4 = k_\theta + 6, \quad \bar{k}_2 = 2k_\theta + 4 \\ \bar{k}_3 = k_\theta + \frac{k_3 k_1}{k_4 k_2} + 1, \quad k_\theta = \frac{k_1}{k_2} + \frac{k_2}{k_3} + \frac{k_3}{k_4} \end{aligned} \quad (15)$$

3.2 Control algorithm based on nested saturation

The following control algorithm uses saturation functions to impose a bound in the control input. The first methodology was proposed by [43] designing the controller with nested saturation functions for linear systems and applied to nonlinear systems. Nevertheless after analysis, it can be observed that the controller structure using saturation functions imposes bounds in each state, such that, the following inequality can be applied $\tan \theta \approx \theta$, see [38, 7]. From this assumption, (13) can be rewritten as follows

$$\begin{aligned} \dot{x}_1 &= x_2 \\ \dot{x}_2 &\approx -g\theta_1 \\ \dot{\theta}_1 &= \theta_2 \\ \dot{\theta}_2 &= u_2 \end{aligned} \quad (17)$$

A controller stabilizing (17) can be denoted by

$$u_2 = -\sigma_a(\theta_2 + \sigma_b(\cdot)) \quad (18)$$

where σ_i defines a generalized saturation function such that $|\sigma_i(\cdot)| \leq i$ for positive constant $i : a, b, c, d$. The argument in $\sigma_b(\cdot)$ will be defined later to assure convergence of the states. Let us define a positive function $V_1 = \frac{1}{2}\theta_2^2$, then its derivative is defined by

$$\dot{V}_1 = \theta_2 \dot{\theta}_2 = \theta_2 u_2 = -\theta_2 \sigma_a(\theta_2 + \sigma_b(\cdot)) \quad (19)$$

if $|\theta_2| \geq b \Rightarrow \dot{V}_1 \leq 0$. Then, $\exists t_1$ such that for $t \geq t_1$ $|\theta_2(t)| \leq b$, implying that $|\theta_2 + \sigma_b(\cdot)| \leq 2b$. Choosing $a \geq 2b$, then (18) can be rewritten $\forall t > t_1$ as

$$u_2 = -\theta_2 - \sigma_b(\cdot). \quad (20)$$

Define $\nu_1 = \theta_1 + \theta_2$, then, $\dot{\nu}_1 = -\sigma_b(\cdot)$. Propose a positive function $V_2 = \frac{1}{2}\nu_1^2$ and imposing $\sigma_b(\cdot) = \sigma_b(\nu_1 + \sigma_c(\cdot))$ then, its derivative is determined by

$$\dot{V}_2 = \nu_1 \dot{\nu}_1 = -\nu_1 \sigma_b(\nu_1 + \sigma_c(\cdot)) \quad (21)$$

if $|\nu_1| \geq c \Rightarrow \dot{V}_2 \leq 0$. Then, $\exists t_2 \geq t_1$ such that for $t \geq t_2$ $|\nu_1(t)| \leq c$ and $|\nu_1 + \sigma_c(\cdot)| \leq 2c$. Choosing $b \geq 2c$, then (20) can be rewritten $\forall t > t_2$ as

$$u_2 = -\theta_2 - \nu_1 - \sigma_c(\cdot). \quad (22)$$

Define $\nu_2 = \nu_1 + \theta_1 - x_2/g = \theta_2 + 2\theta_1 - x_2/g$, thus $\dot{\nu}_2 = -\sigma_c(\cdot)$. Propose the positive function $V_3 = \frac{1}{2}\nu_2^2$ with $\sigma_c(\cdot) = \sigma_c(\nu_2 + \sigma_d(\cdot))$, then its derivative is described by

$$\dot{V}_3 = \nu_2 \dot{\nu}_2 = -\nu_2 \sigma_c(\nu_2 + \sigma_d(\cdot)) \quad (23)$$

if $|\nu_2| \geq d \Rightarrow \dot{V}_3 \leq 0$. Then, $\exists t_3 \geq t_2$ such that $\forall t \geq t_3$ $|\nu_2(t)| \leq d$ and $|\nu_2 + \sigma_d(\cdot)| \leq 2d$. Choosing $c \geq 2d$, then (22) can be rewritten $\forall t > t_3$ as

$$u_2 = -\theta_2 - \nu_1 - \nu_2 - \sigma_d(\cdot). \quad (24)$$

Define $\nu_3 = \theta_2 + 3\theta_1 - 3\frac{x_2}{g} - \frac{x_1}{g}$ then $\dot{\nu}_3 = -\sigma_d(\cdot)$. Propose $V_4 = \frac{1}{2}\nu_3^2$ and $\sigma_d(\cdot) = \sigma_d(\nu_3)$, taking the derivative of V_4 , it follows that

$$\dot{V}_4 = \nu_3 \dot{\nu}_3 = -\nu_3 \sigma_d(\nu_3) \leq 0 \quad (25)$$

The previous implies that $\nu_3 \rightarrow 0$, then from (23) it follows that $\nu_2 \rightarrow 0$. From (21) implies that $\nu_1 \rightarrow 0$, similarly from (19), $\theta_2 \rightarrow 0$. From definition of ν_1 it follows that $\theta_1 \rightarrow 0$. From definition of ν_2 this implies that $x_2 \rightarrow 0$. And finally, from definition of ν_3 it can be deduced that $x_1 \rightarrow 0$.

Rewriting u_2 , it yields

$$\begin{aligned} u_\theta = -\sigma_a(\theta_2 + \sigma_b(\theta_1 + \theta_2 + \sigma_c(\theta_2 + 2\theta_1 + x_2 \\ + \sigma_d(\phi_2 + 3\phi_1 - 3\frac{x_2}{g} - \frac{x_1}{g}))) \end{aligned} \quad (26)$$

3.3 Linear Backstepping controller

This controller is also useful when assuming that the vehicle is moving with small angles. The procedure is similar that for the nonlinear backstepping, nevertheless the obtained algorithm is quite different and has the form of a PD controller. For conceiving the algorithm a linear model is necessary, for our study, system (17) will be used.

Rewriting first equation of (17)

$$\dot{x}_1 = x_2 \quad (27)$$

propose $V_1 = \frac{1}{2}x_1^2$, then $\dot{V}_1 = x_1 x_2$. if $x_2 \rightarrow -k_1 x_1$ with k_1 is a positive constant, then $\dot{V}_1 = -k_1 x_1^2$. Define the following error with the form

$$\zeta_2 = x_2 - x_2^v \quad (28)$$

with $x_2^v = -k_1 x_1 = \alpha_1$ defining the first virtual input. Rewriting (27) and the second equation of (17) with 28 and using $x_1 = \zeta_1$

$$\begin{aligned} \dot{\zeta}_1 &= \zeta_2 + \alpha_1 \\ \dot{\zeta}_2 &= -g\theta_1 - \alpha_1 \end{aligned} \quad (29)$$

Define a positive definite function $V_2 = V_1 + \frac{\zeta_2^2}{2}$, then

$$\dot{V}_2 = \dot{V}_1 + \zeta_2 \dot{\zeta}_2 = \dot{V}_1 + \zeta_2(-g\theta_1 - \dot{\alpha}_1) \quad (30)$$

if $(-g\theta_1 - \dot{\alpha}_1) \rightarrow -k_2\zeta_2$ with k_2 is a positive constant, then, $\dot{V}_2 = \dot{V}_1 - k_2\zeta_2^2$. Define the error

$$\zeta_3 = -g\theta_1 - \alpha_2 \quad (31)$$

with $\alpha_2 = k_2\zeta_2 - \dot{\alpha}_1$ denoting the second virtual input. Rewriting (29) with the previous equation, it follows that

$$\begin{aligned} \dot{\zeta}_1 &= \zeta_2 + \alpha_1 \\ \dot{\zeta}_2 &= -g\theta_1 - \dot{\alpha}_1 \\ \dot{\zeta}_3 &= -g\theta_2 - \dot{\alpha}_2 \end{aligned} \quad (32)$$

Define the positive definite function $V_3 = V_2 + \frac{\zeta_3^2}{2}$, then

$$\dot{V}_3 = \dot{V}_2 + \zeta_3 \dot{\zeta}_3 = \dot{V}_2 + \zeta_3(-g\theta_2 - \dot{\alpha}_2) \quad (33)$$

if $(-g\theta_2 - \dot{\alpha}_2) \rightarrow -k_3\zeta_3$, with k_3 is a positive constant, then $\dot{V}_3 = \dot{V}_2 - k_3\zeta_3^2$. Define the error

$$\zeta_4 = -g\theta_2 - \alpha_3 \quad (34)$$

where $\alpha_3 = (g\theta_2)^v = k_3\zeta_3 - \dot{\alpha}_2$ is the third virtual input. Rewriting (32) with ζ_4 , it follows that

$$\begin{aligned} \dot{\zeta}_1 &= \zeta_2 + \alpha_1 \\ \dot{\zeta}_2 &= -g\theta_1 - \dot{\alpha}_1 \\ \dot{\zeta}_3 &= -g\theta_2 - \dot{\alpha}_2 \\ \dot{\zeta}_4 &= -gu_2 - \dot{\alpha}_3 \end{aligned} \quad (35)$$

Propose the Lyapunov candidate function $V_4 = V_3 + \frac{\zeta_4^2}{2}$, then

$$\dot{V}_4 = \dot{V}_3 + \zeta_4 \dot{\zeta}_4 = \dot{V}_3 + \zeta_4(-gu_2 - \dot{\alpha}_3) \quad (36)$$

Propose the control input

$$u_2 = \frac{1}{g}(k_4\zeta_4 - \dot{\alpha}_3) \quad (37)$$

with $k_4 > 0$ is a constant. Then

$$\dot{V}_4 = \dot{V}_3 - k_4\zeta_4^2 = -k_1\zeta_1^2 - k_2\zeta_2^2 - k_3\zeta_3^2 - k_4\zeta_4^2 < 0 \quad (38)$$

The above implies system (35) goes to zero implying that (17) is globally asymptotically stable. Rewriting (37) with respect to the state variables, it follows that

$$u_2 = -\frac{\bar{k}_1}{g}x_1 - \frac{\bar{k}_2}{g}x_2 + \bar{k}_3\theta_1 + \bar{k}_4\theta_2 \quad (39)$$

where

$$\begin{aligned} \bar{k}_1 &= k_1k_2k_3k_4 \\ \bar{k}_2 &= k_1k_2(k_3 + k_4) + k_3k_4(k_1 + k_2) \\ \bar{k}_3 &= k_1(k_2 + k_3 + k_4) + k_2(k_3 + k_4) + k_3k_4 \\ \bar{k}_4 &= k_1 + k_2 + k_3 + k_4 \end{aligned} \quad (40)$$

3.4 Fully actuated approach

Previous approaches, as explained, use mainly the control input u_1 to stabilize the altitude of a VTOL while u_2 is necessary to control the underactuated subsystem. Some works in literature propose to rewrite the system constructing a cascade structure which only consists of two fully actuated subsystems by means of a virtual input. Motivated by the works developed by [46,2], we propose simple smooth bounded controllers that can easily be implemented in VTOL aircrafts with parallel motors for tracking set-points and time-varying trajectories.

The methodology is explained as follows; propose a vector control input, F , containing the virtual control laws to stabilize the translational states in the VTOL vehicle. These virtual control inputs will be related with a desired orientation matrix, R_d , of the vehicle and with its main control input, u , as follows

$$F = uR_d \quad (41)$$

In our case of study, the virtual control laws will be designed for the vertical and longitudinal axis of the PVTOL with the form $F = [F_1 \ F_2]^T$, similarly $R_d \in \mathbb{R}^{2 \times 1}$, is defined as

$$R_d = \begin{bmatrix} \sin \theta_d \\ \cos \theta_d \end{bmatrix}.$$

Next step, is to rewrite the original dynamic system, in our case it is represented by equations (5)-(7). Thus the new system is described as

$$\begin{aligned} m\ddot{q} &= u_1R + mge_2 \\ J\ddot{\theta} &= u_2 \end{aligned} \quad (42)$$

where $q = [x \ z]^T$, $R = [\sin \theta \ \cos \theta]^T$, and $e_2 = [0 \ 1]^T$. Notice that in (41), u is the main control input, in our case, $u = u_1$. Therefore, using (41) into first equation of (42)

$$\begin{aligned} m\ddot{q} &= u_1R + mge_2 + u_1R_d - u_1R_d \\ &= u_1(R - R_d) + mge_2 + u_1R_d \end{aligned}$$

Define $\omega = u_1(R - R_d)$, thus

$$m\ddot{q} = \omega + mge_2 + F. \quad (43)$$

The desired orientation can be found using (41), then

$$\theta_d = \tan \left(\frac{F_1}{F_2} \right)^{-1}. \quad (44)$$

Notice from (41) that R_d is an orthogonal vector, therefore

$$u_1 = \sqrt{F_1^2 + F_2^2}. \quad (45)$$

Define the control objective

$$\lim_{t \rightarrow \infty} \|q(t) - q_d(t)\| = 0. \quad (46)$$

where $q_d = [x_d \ y_d]^T$ are the longitudinal and vertical references. Denoting the tracking error as $p = [p_1 \ p_2]$ such that

$$p_1 = q - q_d \quad p_2 = \dot{q} - \dot{q}_d \quad (47)$$

Differentiating (47) and by means of (43)

$$\begin{aligned} \dot{p}_1 &= p_2 \\ \dot{p}_2 &= \frac{\omega}{m} - g e_2 + \frac{F}{m} - \ddot{q}_d. \end{aligned} \quad (48)$$

where \ddot{q}_d means the desired acceleration.

3.4.1 Control based on hyperbolic functions

Let us propose F in (48) as

$$F = m(-\sigma_1(Ap_1 + Bp_2) - \sigma_2(Bp_2) + g e_2 + \ddot{q}_d) \quad (49)$$

where $A, B \in \mathbb{R}^{2 \times 2}$ are diagonal positive matrices constant, σ_i signifies a saturation function with the form $\sigma(\varsigma) = \bar{\sigma} \tanh(\varsigma)$ and $\bar{\sigma}$ means a bounded constant.

Hence, the bounded control F in (49) makes system (48) globally asymptotically stable.

Proof: First, to prove the asymptotic convergence of solution $p(t)$ to the origin, observe from (41) and (43) that $\|\omega\| \leq 2\bar{u}$. This implies that the closed-loop system using (48) and (49) satisfies the Lipschitz condition¹, and hence has a unique solution over $[0, T]$ with $T \geq 0$. Moreover, if T is bounded, then $p(t)$ is bounded $\forall t \in [0, T]$. This proof is inspired in [52].

Furthermore, if $\lim_{t \rightarrow \infty} \|\omega(t)\| = 0$ then $\lim_{t \rightarrow \infty} \|p(t)\| = 0 \forall p(0)$, that is, for every ϵ_1 , for every $p(0)$, there exists $\epsilon_2, T_{\epsilon_1}, T_{\epsilon_2} > 0$ and $T_{\epsilon_1} \geq T_{\epsilon_2}$ such that

$$\|\omega(t)\| < \epsilon_2, \forall t \geq T_{\epsilon_2} \Rightarrow \|p(t)\| < \epsilon_1, \forall t \geq T_{\epsilon_1}. \quad (50)$$

Towards this end, inspired by the work developed by [46], let us propose the following candidate Lyapunov function

$$V_1(p) = \sum_{i=1}^2 \int_0^{(a_i p_{1i} + b_i p_{2i})} \sigma_{1i}(\varsigma_i) d\varsigma_i + \frac{1}{2} p_2^T A p_2 \quad (51)$$

where $a_i, p_{1i}, b_i, p_{2i}, \sigma_{1i}$ and ς_i for $i = 1, 2$ are the elements of A, B, p_1, p_2, σ_1 and ς respectively. The time derivative of $V_1(p)$ along (48) and (49) is

$$\begin{aligned} \dot{V}_1 &= p_1^T A \sigma_1(Ap_1 + Bp_2) + p_2^T B \sigma_1(Ap_1 + Bp_2) \\ &\quad + p_2^T A p_2 \\ &= -\sigma_1(Ap_1 + Bp_2) B \sigma_1(Ap_1 + Bp_2) - \sigma_2(Bp_2) B \\ &\quad \sigma_1(Ap_1 + Bp_2) + \frac{\omega^T}{m} B \sigma_1(Ap_1 + Bp_2) \\ &\quad - \sigma_2(Bp_2) A p_2 + \frac{\omega^T}{m} A p_2 \\ &\leq -\lambda_{\min}(B) \|\sigma_1(Ap_1 + Bp_2)\|^2 + \lambda_{\max}(B) \\ &\quad \|\sigma_1(Ap_1 + Bp_2)\| \|\sigma_2(Bp_2)\| - \lambda_{\min}(A) p_2^T \sigma_2(Bp_2) \\ &\quad + \frac{\omega^T}{m} (A p_2 + B \sigma_1(Ap_1 + Bp_2)) \end{aligned} \quad (52)$$

Using Young's inequality¹, the second term of (52), satisfies

$$\begin{aligned} &\lambda_{\max}(B) \|\sigma_1(Ap_1 + Bp_2)\| \|\sigma_2(Bp_2)\| \\ &\leq \frac{\lambda_{\max}(B)}{2} \|\sigma_1(Ap_1 + Bp_2)\|^2 \\ &\quad + \frac{\lambda_{\max}(B)}{2} \|\sigma_2(Bp_2)\|^2 \end{aligned} \quad (53)$$

Furthermore, there exists $\nu > 0$ such that $\|\sigma_2(Bp_2)\| \leq \nu \lambda_{\max}(B) \|p_2\|$, for all p_2 , then the third term of (52) satisfies

$$\|\sigma_2(Bp_2)\|^2 \leq \nu \lambda_{\max}(B) p_2^T \sigma_2(Bp_2) \quad (54)$$

Substituting (4) into (53), yields

$$\begin{aligned} \|\sigma_1(Ap_1 + Bp_2)\| \|\sigma_2(Bp_2)\| &\leq \frac{\|\sigma_1(Ap_1 + Bp_2)\|^2}{2} \\ &\quad + \frac{\nu \lambda_{\max}(B)}{2} p_2^T \sigma_2(Bp_2) \end{aligned} \quad (55)$$

From (55), the derivative \dot{V}_1 in (52) satisfies

$$\dot{V}_1 \leq -W_1(p, \omega) \quad (56)$$

where

$$\begin{aligned} W_1(p, \omega) &= c_1 \|\sigma_1(Ap_1 + Bp_2)\|^2 + c_2 p_2^T \sigma_2(Bp_2) \\ &\quad - \frac{\|\omega\|}{m} (\lambda_{\max}(A) \|p_2\| + c_3) \end{aligned} \quad (57)$$

with $c_1 = \frac{2\lambda_{\min}(B) - \lambda_{\max}(B)}{2}$, $c_2 = \frac{2\lambda_{\min}(A) - \nu \lambda_{\max}^2(B)}{2}$, $c_3 = \lambda_{\max}(B) \bar{\sigma}_1$ with $\bar{\sigma}_1 \geq \|\sigma_1(\cdot)\|$. In order to ensure $c_1, c_2, c_3 > 0$, the matrices A, B are chosen as follows:

$$2\lambda_{\min}(B) > \lambda_{\max} B; \quad 2\lambda_{\min}(A) > \nu \lambda_{\max}(B)^2 \quad (58)$$

(56) is used to get (50). Let the ball $\mathcal{B}_{\epsilon_1} = \{p \in \mathbb{R}^3 \mid \|p\| \leq \epsilon_1\}$, and show that $p(t)$ approaches \mathcal{B}_{ϵ_1} after the time

¹See [20], Theorem 3.1 and page 446

T_{ϵ_1} . Let $\alpha_1 = \min_{\|p\|=\epsilon_1} V_1(p)$, then $\alpha_1 > 0$. Take $\beta_1 \in (0, \alpha_1)$ and define the set $\Omega_{\beta_1} = \{p \in \mathcal{B} \mid V_1(p) \leq \beta_1\}$, then Ω_{β_1} is in the interior of \mathcal{B}_{ϵ_1} . As $V_1(p)$ is continuous and $V_1(0) = 0$, there exists γ_1 such that $0 < \gamma_1 < \epsilon_1$ and $\beta_2 = \max_{\|p\|=\gamma_1} V_1(p) < \beta_1$. Let the ball $\beta_{\gamma_1} = \{p \in \Omega_{\beta_1} \mid \|p\| \leq \gamma_1\}$, and take $\alpha_2 = \min_{\|p\|=\gamma_1} V_1(p)$, then we have $0 < \alpha_2 < \beta_2 < \beta_1 < \alpha_1$. Consequently, let the ball $\beta_{\gamma_2} = \{p \in \mathcal{B}_{\gamma_1} \mid \|p\| \leq \gamma_2\}$ where γ_2 satisfies $0 < \gamma_2 < \gamma_1$ and $0 < \min_{\|p\|=\gamma_2} V_1(p) \leq \max_{\|p\|=\gamma_2} V_1(p) < \alpha_2$, then from the above construction we have

$$\mathcal{B}_{\gamma_2} \subset \mathcal{B}_{\gamma_1} \subset \Omega_{\beta_1} \subset \mathcal{B}_{\epsilon_1} \quad (59)$$

and the Lyapunov function $V_1(p)$ satisfies

$$\begin{aligned} \max_{\|p\|=\gamma_2} V_1(p) &< \min_{\|p\|=\gamma_1} V_1(p) \\ &< \max_{\|p\|=\gamma_1} V_1(p) < \beta_1 < \alpha_1. \end{aligned} \quad (60)$$

In view of (59) it is sufficient to show that $p(t)$ approaches \mathcal{B}_{γ_2} after the finite time T_{ϵ_1} . Then, the proof is completed. \square

As the PVTOL is composed by mechanical and electrical parts, constraints are needed in order to respect the limits of its actuators and mechanical movements. Using (45) and (49) it yields

$$\begin{aligned} u^2 &= F_1^2 + F_2^2 \\ u^2 &\leq (\sigma_{11} + \sigma_{21})^2 m^2 + (\sigma_{12} + \sigma_{22} + g)^2 m^2. \end{aligned} \quad (61)$$

In order to ensure $u \leq \bar{u}$, where \bar{u} is a thrust bounded constant defined by the properties of the actuators, and from (61), then the following condition is necessary

$$(\sigma_{11} + \sigma_{21})^2 + (\sigma_{12} + \sigma_{22} + g)^2 \leq \frac{\bar{u}^2}{m^2}. \quad (62)$$

To guarantee the orientation angle θ tracks its desired value θ_d in the sense that $\lim_{t \rightarrow \infty} \|\theta(t) - \theta_d(t)\| = 0$, such that

$$\lim_{t \rightarrow \infty} \|\omega(t)\| = 0 \quad (63)$$

Let us propose a Proportional Derivative controller with the form

$$u_2 = -k_\theta(\theta - \theta_d) - k_\dot{\theta}(\dot{\theta} - \dot{\theta}_d) \quad (64)$$

where k_θ and $k_\dot{\theta}$ are positive constants that must be well chosen to ensure (63). $\dot{\theta}_d$ rate is defined as

$$\dot{\theta}_d = \frac{\dot{F}_1 F_2 - F_1 \dot{F}_2}{F_2^2 + F_1^2} \quad (65)$$

with

$$\begin{aligned} \dot{F} &= -\frac{1}{m}((1 - \sigma_1 \tanh(Ap_1 + Bp_2))^2 Ap_2 \\ &\quad + B((1 - \sigma_1 \tanh(Ap_1 + Bp_2))^2 \\ &\quad + (1 - \sigma_2 \tanh(Bp_2))^2) \dot{p}_2). \end{aligned}$$

4 Simulations and experimental results

This section presents the simulation and experimental results to validate the control strategies developed in (15), (26), (39), (49). The scenario consists in moving the longitudinal axis of the vehicle as following: the vehicle starts at $x(0) = -2m$ and $z(0) = 1m$, then, three set-points defined by $x_{r_1} = 2$, $x_{r_2} = 4$ and $x_{r_3} = 5$ in meters are imposed as desired values.

For better illustrating the graphs obtained in simulation and experimental results, we denote as *LB*, *NLB*, *NS*, the Linear/NonLinear Backstepping and the Nested Saturation controllers, respectively, called also *classical* approaches. The *HSC* means the Hyperbolic Saturation Controller based on the fully actuated approach, named also *virtual* method. Parameters for numerical and practical validation are shown in Table 1.

Table 1: Parameters values used in the control laws in (15), (26), (39) and (49) used in simulation and experimental validation.

Controller	Control parameters	Sim. Values	Exp. Values
LB	k_1	1	1.31
	k_2	1.2	1
	k_3	0.8	0.8
	k_4	0.1	0.1
NLB	\bar{k}_1	0.04	0.04
	\bar{k}_2	0.035	0.0025
	\bar{k}_3	0.4	0.4
	\bar{k}_4	0.1	0.1
NS	a	0.15	0.15
	b	0.22	0.22
	c	0.1	0.1
	d	0.3	0.3
HSC	σ_1	0.5	0.5
	σ_2	0.1	0.2
	σ_3	0.5	0.5
	σ_4	0.35	0.4
	a_{11}	5	1
	a_{22}	0.9	1.3
	b_{11}	3	0.3
	b_{22}	0.55	0.6

4.1 Simulations

The performance of the PVTOL vehicle when using controllers (15), (26), (39) and (49) tracking the set-points previously defined is shown in Figures 2 and 4. Note from Figure 2 that, the convergence using con-

trollers without bounded functions are faster than the others that use saturation functions. This performance is 'normal' because the saturation functions are chosen so small to guarantee a convergence from any initial condition. To increase the speed of convergence of these controllers, the bound of the saturation functions must be increased. Besides, as depicted in Figure 3, the θ angle responses for controllers NS and HSC are bounded, even if initial conditions are far from the desired position. In addition, for the controllers LB and NLB the angular position is bigger than for the NS and HSC controllers and its magnitude is related with the error position. Notice that only for the HSC algorithm the angular position tracks the imposed reference.

Figure 4 is the most important graph to analyze the performance of the four controllers. In this figure z performance is depicted showing the advantage of using virtual controller from the classical one. Observe here that the controllers using classical procedure their altitude performance is degraded when tracking the references. In practical validation, this is observed as a drop in the altitude performance (z decreases). This drop is related to how far the desired value is. Similarly, the performance of the vehicle comparing the four control algorithms in the $x - z$ plane is displayed in Figure 5. In one hand, it can be verified that when using control algorithms based on *classical* method, the behaviour of the vehicle in the z -axis is compromised. On the other hand, controllers based on the *virtual* approach compensate this error by means of the correction term F_2 in the total thrust.

In Figures 6 and 7 the control input responses for the four controllers are depicted. Notice from Figure 6

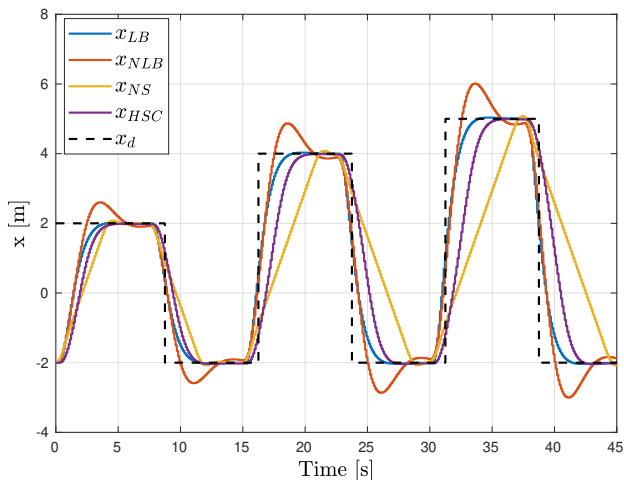


Fig. 2: Longitudinal position performance when comparing the four control algorithms.

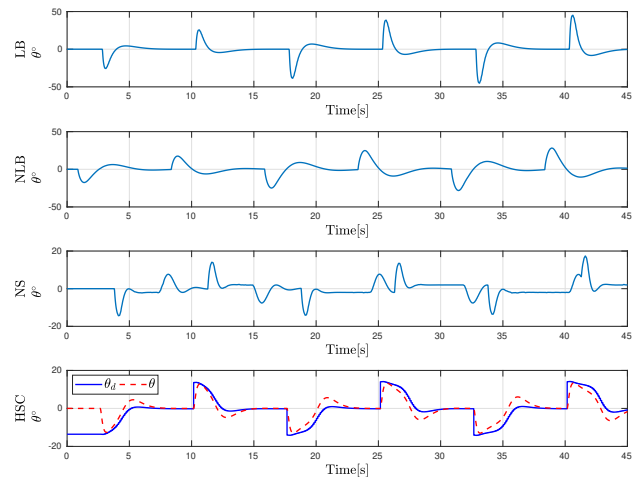


Fig. 3: θ angle response when applying four controllers into the PVTOL dynamics. θ_d denotes the desired angle computed in (44).

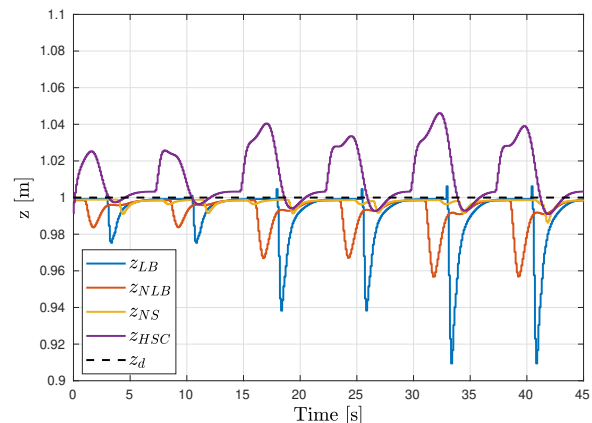


Fig. 4: Vertical position behavior when comparing the four control algorithms.

that u_1 response is different for the HSC controller with respect to the others that decreases when the desired point is modified. Notice that for the HSC controller u_1 increases to compensate the lateral displacement. Nevertheless, in Figure 7 the u_2 performance is quite different. Notice here that u_2 for the HSC controller is smaller with respect to the others one.

4.2 Experimental results

The control algorithms (15), (26), (39) and (49) were validated in real time in a quadcopter vehicle evolving as a PVTOL vehicle. For this, front motors (f_{L_2}, f_{R_2}) and rear motors (f_{L_1}, f_{R_1}) in the aerial vehicle are considered producing only a force in the front and rear side of the vehicle, as depicted in Figure 8.

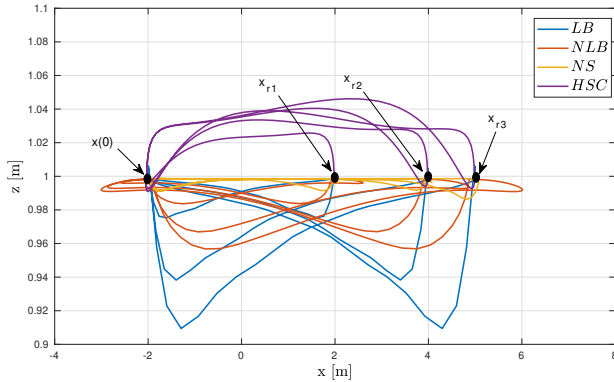


Fig. 5: Vehicle's performance evolution in the $x - z$ plane when applying the four control algorithms into the PVTOL dynamics. From figure x_{r_i} for $i : 1, 2, 3$ describes the desired way-point.

From Figure 8 notice that $f_1 = f_{L1} + f_{R1}$ and $f_2 = f_{L2} + f_{R2}$. Vertical displacement on the z axis is produced when increasing or decreasing, by the same magnitude the speed in the motors, while pitch moment is produced with the difference between f_1 and f_2 .

This quadcopter vehicle evolving as a PVTOL is an AR Drone 2. Its firmware was modified to work under the software *Fl-AIR - Framework libre AIR* which is open source and runs a Linux-based operating system, capable of implementing a wide range of control schemes, see [21]. An OptiTrack motion capture system was used to estimate the vehicles position, while its internal Inertial Measurement Unit (IMU) measures its orientation and angular rates.

Conditions for flight tests are the same used in sim-

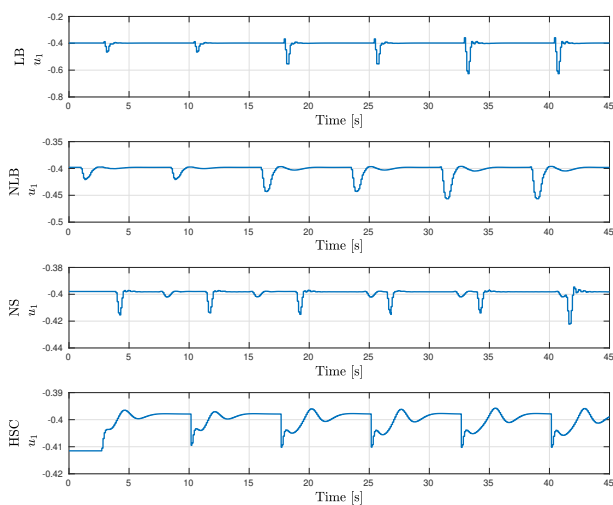


Fig. 6: u_1 behavior when applying the controllers (15), (26), (39) and (49).

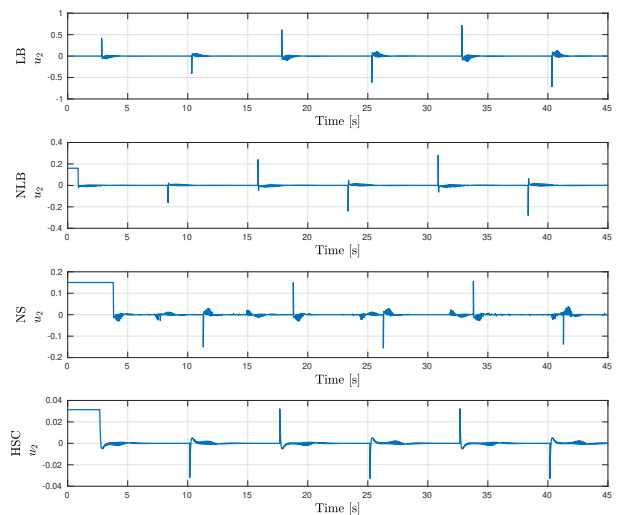


Fig. 7: Torque control u_2 response when applying the four control algorithms.

$z(0) = 1m$ while $x_{r_1} = 2$, $x_{r_2} = 4$ and $x_{r_3} = 5$ in meters are the references. In Figures 9- 9 the state response when applying the controllers are illustrated.

Notice from Figure 9, as in the simulation case, the convergence of controllers based on saturation functions are quite slower due to the bound of these functions, that helps to maintain an relative small error with respect to the others controllers.

The angular performance of the system is depicted in Figure 11. Notice here that for the HSC algorithm a desired angle is imposed that is tracked satisfactory. Similarly in this figure observe that θ -response is bigger for the other controllers.

Observe in Figure 11, that the magnitude of the error in the z -axis is directly related to the x -axis error.

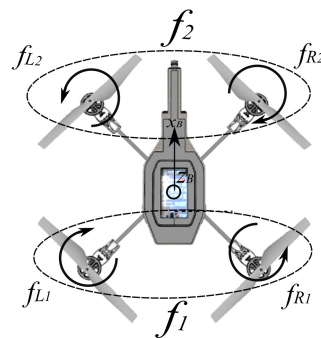


Fig. 8: Quadcopter configuration evolving in its longitudinal plane with the main forces acting in the vehicle.

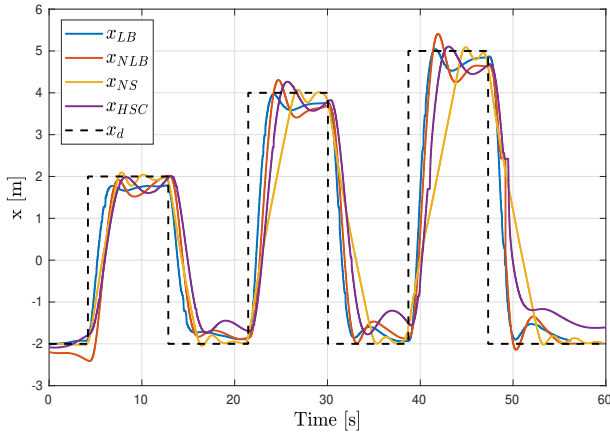


Fig. 9: Longitudinal behavior when applying experimentally the four control algorithms into the aerial vehicle.

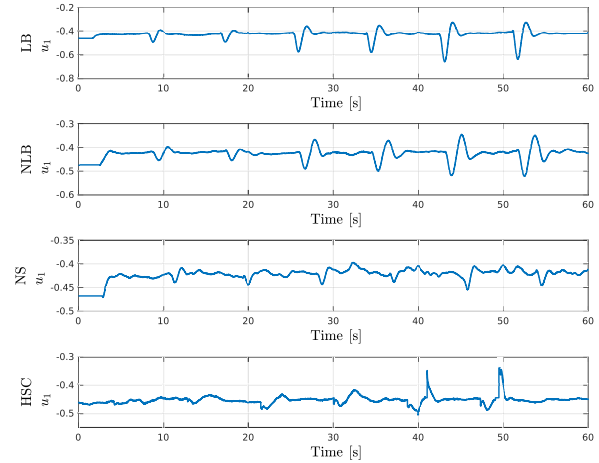


Fig. 13: Thrust control u_1 response for the four control algorithms applied to the aerial vehicle.

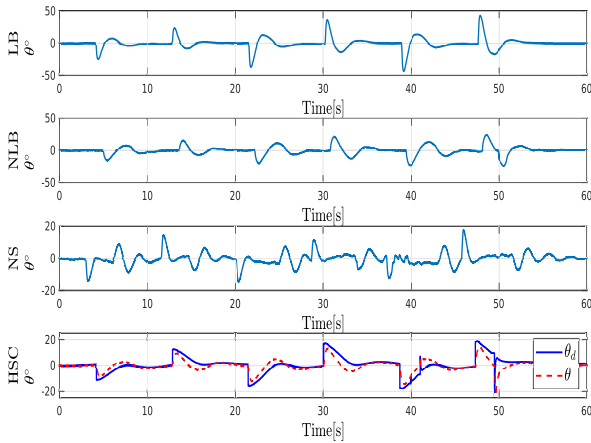


Fig. 10: θ angle response when the four control algorithms were applied in real time. θ_d describes the desired angle.

This means that if the desired x -position is far, then, the error in the vertical axis increases. And as a consequence, the magnitude of the θ angle is affected as can be verified in Figure 10. Therefore, as can be observed in Figure 10, it is possible to use small values for the saturation functions, nevertheless it will compromise the performance of the system.

Moreover, a big angle response should increase the vertical error in the vehicle when using control algorithms based on *classical* approach as can be seen in Figure 12a. Besides, when using controllers based on the *virtual* approach or controllers using bounded functions, this error is minimized as shown in Fig. 12b.

In Figures 13 and 14 the control input responses are illustrated. Notice that they have similar performance obtained in simulations.

5 Discussion

We have carried out an analysis of the performance of the four control algorithms. For this purpose, the performance indices and a qualitative comparison of the controller are obtained. The performance indices are the Integral Square Error (ISE), the Integral Absolute Error (IAE), the Integral time Squared Error (ITSE) and the Integral time Absolute Error (ITAE). A summary of this analysis is presented in Tables 2 - 4.

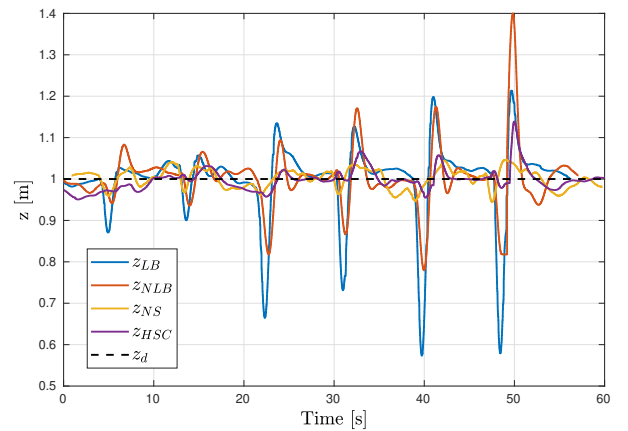
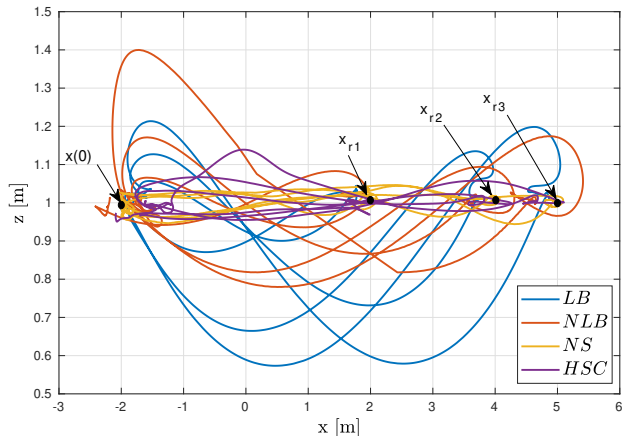
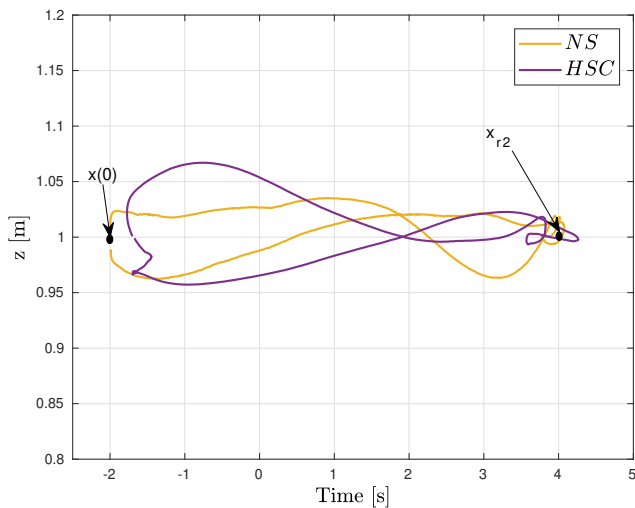


Fig. 11: Vertical position comparing the four control algorithms.



(a) Vehicle's performance evolution in the $x-z$ plane when the four control algorithms are applied in real time.



(b) xz performance for the NS and HSC control algorithms.

Fig. 12: Vehicle's performance evolution in the $x-z$ plane comparing the four control algorithms.

From the steady state regime experimental results, in Table 2 it can be observed that the Nested Saturation and Hyperbolic Saturation controllers achieve the best performance in terms of the vertical error. The HSC presents less error than NS. However, the HSC algorithm includes a bigger computational effort with respect to the other algorithms due the derivatives of the desired angle θ_d . It can be analyzed from the design of the HSC controller that it imposes desired angles, therefore, this increases the time of convergence to the desired translational values. Nevertheless, it could be an advantage because when imposing desired angles, we can ensure converge even if the desired position are far away. Observe also from tables that the Nested Saturation controller is quite small with respect to the other

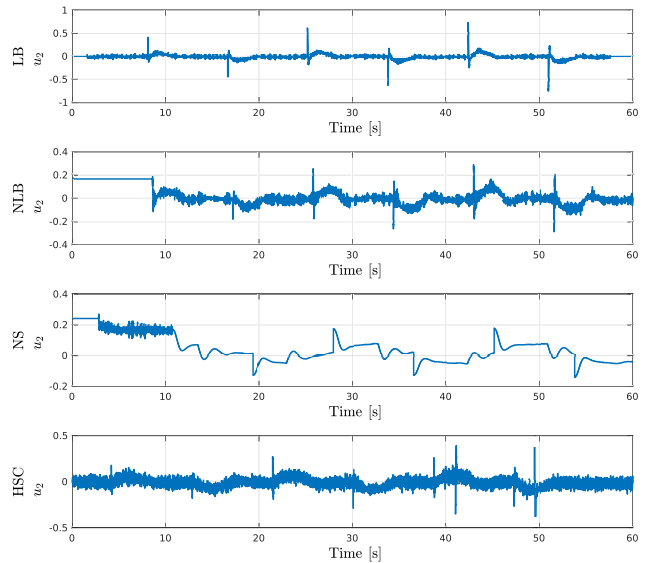


Fig. 14: Torque control response, u_2 , obtained during the flight tests.

Table 2: Performance indices of the z -axis

Controllers/Indices	IAE	ISE	ITAE	ITSE
LB	2.97	0.48	99.48	17.98
NLB	2.40	0.30	82.38	12.33
NS	1.16	0.03	35.63	1.02
HSC	1.09	0.04	30.04	1.35

approaches due mainly to the values of the bound of the saturation functions.

Table 3: Performance indices of the x -axis

Controllers/Indices	IAE	ISE	ITAE	ITSE
LB	60.21	221.56	1453	7132
NLB	64.53	241.37	1937	7780
NS	77.96	290.13	2324	10188
HSC	83.51	308.68	266.76	14752

The linear backstepping controller presents a small error in the longitudinal axis and faster convergence time as can be observed in tables. Nonetheless, according to Table 2 this control algorithm has a bigger vertical error and by consequence control effort u_2 required is also bigger.

Regarding the design, tuning and the implementation effort, the linear backstepping and nested saturation algorithms are easier to tune because their structure can be seen as Proportional Derivative controllers. In contrast, the NLB and the HSC are tedious to implement since both present several complex derivatives

and they have more parameters to tune. In this context, Table 4 presents a behavior comparison between the four algorithms including numerical and practical validation and other qualitative indicators.

Concerning the methods to design these control laws, one of the advantages of the *virtual* method is the possibility to design any control law (adaptive control, sliding mode control, etc) in F_1 and F_2 that could significantly improve the performance of the system. Here, the challenge will be to design control laws capable to provide desired angles respecting the mechanical constraints of the vehicle for avoiding singularities in the control input u_2 . The disadvantage, as previously mentioned, is the complexity of the implementation of the algorithms due of the first and second derivative of F_1 and F_2 .

Table 4: Qualitative comparison of the control algorithms. The best result is denoted with 1 and the worst with 4.

	LB	NLB	NS	HSC
Converge to the reference	1	4	3	2
Computational effort	1	3	2	3
Control effort	4	3	1	1
Design & tuning effort	1	4	2	3
Real implementation	1	4	3	3

In contrast, the *classical* method gives the advantage of an easy fashion to get the analysis and design of the control algorithms based on the assumption that the vertical axis is controlled. However, the control effort of the inputs u_1 and u_2 could be bigger if the desired positions are far from the initial conditions.

6 Conclusion

This paper deals with a brief overview evaluation of popular control algorithms for VTOLs vehicles. Our goal was to provide a unified and accessible analysis, placing the classical model of a multirotor aerial vehicle and the proposed control methods (*Classical* and *Virtual*) into a proper context. In this case, four control algorithms were obtained, explaining the main procedure for their design. The studied control methodologies were the linear and nonlinear backstepping and the nested saturation functions for the well-known underactuated dynamics and the hyperbolic saturation control for a virtual fully actuated system. The Lyapunov theory was used to prove the stability analysis.

Simulation and experimental results revealed that using the *virtual* approach for the control design results in a smoother convergence to the desired reference and a smaller control effort demand. After analysis, we consider that *virtual* approach should be a good solution for some mission and need to be explored in more detail. Nevertheless, the choice between the controllers will depend on the problem requirements.

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Compliance with Ethical Standards

Conflict of interests The authors declare that they have no conflict of interest.

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