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► **To cite this version:**

Mahmoud Hamandi, Kapil Sawant, Marco Tognon, Antonio Franchi. Omni-Plus-Seven (O 7 + ): An Omnidirectional Aerial Prototype with Minimal Uni-directional Thrusters. 2020 International Conference on Unmanned Aircraft Systems (ICUAS), Sep 2020, Athens, Greece. 10.1109/ICUAS48674.2020.9214065 . hal-02498482v3

**HAL Id: hal-02498482**

**<https://hal.science/hal-02498482v3>**

Submitted on 16 Oct 2020

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# Omni-Plus-Seven ( $O_+^7$ ): An Omnidirectional Aerial Prototype with a Minimal Number of Uni-directional Thrusters

Mahmoud Hamandi<sup>1</sup>, Kapil Sawant<sup>1</sup>, Marco Tognon<sup>2,1</sup>, Antonio Franchi<sup>3,1</sup>

**Abstract**—The aim of this paper is to present the design of a novel omnidirectional Unmanned Aerial Vehicle (UAV) with seven uni-directional thrusters, called  $O_+^7$ . The paper formally defines the  $O_+$  design for a generic number of propellers and presents its necessary conditions; then it illustrates a method to optimize the placement and orientation of the platform’s propellers to achieve a balanced  $O_+$  design. The paper then details the choice of the parameters of the  $O_+^7$  UAV, and highlights the required mechanical and electrical components. The resultant platform is tested in simulation, before being implemented as a prototype. The prototype is firstly static-bench tested to match its nominal and physical models, followed by hovering tests in multiple orientations. The presented prototype shows the ability to fly horizontally, upside down and at a tilted angle.

## I. INTRODUCTION

Unmanned Aerial Vehicles (UAVs) have been widely studied in the literature, with many applications that attempted to push their ability to the limits, such as aerial physical interaction [1], [2], surveying, photography, etc.

In its generic form, a UAV hovering (or flying) in the three-dimensional world is a rigid body able to apply forces and moments in (at most) a six-dimensional space. The applied forces and moments usually have the task to counteract external forces such as gravity, wind, etc, in addition to internal and external gyroscopic effects. The applied control wrench is generally allocated with a group of motors placed around the Center of Mass (CoM), with fixed or actuated orientations. Their placement and aerodynamic properties define the shape of the platform’s feasible wrench space.

The most common UAV in the literature is the quadrotor [3], [4], which can apply uni-directional forces and three-dimensional moments. Given its wrench space, such a vehicle has to modify its orientation to be able to move in the three-dimensional world.

To decouple the platform’s forces from its orientation, multiple designs in the literature presented platforms that exploit the full six-dimensional wrench space. [5] and [6] use tilted uni-directional thrusters to apply 3-dimensional

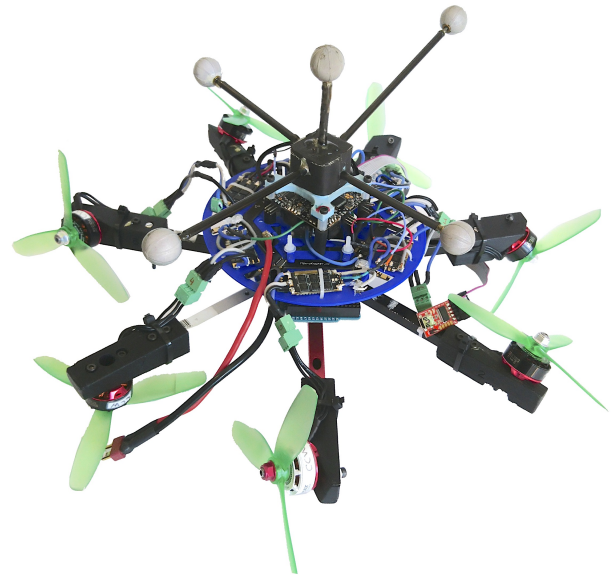


Fig. 1: The built prototype showing the motor assembly, the electronics assembly and the motion capture tracking markers.

forces independently from the applied moment. However, these platforms cannot apply forces in any direction, and are usually limited to force directions in the upper hemi-sphere.

Conversely, [7]–[9] actively tilt the propellers to achieve omnidirectional flight: [7] synchronizes the tilt angle of the propellers of a hexarotor, [8] actively tilts the angles of a quadrotor independently about their radial axes, and [9] actively tilts the propellers of a trirotor, while adding a fixed central propeller to carry the weight of the vehicle. While popular in the literature, actuated propellers add weight to the platform due to the extra actuators, which cannot guarantee instantaneous force exertion because of the time required to re-orient the propellers.

On the other hand, [10]–[12] achieve omnidirectional flight with 6 or 8 bidirectional thrusters. While their solutions are very interesting, as illustrated in [13], bidirectional thrusters exhibit a singularity near the zero thrust region. Moreover, commercial hardware solutions for bidirectional thrusters are not satisfactory, where commercial ESCs allowing the control of bidirectional propellers are scarce. Additionally, bidirectional propellers provide less thrust than their uni-directional equivalent.

In our previous work [14], we investigated the required

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This work was partially funded by the European Union’s Horizon 2020 research and innovation programme under grant agreement ID: 871479 AERIAL-CORE

properties and conditions to achieve omnidirectional flight with fixed uni-directional thrusters, and proved the number of propellers  $n \geq 7$  to be a necessary condition. Furthermore, we presented an optimization method to find propeller tilts for any generic number of propellers ( $n \geq 7$ ), that can guarantee the omnidirectional property of the platform, while enforcing equal sharing of the desired forces between the propellers.

In this work we aim to design, manufacture and test an omnidirectional platform with the minimal number ( $n = 7$ ) of uni-directional thrusters. Our design will rely on the optimization proposed in [14]. To the knowledge of the authors, this is the first manuscript in the literature showing such a working prototype with the properties mentioned above; the manufactured platform is shown in Fig. 1.

The rest of this paper is organized as follows. In section II we model a generic UAV platform. In sections III and IV we define the properties that guarantee the omnidirectionality of a UAV platform with uni-directional thrusters, and define the optimization problem with an emphasis on the assumptions made for the prototype. Section V presents the controller used to fly the platform, while section VI presents the final prototype. Sections VII-VIII show the numerical and real experiments that test the feasibility of the platform. Finally, section IX concludes the paper.

## II. MODELING

Let us define a world frame  $\mathcal{F}_W$  with origin  $O_W$  following the East-North-Up (ENU) convention with axes  $\{\mathbf{x}_W, \mathbf{y}_W, \mathbf{z}_W\}$ . Let us define a body frame  $\mathcal{F}_R$  with center  $O_R$  fixed to the geometric center of the robot (assumed to coincide with the robot's CoM), and with axes  $\{\mathbf{x}_R, \mathbf{y}_R, \mathbf{z}_R\}$ . We refer to  $\mathbf{p}_R \in \mathbb{R}^3$  as the position of  $O_R$  in  $\mathcal{F}_W$  and to  $\mathbf{R}_R \in SO(3)$ <sup>1</sup> as the orientation of  $\mathcal{F}_R$  with respect to (w.r.t.)  $\mathcal{F}_W$ . We parameterize the rotation matrix  $\mathbf{R}_R$  with the classical Euler angles roll, pitch and yaw  $(\phi, \theta, \psi)$ , such that (s.t.)  $\mathbf{R}_R = \mathbf{R}_z(\psi)\mathbf{R}_y(\theta)\mathbf{R}_x(\phi)$ . We refer to  $\mathbf{v}_R \in \mathbb{R}^3$  as the translational velocity of  $O_R$  in  $\mathcal{F}_W$ , and to  $\boldsymbol{\omega}_R \in \mathbb{R}^3$  as the angular velocity of  $\mathcal{F}_R$  w.r.t.  $\mathcal{F}_W$ , expressed into  $\mathcal{F}_R$ .

Let  $m_R \in \mathbb{R}_{>0}$  and  $\mathbf{J}_R \in \mathbb{R}_{>0}^{3 \times 3}$  define the mass and the positive definite inertia matrix of the robot w.r.t.  $\mathcal{F}_R$ . Then, following the Newton-Euler formalism, we can write the robot equations of motion as  $\dot{\mathbf{p}}_R = \mathbf{v}_R$ ,  $\dot{\mathbf{R}}_R = \mathbf{R}_R \boldsymbol{\Omega}_R$ , and

$$\begin{bmatrix} m_R \dot{\mathbf{v}}_R \\ \mathbf{J}_R \dot{\boldsymbol{\omega}}_R \end{bmatrix} = - \begin{bmatrix} g m_R \mathbf{e}_3 \\ \boldsymbol{\omega}_R \times \mathbf{J}_R \boldsymbol{\omega}_R \end{bmatrix} + \mathbf{G} \mathbf{w}, \quad (1)$$

where  $\boldsymbol{\Omega}_R = \mathcal{S}(\boldsymbol{\omega}_R)$  is the skew symmetric matrix relative to  $\boldsymbol{\omega}_R$ ,  $\mathbf{e}_3 = [0 \ 0 \ 1]^\top$ ,  $g$  is the gravitational constant, and  $\mathbf{w} \in \mathbb{R}^{6 \times 1}$  is the total wrench applied on  $O_R$  w.r.t.  $\mathcal{F}_R$ . In particular,  $\mathbf{w} = [\mathbf{f}^\top \ \mathbf{m}^\top]^\top$ , where  $\mathbf{f} \in \mathbb{R}^3$  and  $\mathbf{m} \in \mathbb{R}^3$  are the corresponding force and moment components of  $\mathbf{w}$ . Finally,  $\mathbf{G}$  is the 6-by-6 matrix of the form

$$\mathbf{G} = \begin{bmatrix} \mathbf{R}_R & \mathbf{0}_3 \\ \mathbf{0}_3 & \mathbf{I}_3 \end{bmatrix}. \quad (2)$$

<sup>1</sup> $SO(3) = \{\mathbf{R} \in \mathbb{R}^{3 \times 3} \mid \mathbf{R}^\top \mathbf{R} = \mathbf{I}_3, \det(\mathbf{R}) = 1\}$ , where  $\mathbf{I}_i \in \mathbb{R}^{i \times i}$  is the identity matrix of dimension  $i$ .

We denote by  $n$  the number of propellers, and by  $\mathbf{F} \in \mathbb{R}^{6 \times n}$  the full allocation matrix.  $\mathbf{F}_1 \in \mathbb{R}^{3 \times n}$  and  $\mathbf{F}_2 \in \mathbb{R}^{3 \times n}$  are the force and moment allocation matrices, s.t.

$$\mathbf{w} = [\mathbf{F}_1^\top \ \mathbf{F}_2^\top]^\top [u_1 \ \dots \ u_n]^\top = \mathbf{F} \mathbf{u}, \quad (3)$$

where  $u_i$  is the control thrust of the corresponding  $i$ -th propeller, and  $\mathbf{u} \in \mathbb{R}^{n \times 1}$  their concatenation. It is noted that in this formalism it was assumed that the propellers are the only source of wrench being applied on the robot, and that any other sources are neglected as secondary disturbances. Following this notation, we can write  $\mathbf{F}_1$  and  $\mathbf{F}_2$  as follows:

$$\mathbf{F}_1 = [v_1 \ \dots \ v_n], \quad (4)$$

$$\mathbf{F}_2 = [d_1 \times v_1 \ \dots \ d_n \times v_n] + [c_1 k v_1 \ \dots \ c_n k v_n], \quad (5)$$

where  $v_i \in \mathbb{R}^3$  and  $d_i \in \mathbb{R}^3$  are the thrust direction and CoM position of the  $i$ -th propeller in  $\mathcal{F}_R$ , respectively.  $c_i = -1$  ( $c_i = 1$ ) if the  $i$ -th propeller angular velocity vector has the same direction of  $v_i$  ( $-v_i$ ) when  $u_i > 0$ , i.e., the propeller spins counter-clockwise (clockwise);  $k \in \mathbb{R}$  is the drag to lift ratio of each propeller, where we assumed all propellers to be identical.

## III. OPTIMUM OMNIPLUS

From the previous section, we can define a fixed propeller aerial vehicle design as the tuple  $\mathcal{T} = (n, m_R, \mathbf{v}, \mathbf{d}, \mathbf{c}, k, u_{min}, u_{max})$  representing the number of propellers  $n$ , the platform mass  $m_R$ , the thrust direction and position of each propeller in  $\mathcal{F}_R - \mathbf{v}$  and  $\mathbf{d}$ , respectively, the rotation direction of the corresponding propellers  $\mathbf{c}$ , the aerodynamic drag to lift coefficient  $k$ , and minimum and maximum thrust of each propeller,  $u_{min}$  and  $u_{max}$ , where  $0 \leq u_{min} < u_{max}$ .

While  $u_{min}$  and  $u_{max}$  are not shown in the previous formalism, they are crucial for any platform design to guarantee the feasibility of a desired wrench  $\mathbf{w}_d \in \mathbb{W}$ , where  $\mathbb{W}$  is the set of desired wrenches necessary for platform flight. As such, we can write the following condition:

$$\forall \mathbf{w}_d \in \mathbb{W} \quad \exists \mathbf{u}_d \text{ s.t. } \mathbf{w}_d = \mathbf{F} \mathbf{u}_d \text{ and } \mathbf{u}_d \in \mathcal{U}, \quad (6)$$

where  $\mathcal{U}$  is the set of allowable control thrust, defined as the  $n$ -dimensional hypercube s.t.  $\mathcal{U} = \times_{i=1}^n [u_{min}, u_{max}]$ .

We denote with  $\mathbf{1}$  the column vector with all ones. Its size is understood from the context. Given two vectors  $\mathbf{x}$  and  $\mathbf{y}$ , the notations  $\mathbf{x} \geq \mathbf{y}$ ,  $\mathbf{x} > \mathbf{y}$  are intended component-wise.

**Definition 1.** *a design tuple  $\mathcal{T}$  is said to be OmniPlus  $O_+$  if one of the following equivalent conditions holds [14]*

$$\forall \mathbf{w} \in \mathbb{R}^6 \quad \exists \mathbf{u} \geq u_{min} \mathbf{1} \text{ s.t. } \mathbf{F} \mathbf{u} = \mathbf{w} \quad (7)$$

$$\forall \mathbf{w} \in \mathbb{R}^6 \quad \exists \mathbf{u} \geq \mathbf{0} \text{ s.t. } \mathbf{F} \mathbf{u} = \mathbf{w} \quad (8)$$

$$\text{rank}(\mathbf{F}) = 6, \quad \exists \mathbf{b} = [b_1 \ \dots \ b_n]^\top > \mathbf{0} \text{ s.t. } \mathbf{F} \mathbf{b} = \mathbf{0} \quad (9)$$

### A. Allocation Strategy

Given an  $O_+$  design and a desired wrench, following *condition* (9), one may calculate the thrust  $\mathbf{u}^*$  s.t.  $\mathbf{u}^* = \mathbf{F}^\dagger \mathbf{w}_d$  where  $\mathbf{F}^\dagger$  is the *Moore-Penrose* pseudo inverse of  $\mathbf{F}$ . As was proven in [14],  $\mathbf{u}^*$  always has at least a negative entry, and as such violates *condition* (7).

Let us consider an ellipsoid that represents the attainable wrench space  $S_w = \{\mathbf{w} \in \mathbb{R}^6 | \mathbf{w}^\top \Sigma \mathbf{w} \leq 1\} \subset \mathbb{R}^6$ , where  $\Sigma \in \mathbb{R}^{6 \times 6}$  is a positive definite matrix. The set  $U_w$  maps  $S_w$  through the linear transformation  $\mathbf{F}$  such that  $U_w = \{\mathbf{u} \in \mathbb{R}^n | \mathbf{w} = \mathbf{F}\mathbf{u}, \forall \mathbf{w} \in S_w\}$ .  $U_w$  maps  $S_w$  one to one, however, as stated in the previous paragraph, not all solutions  $\mathbf{u}^* \in U_w$  have all positive entries, and as such  $U_w \not\subset \mathcal{U}$ .

Let's define a vector  $\mathbf{b} \in \text{null}(\mathbf{F}) \cap \mathbb{R}_+^n$ , then  $\mathbf{b} \perp \mathbf{u}^*$ . Any solution such that  $\mathbf{u}^{**} = \mathbf{u}^* + \lambda \mathbf{b}$  with  $\lambda > 0 \in \mathbb{R}$  satisfies  $\mathbf{w}_d = \mathbf{F}\mathbf{u}^{**}$ . As such, the objective of the allocation strategy would be to find  $\lambda$  such that  $\mathbf{u}^{**} \in \mathcal{U}$  as follows:

$$\lambda = \underset{\mathbf{u}^{**} \in \mathcal{U}}{\text{argmin}} \|\mathbf{u}^{**}(\mathbf{u}^*, \mathbf{b})\| = \underset{\mathbf{u}^{**} \in \mathcal{U}}{\text{argmin}} \|\mathbf{u}^* + \lambda \mathbf{b}\|. \quad (10)$$

The control thrust found in (10) satisfies  $\mathbf{u}^{**} \in U_w^*$ , where  $U_w^* = U_w \cap \mathcal{U}$ . It is noted that  $U_w^*$  also maps  $S_w$  one to one, and as such, in what follows we refer to  $\mathbf{u}^{**}(\mathbf{w})$  as the control thrust in  $U_w^*$  that allows the platform to apply wrench  $\mathbf{w}$ .

**Definition 2.** *an  $O_+$  design is said to be optimal if its space  $U_w^*$  has minimum eccentricity, and if its propellers equally share the effort to keep  $\mathbf{u}^{**} \in \mathcal{U}$  calculated in (10).*

Minimizing the eccentricity of  $U_w^*$  allows the platform to apply lower maximum thrust for each desired wrench since the platform will be sharing the load equally among its propellers; this problem can be solved by minimizing the condition number of  $\Sigma^{-1}\mathbf{F}$ . On the other hand, to satisfy the second condition of *Definition 2*, it is easy to be convinced that the best choice to have  $\mathbf{b} = \mathbf{1}$ . For more details we refer the reader to [14].

## IV. PARAMETER OPTIMIZATION

In this section we detail the choice of parameters that allow the design  $\mathcal{T}$  to satisfy the conditions and requirements mentioned above. First, we make the following assumptions:

- platform dimensions are chosen separately and fixed throughout the optimization,
- all motors and propellers used in the platform are identical,
- motor and propeller choice is made separately from this optimization problem,
- propeller rotation directions are chosen prior to the optimization, with these directions alternating between one propeller and the next, i.e.  $c_i = (-1)^i$  for  $i = 1 \dots n$ .

With these assumptions, we can clearly see that the *eterovectoring* part of  $\mathcal{T}(n, m_R, \mathbf{d}, \mathbf{c}, k, u_{min}, u_{max})$  is fixed, in addition to the norm of the vectoring part  $\|\mathbf{v}_i\|$  for  $i = 1 \dots n$ , while the optimization problem should choose the direction of the vectoring part ( $\mathbf{v}$ ). It is noted that  $\|\mathbf{v}_i\| = 1$  as the

allocation matrix  $\mathbf{F}$  is assumed to map wrench  $\mathbf{w}$  to propeller thrust  $\mathbf{u}$ .

To highlight the optimization problem, let us rewrite  $\mathbf{F}_1$  and  $\mathbf{F}_2$  as follows:

$$\mathbf{F}_1 = [\mathbf{I}_3 \mathbf{v}_1 \dots \mathbf{I}_3 \mathbf{v}_n] \quad (11)$$

$$\mathbf{F}_2 = [(S(\mathbf{d}_1) + c_1 k \mathbf{I}_3) \mathbf{v}_1 \dots (S(\mathbf{d}_n) + c_n k \mathbf{I}_3) \mathbf{v}_n]. \quad (12)$$

Then we can rewrite the second part of (9) as:

$$\underbrace{\begin{bmatrix} \mathbf{I}_3 \mathbf{b}_1 & \dots & \mathbf{I}_3 \mathbf{b}_n \\ (S(\mathbf{d}_1) + c_1 k \mathbf{I}_3) \mathbf{b}_1 & \dots & (S(\mathbf{d}_n) + c_n k \mathbf{I}_3) \mathbf{b}_n \end{bmatrix}}_{\mathbf{A}(n, \mathbf{d}, \mathbf{c}, k, \mathbf{b})} \mathbf{v} = \mathbf{0}, \quad (13)$$

where  $\mathbf{I}_i$  is the  $i$ -by- $i$  identity matrix.

Following this formalism, the  $O_+$  parameter optimization can be written as follows:

$$\min \text{cond}(\Sigma^{-1}\mathbf{F}) \quad (14)$$

subject to the following constraints

$$\mathbf{v}^\top \mathbf{D}_1 \mathbf{v} = 1, \dots, \mathbf{v}^\top \mathbf{D}_n \mathbf{v} = 1 \quad (15)$$

$$\text{rank}(\mathbf{F}(\mathbf{c}, k, \mathbf{d}, \mathbf{v})) = 6 \quad (16)$$

$$\mathbf{A}(n, \mathbf{d}, \mathbf{c}, k, \mathbf{1}) \mathbf{v} = \mathbf{0} \quad (17)$$

where  $\mathbf{D}_i = \text{diag}(\mathbf{D}_{i1} \dots \mathbf{D}_{in})$  is a  $3n$ -by- $3n$  diagonal matrix, with  $\mathbf{D}_{ij} = 0$  if  $j \neq i$  and  $\mathbf{D}_{ii} = \mathbf{I}_3$  otherwise.

We note that in the choice of the propeller placements, we chose all propellers to be coplanar, placed in a star shape with the first propeller arm along  $\mathbf{x}_R$ , i.e.  $\mathbf{d}_1 = [d, 0, 0]^\top$ , and  $\mathbf{d}_i = d_1 \mathbf{R}_z(2\pi(i-1)/n)$  for  $i = 2 \dots n$ , where  $d$  is the norm of the arm connecting  $O_R$  to the CoM of any propeller, and  $\mathbf{R}_z$  is the transformation matrix corresponding to the rotation about  $\mathbf{z}_R$ .

As the aim of this paper is to build an omnidirectional platform, it is desired that the force and moment ellipsoids to resemble a sphere, where the platform will be invariant to its flight direction. As such we choose  $\Sigma$  of the form

$$\Sigma = \begin{bmatrix} \sigma_f \mathbf{I}_3 & \mathbf{0}_{3 \times 3} \\ \mathbf{0}_{3 \times 3} & \sigma_m \mathbf{I}_3 \end{bmatrix} \quad (18)$$

where  $\sigma_f, \sigma_m \in \mathbb{R}_{>0}$ .

Finally, we analyze the maximum propeller thrust  $u_{max}^d$  while the platform is in hover, where we define

**Definition 3.** *hovering (or static hovering) as the ability of the platform to stabilize its position and orientation for some orientation  $\mathbf{R}_d \in SO(3)$  with zero linear and angular velocity, i.e.  $(\mathbf{p}_R^d, \mathbf{R}_d, \mathbf{v}_R^d, \boldsymbol{\omega}_R^d) = (\mathbf{p}_R^d, \mathbf{R}_d, \mathbf{0}, \mathbf{0})$ .*

Hovering is of particular interest for the design as it is a base point for the platform to apply forces and moments in any direction. The analysis of  $u_{max}^d$  is required for the motor choice and it is an important feature to study the feasibility of the design.

The chosen  $\Sigma$ , and the ensuing minimization of  $\text{cond}(\Sigma^{-1}\mathbf{F})$ , enforces a uniformity in the platform's generated force (moment) about its corresponding directions,

and as such guarantees invariance of the platform to  $\mathbf{R}$  at hovering.

While hovering, the thrust of each propeller can be calculated as

$$\mathbf{u}_d(\mathbf{R}_d) = \mathbf{u}_d^{**} \left( \begin{bmatrix} m_R g \mathbf{R}_d e_3 \\ \mathbf{0}_3 \end{bmatrix} \right) \quad (19)$$

Due to the invariance of the platform to its hovering direction and to the chosen optimization constraints, it is straight forward to prove that

$$\mathbf{u}_{max}^d = \max \mathbf{u}_d(\mathbf{R}_d) = \max \mathbf{u}_d(\mathbf{I}), \quad (20)$$

where the right hand part of the equality is for hovering at an identity rotation matrix. While in theory maximum propeller thrust should be the same irrespective of the hovering orientation, in practice the condition number never reaches unity, and as such there is always a difference between the  $\max \mathbf{u}_d(\mathbf{R}_d)$  at different  $\mathbf{R}_d$ , and as such,  $\mathbf{u}_{max}^d$  is found with a grid search algorithm over possible orientations.

## V. CONTROLLER

Given a desired position and orientation  $\mathbf{p}_R^d(t)$  and  $\mathbf{R}_d(t)$ , the control strategy is straight forward as the allocation matrix  $\mathbf{F}$  is full rank. The desired wrench is calculated as the one that brings the platform to the desired position and orientation (along with their corresponding derivatives) while compensating for gravity and gyroscopic moments. As such the desired wrench can be written as follows:

$$\mathbf{w}^d = \mathbf{G}^{-1} \begin{bmatrix} m_R(g\mathbf{e}_3 + \dot{\mathbf{v}}_R^d) + \mathbf{K}_P \mathbf{e}_p + \mathbf{K}_D \dot{\mathbf{e}}_p + \mathbf{K}_{IP} \int_0^t \mathbf{e}_p \\ \boldsymbol{\omega}_R \times \mathbf{J}_R \boldsymbol{\omega}_R + \mathbf{K}_R \mathbf{e}_R + \mathbf{K}_\omega \mathbf{e}_\omega + \mathbf{K}_{IR} \int_0^t \mathbf{e}_R \end{bmatrix} \quad (21)$$

where  $\mathbf{K}_P, \mathbf{K}_D, \mathbf{K}_{IP}, \mathbf{K}_R, \mathbf{K}_\omega$  and  $\mathbf{K}_{IR}$  are diagonal positive definite matrices  $\in \mathbb{R}^{3 \times 3}$  representing the controller tunable gains.  $\mathbf{e}_p = \mathbf{p}_R^d - \mathbf{p}_R$ ,  $\mathbf{e}_\omega = \boldsymbol{\omega}_R^d - \boldsymbol{\omega}_R$  and  $\mathbf{e}_R = 1/2(\mathbf{R}_d^\top \mathbf{R} - \mathbf{R}^\top \mathbf{R}_d)^\wedge$ , where  $[\cdot]^\wedge$  is the inverse skew symmetric operator. Then for each desired wrench, a control thrust is calculated as described earlier in (10).

### A. State Estimation

The platform is endowed with an IMU that captures the platform's specific linear acceleration and angular velocity. Furthermore, its position and orientation are tracked with a motion capture system.

All measurements from the IMU are filtered with the regression-based filter introduced in [15]. The filter is designed to reduce the noise caused by the propellers' vibration; however, as the motors are controlled in open-loop (*i.e.* propeller rotational velocities are not measured), the filter fit the second order polynomial to the IMU signal without separation between its signal and noise constituents.

Both filtered IMU measurements and motion capture measurements are fused using an Unscented Kalman Filter (UKF) [16] to retrieve the full pose estimate of the platform.

Component	weight per unit [g]	# units	weight [g]
Motors	40	7	280
Propellers	5	7	35
Electronics	200	-	200
Mechanical parts	350	-	350
Battery	214	-	214
Total			1079

TABLE I: Representing the estimated weight of each of the platform components prior to its final design and construction. It should be noted that the weight of parts that are certainly used were reported as they are, while others are estimated.

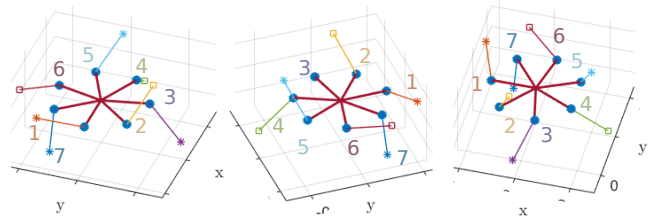


Fig. 2: Optimized propeller direction of the  $O_+^7$  design, with  $n = 7$ ,  $d = 0.16$  [m] and  $k = 0.002$  [m] at  $\text{cond}(\Sigma^{-1} \mathbf{F}) = 2.052$ .

## VI. PROTOTYPE

For our prototype we chose to construct a platform with  $n = 7$ , the least number of uni-directional propellers necessary to achieve omnidirectional thrust. The platform is built to be the smallest possible to increase its stability by reducing any possible oscillations in the arms connecting the motors to the body; as such, we chose an arm length  $d = 0.16$  [m]. We then chose 5" propellers, as it is the largest diameter that can be installed on the platform without any collision between adjacent propellers. The propellers we chose enforced a drag to lift coefficient  $k = 0.002$  [m], and a lift coefficient  $k_f = 0.5e - 4$  [N/Hz<sup>2</sup>]. Finally, the wrench ellipsoid was chosen such that  $\Sigma = \text{diag}([1, 1, 1, 0.5, 0.5, 0.5])$ . We estimated the platform mass before the platform construction to be around 1.1 [kg] following the component-wise weight estimation shown in Tab. I.

### A. Numerical Optimization

The  $O_+$  optimization algorithm calculated the vectoring part of the design, and reached a minimum condition number of  $\text{cond}(\Sigma^{-1} \mathbf{F}) = 2.052$ , with a vectoring part as follows:

$$\mathbf{v} = \begin{bmatrix} 0.36 & -0.35 & 0.29 & -0.81 & -0.37 & 0.78 & 0.10 \\ -0.90 & 0.44 & 0.76 & -0.12 & 0.45 & -0.57 & -0.07 \\ 0.25 & 0.83 & -0.58 & -0.58 & 0.81 & 0.26 & -0.99 \end{bmatrix}. \quad (22)$$

The above thrust directions are illustrated in Fig. 2.

With the current parameters, the maximum propeller thrust was found to be  $\mathbf{u}_{max}^d = 11.18$  [N], corresponding to a maximum rotational speed of  $W_{max} = 472$  [Hz] for the chosen propellers; as such we chose a motor that can provide a

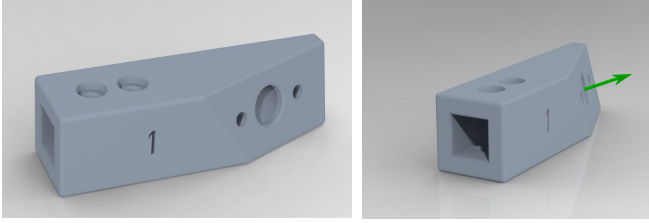


Fig. 3: CAD design of the motor adapter for motor # 1. The adapter is designed to fasten the motor to the body frame at position  $d_1$  and orientation  $v_1$  in  $\mathcal{F}_R$ .

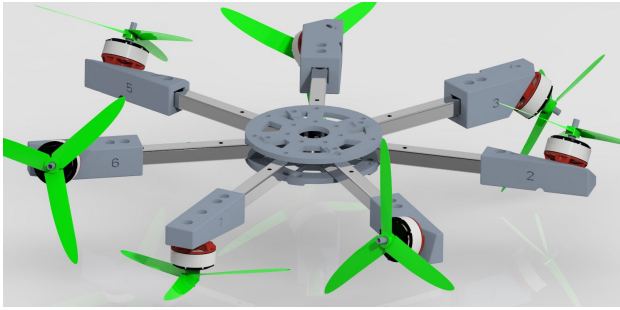


Fig. 4: CAD design showing the platform body assembly. The body assembly shows the motors and propellers connected to the body frame with aluminum bars, where each motor was fastened to the corresponding bar with a CAD designed adapter similar to the one shown in fig. 3. Each of the shown adapters was designed to fit the corresponding orientation  $v_i$  in  $\mathcal{F}_R$ .

peak thrust of 14 [N] with the chosen propellers, at which the motor is required to rotate at 530 [Hz].

The chosen motor is controlled in PWM via an Electronic Speed Controller (ESC), *i.e.*, the motor is controlled in open-loop, where the thrust generated at each PWM was identified with a force-torque sensor. Finally, the PWMs delivered to the ESCs are generated using an onboard microcontroller.

### B. Platform Implementation

The platform body is constructed with 7 aluminum bars connecting the CoM of the platform to the CoM of the propellers. Each propeller is connected with a separate arm to ensure the stability of the platform. Aluminum bars are fastened together using 3D printed plates connected to one of their edges, while the second edge is connected to a 3D printed adapter that ensures the motors' connection at the calculated direction. Fig. 3 shows the CAD drawing of one of the adapters, while Fig. 4 shows the CAD drawing of the body frame assembly.

Finally, the necessary electronics and motion capture markers are placed on top of the platform, with the full prototype shown in Fig. 1.

The final weight of this setup without a battery is measured at  $m_R = 0.835$  [kg].

### C. Design Drawbacks

The platform prototype as presented above exhibits the following drawbacks that can affect its performance:

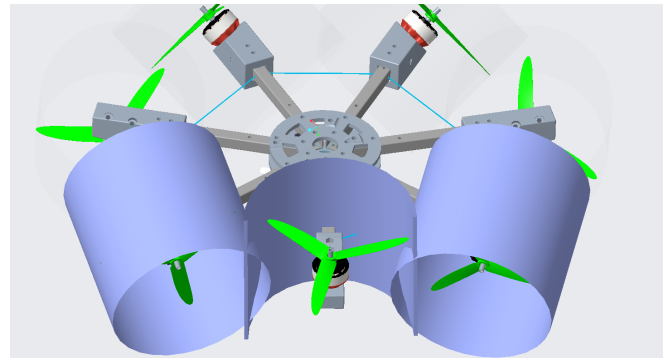


Fig. 5: Example intersection between adjacent propeller airflow cylinders, showing the intersection between 7<sup>th</sup> propeller airflow cylinder with the airflow cylinders of the 1<sup>st</sup> and 6<sup>th</sup> propeller.

- Propeller open-loop control: as the propellers are controlled in open-loop, the propeller speed is not guaranteed, and correspondingly the individual motor thrust. Therefore, the applied wrench can differ from the desired one.
- Propeller airflow cylinder intersection. We define the airflow cylinder as the cylinder containing the corresponding propeller, of radius equal to the propellers', and direction similar to the corresponding propeller. Since the propellers are placed close to each other, with each producing thrust in *any* direction, it is impossible for the airflow cylinders of adjacent propellers not to intersect as shown in Fig. 5. This intersection affects their aerodynamics, as propellers have to withdraw air from the inflow/outflow of their adjacent propellers instead of withdrawing air from the free stream assumed static. It should be noted that it is difficult to estimate the effect of this intersection, as the change in the thrust produced by each propeller will depend on the amount of thrust provided by the adjacent propeller.

While these drawbacks can induce an error in the applied wrench, we assume it equivalent to an external disturbance that can be compensated by the feedback controller shown in section V.

## VII. PRELIMINARY TESTS

### A. Dynamic Simulation

To assess the performance and flyability of the prototype described in section VI, we simulated its dynamical system in Matlab/Simulink with the estimated mass. The simulation is made closer to reality with the addition of measurement noise and signal delays.

Figure 6 shows the performance of the platform's flight with  $z_R$  circling the unit radius sphere. This figure shows that the platform is able to fly while in a variety of orientations. The simulation also shows that the platform can apply independent force and moments, and as such, orient  $\mathcal{F}_R$  independently of its translation, while the propeller rotational velocities are kept within the allowable range.

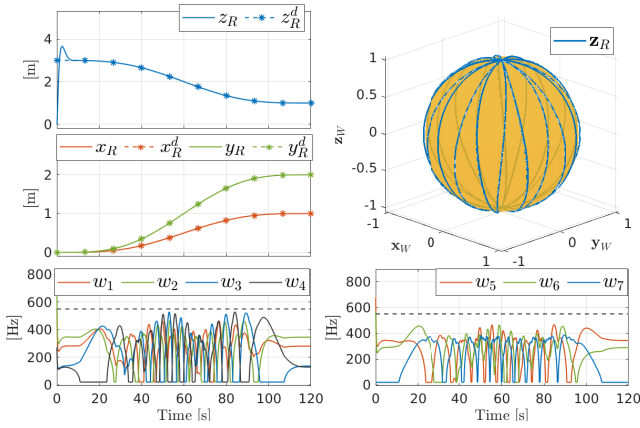


Fig. 6: Tracking results of the simulated platform while following a desired position and orientation. The platform orientation is chosen such that  $z_R$  circles the unit sphere multiple times, while the position is chosen to change smoothly and simultaneously on all axes. The desired and actual  $z_R$  are superimposed in the top right plot. A step change in the  $z$ -position is required at time  $t = 0s$ , where the platform is required to lift from the height of 0 [m] to 4 [m]. The figure also shows the propeller rotational velocities  $w_i$  for  $i = 1 \dots 7$ , where the dashed line constitutes the limiting maximum velocity.

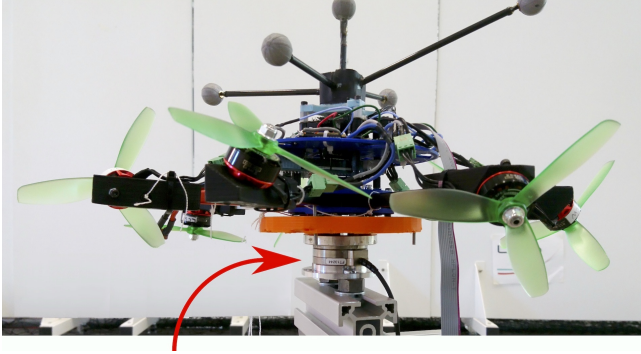


Fig. 7: Platform fixed to the force-torque sensor.

### B. Wrench Tests

To assess the discrepancy between the ideal model and the built prototype, a force-torque sensor was used to measure the generated wrench in a static experiment as shown in Fig. 7. Tab. II shows the force and moment constituents of the desired nominal wrench  $w_d$  and the corresponding error between the measured and nominal wrench  $w_{error} = w_{measured} - w_d$ .

We can observe from the data in Tab. II an error between the measured and nominal values of the applied wrench; it can also be observed that the value of this error changes depending on the desired nominal wrench. While we could not identify clearly the cause of these errors, they were expected due to the motor speed control and aerodynamic interaction between adjacent propellers, in addition to manufacturing imperfections (see section VI-C).

test	$f_d$ [N]	$f_{error}$ [N]	$m_d$ [N.m]	$m_{error}$ [N.m]
1	[008]	[0.4 +0.0 -0.2]	[000]	[-0.1 +0.1 +0.1]
2	[428]	[0.0 -0.1 -0.3]	[000]	[-0.1 +0.3 -0.2]
3	[048]	[0.2 -0.3 -1.1]	[000]	[-0.3 +0.1 +0.0]
4	[008]	[0.6 -0.1 -0.6]	[0.200]	[-0.1 +0.1 +0.1]
5	[008]	[0.3 -0.3 -0.4]	[00.20]	[-0.0 -0.1 +0.1]
6	[008]	[0.3 -0.1 -0.4]	[000.2]	[-0.0 +0.1 -0.0]

TABLE II: Representing the nominally applied force  $f_d$  and moment  $m_d$ , and their respective measured error.

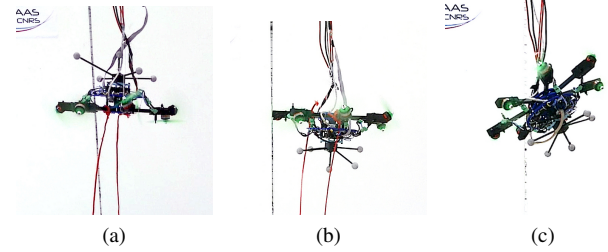


Fig. 8: Preliminary hovering tests of the prototype platform: a) platform hovering horizontally, b) platform hovering upside down, c) platform hovering at a tilted orientation such that  $\phi_d = 130^\circ$ .

## VIII. EXPERIMENTS

### A. System Setup

As stated in section V-A, the platform is endowed with an IMU, which exports the raw specific linear acceleration and angular velocity measurements at 1 [kHz]; in addition, the platform position and orientation are tracked with a motion capture system at 100 [Hz]. Both measurements are fused by a UKF running at 1 [kHz], and providing an estimate of the platform state. The platform controller is implemented in Matlab/Simulink at 500 [Hz], while the onboard micro-controller delivers the desired PWM to the ESCs. Most software (excluding the controller), including those used for the communication between Matlab and the platform, are developed in C++ using Genom3 [17], a code generator and formal software component description language that allows assembling middleware-independent components in a modular system. These software can be found here: <https://git.openrobots.org/projects/telekyb3>

The platform is top-connected to a power supply cable, in addition to multiple data cables allowing the back and forth communication with the off-board controller PC.

### B. Hovering

To preliminarily test the omnidirectional flight ability of the platform, we ask the vehicle to lift off from its hanged position and hover in place in multiple orientations as shown in Fig. 8. The performance of the platform in each of the desired orientations is shown in Figures 9 through 11. These figures show that the platform is able to hover horizontally, upside down, and at a tilted angle. Furthermore, for these orientations, the desired propeller rotational velocities are within the acceptable range.

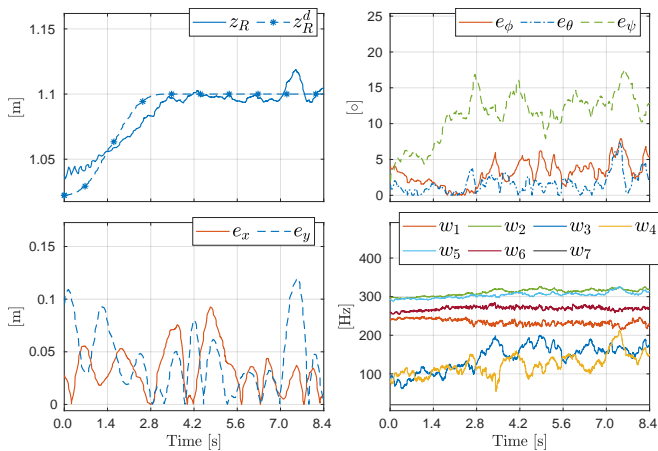


Fig. 9: Performance of the platform while hovering horizontally: (top-left) shows the desired and estimated platform height, (top-right) shows the angular errors, (bottom-left) shows the average  $xy$  error, and (bottom-right) shows the propeller desired rotational velocities.

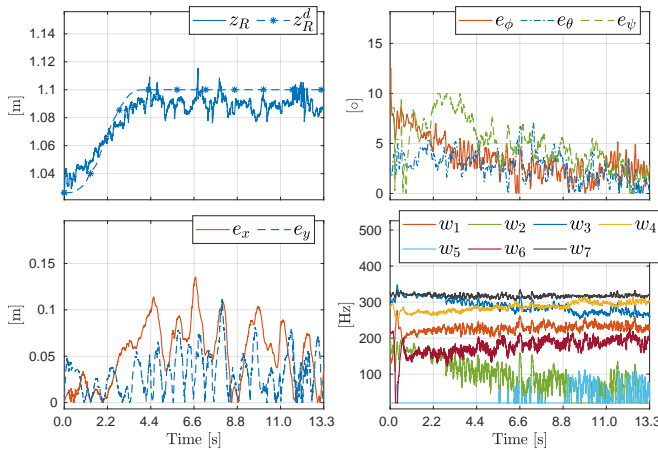


Fig. 10: Performance of the platform while hovering with  $z_R$  pointing in the negative  $z_W$  direction, *i.e.*,  $\phi_d = 180^\circ$ : (top-left) shows the desired and estimated platform height, (top-right) shows the angular errors, (bottom-left) shows the average  $xy$  error, and (bottom-right) shows the propeller desired rotational velocities.

*Remark 1.* In all these experiments, the propeller rotational velocities are within the acceptable range. Moreover, the maximum propeller rotational velocity is around 380 [Hz], far from the maximum allowed rotational velocity of 550 [Hz].

*Remark 2.* While the platform is able to hover at these orientations, there is a noticeable tracking error in all experiments even if the controller gains have been tuned specifically for each experiment to reduce tracking error while assuring the platform's stability.

## IX. CONCLUSIONS

In this paper we have presented a novel prototype of an  $O_+$  UAV with 7 propellers. To the knowledge of the authors, the design is the first of its kind. We conducted an experimental campaign to assess the model and the prototype. Our experiments show that the platform is able to hover in many directions. From our tests, we have realized that there

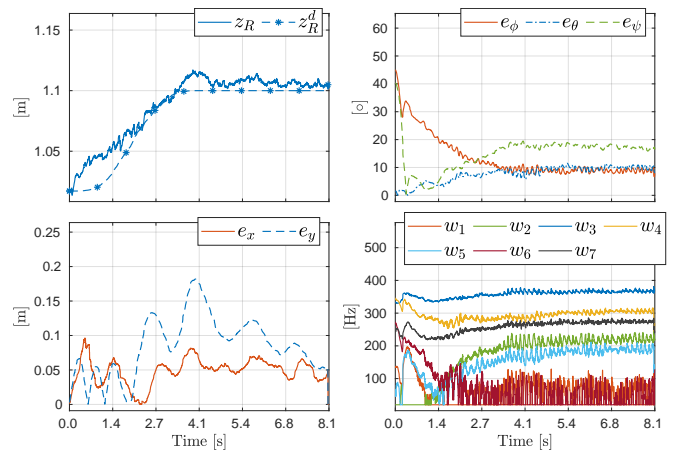


Fig. 11: Performance of the platform while hovering at a tilted orientation such that  $\phi_d = 130^\circ$ ,  $\theta_d = 0^\circ$ ,  $\psi_d = 90^\circ$ : (top-left) shows the desired and estimated platform height, (top-right) shows the angular errors, (bottom-left) shows the average  $xy$  error, and (bottom-right) shows the propeller desired rotational velocities. The platform starts its maneuver while being oriented near upside down, *i.e.*  $\phi(t = 0s) \approx 180^\circ$ .

is a discrepancy between the nominal desired wrench and the one actually applied by the platform. This discrepancy is expected to be caused by : 1) the open-loop control of the propeller rotational velocities, 2) and the aerodynamic interactions between adjacent propellers due to the small size of the frame. While the aerodynamic interactions are caused by the proximity between the propellers, the open-loop control of the propellers is necessary to ensure such high speed rotational velocity of the propellers, where off the shelf ESCs providing closed loop speed control at the required rotational speed are still rare.

In the future, we aim at ameliorating the design along the following lines

- Reduce the weight of the platform by replacing aluminum bars with lighter carbon fiber material. This will allow the platform to fly with lower propeller rotational velocities, and as such be able to achieve close-loop control of the propellers.
- Increase the platform's arm length to reduce the aerodynamic interactions between propellers.
- Implement an adaptive/learning based controller that can compensate for the unmodeled aerodynamic interactions.

## REFERENCES

- [1] A. Ollero, G. Heredia, A. Franchi, G. Antonelli, K. Kondak, A. Sanfeliu, A. Viguria, J. R. Martinez-de Dios, F. Pierri, J. Cortés, A. Santamaria-Navarro, M. A. Trujillo, R. Balachandran, J. Andrade-Cetto, and A. Rodriguez, "The AEROARMS project: Aerial robots with advanced manipulation capabilities for inspection and maintenance," *IEEE Robotics & Automation Magazine, Special Issue on Floating-base (Aerial and Underwater) Manipulation*, vol. 25, no. 4, pp. 12–23, 2018.
- [2] A. Ollero, "Aerial robotics cooperative assembly system (arcas): First results," in *Aerial Physically Acting Robots (AIRPHARO) Workshop, IROS*, 2012, pp. 7–12.



- [3] R. Mahony, V. Kumar, and P. Corke, "Multirotor Aerial Vehicles: Modeling, Estimation, and Control of Quadrotor," *IEEE Robotics & Automation Magazine*, vol. 19, no. 3, pp. 20–32, 2012.
- [4] M. Faessler, A. Franchi, and D. Scaramuzza, "Differential flatness of quadrotor dynamics subject to rotor drag for accurate tracking of high-speed trajectories," *IEEE Robotics and Automation Letters*, vol. 3, no. 2, pp. 620–626, 2018.
- [5] M. Ryll, G. Muscio, F. Pierri, E. Cataldi, G. Antonelli, F. Caccavale, D. Bicego, and A. Franchi, "6D interaction control with aerial robots: The flying end-effector paradigm," *The International Journal of Robotics Research*, vol. 38, no. 9, pp. 1045–1062, 2019.
- [6] H. Romero, S. Salazar, A. Sanchez, and R. Lozano, "A new UAV configuration having eight rotors: dynamical model and real-time control," in *46th IEEE Conf. on Decision and Control*, New Orleans, LA, Dec. 2007, pp. 6418–6423.
- [7] M. Ryll, D. Bicego, and A. Franchi, "Modeling and control of FAST-Hex: a fully-actuated by synchronized-tilting hexarotor," in *2016 IEEE/RSJ Int. Conf. on Intelligent Robots and Systems*, Daejeon, South Korea, Oct. 2016, pp. 1689–1694.
- [8] M. Ryll, H. H. Bühlhoff, and P. Robuffo Giordano, "A novel over-actuated quadrotor unmanned aerial vehicle: modeling, control, and experimental validation," *IEEE Trans. on Control Systems Technology*, vol. 23, no. 2, pp. 540–556, 2015.
- [9] Y. Long and D. J. Cappelleri, *Omnicopter: A Novel Overactuated Micro Aerial Vehicle*. Heidelberg: Springer International Publishing, 2013, pp. 215–226.
- [10] D. Brescianini and R. D'Andrea, "Design, modeling and control of an omni-directional aerial vehicle," in *2016 IEEE Int. Conf. on Robotics and Automation*, Stockholm, Sweden, May 2016, pp. 3261–3266.
- [11] S. Park, J. J. Her, J. Kim, and D. Lee, "Design, modeling and control of omni-directional aerial robot," in *2016 IEEE/RSJ Int. Conf. on Intelligent Robots and Systems*, Daejeon, South Korea, 2016, pp. 1570–1575.
- [12] E. Dyer, S. Sirouspour, and M. Jafarinasab, "Energy optimal control allocation in a redundantly actuated omnidirectional uav," in *2019 IEEE Int. Conf. on Robotics and Automation*, Montreal, Canada, may 2019.
- [13] M. Hamandi, F. Usai, Q. Sablé, N. Staub, M. Tognon, and A. Franchi, "Survey on Aerial Multirotor Design: a Taxonomy Based on Input Allocation," Jan. 2020, working paper or preprint. [Online]. Available: <https://hal.archives-ouvertes.fr/hal-02433405>
- [14] M. Tognon and A. Franchi, "Omnidirectional aerial vehicles with unidirectional thrusters: Theory, optimal design, and control," *IEEE Robotics and Automation Letters*, vol. 3, no. 3, pp. 2277–2282, 2018.
- [15] M. Hamandi, M. Tognon, and A. Franchi, "Direct acceleration feedback control of quadrotor aerial vehicles," *2020 IEEE Int. Conf. on Robotics and Automation*.
- [16] E. A. Wan and R. Van Der Merwe, "The unscented kalman filter for nonlinear estimation," in *Proceedings of the IEEE 2000 Adaptive Systems for Signal Processing, Communications, and Control Symposium (Cat. No. 00EX373)*. Ieee, 2000, pp. 153–158.
- [17] A. Mallet, C. Pasteur, M. Herrb, S. Lemaignan, and F. Ingrand, "Genom3: Building middleware-independent robotic components," *2010 IEEE International Conference on Robotics and Automation*, pp. 4627–4632, 2010.