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A low-diffusion self-adaptive flux-vector splitting approach for compressible flows

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Abstract

A low-diffusion self-adaptive flux-vector splitting method is presented for the Euler equations. The flux-vector is here split into convective and acoustic parts following the formulation recently proposed by the authors. This procedure is based on the Zha-Bilgen (or previously Baraille *et al.* for the Euler barotropic system) approach enriched by a dynamic flow-dependent splitting parameter based on the local Mach number. As a consequence, in the present self-adaptive splitting, the convective and acoustic parts decouple in the low-Mach number regime whereas the complete Euler equations are considered for the sonic and highly subsonic regimes. The low diffusive property of the present scheme is obtained by adding anti-diffusion terms to the momentum and the energy components of the pressure flux in the acoustic part of the present splitting. This treatment results from a formal invariance principle preserving the discrete incompressible phase space through the pressure operator. Numerical results for several carefully chosen one- and two-dimensional test problems are finally investigated to demonstrate the accuracy and robustness of the proposed scheme for a wide variety of configurations from subsonic to highly subsonic flows.

Keywords: Euler equations, Flux-Vector Splitting, Low-Mach number flows, Stationary incompressible flows, Operator kernel

1. Introduction

The objective of the present work is to provide a time-explicit numerical scheme able to capture fast acoustic waves related to the *compressible* Euler system while remaining accurate, when the flow Mach number tends towards zero, with stationary solutions of the *incompressible* Euler system. Although the development of accurate methods for the Euler and Navier-Stokes equations is the focus of numerous research studies over the last few decades, fulfilling both criteria is still challenging. Indeed, upwind schemes are known to be well suited for the solution of hyperbolic conservation laws, and are thus extensively used to solve the compressible Euler equations [1, 2, 3]. These approaches can broadly be divided into two main categories: Godunov-type methods and Flux-Vector Splitting (FVS) methods.

Godunov-type methods are based on the approximate solution of the local Riemann problem between two adjacent states for the calculation of the numerical flux through the interface between these two states. Complete Riemann solvers as the Roe's approximate Riemann solver [4] or the HLLC (Harten-Lax-van Leer + Contact) scheme of Toro [5, 3], respect the full wave structure of the compressible Euler equations in the construction of the numerical flux. As a consequence, these solvers are known for their accuracy and for their ability to capture both linearly degenerate and genuinely non-linear waves. Generally, this family of methods is usually complex and computationally expensive.

In Flux-Vector Splitting methods, the flux is usually split into positive and negative components as in the pioneer work of Steger and Warming [6]. These techniques are usually less expensive than Godunov-type methods and provide an efficient upwind discretisation. However their inability to resolve intermediate characteristic fields badly affects

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18 the correct resolution of contact waves, material interfaces, shear waves and vortices for example. To circumvent
19 this problem, Liou and Steffen [7] have proposed a flux-vector splitting named AUSM (Advection Upstream Splitting
20 Method) able to recognize contact waves. The basic idea of AUSM-type solvers is to split the flux-vector into its
21 convective and acoustic parts at the continuous level. Subsequently, both parts are treated separately based on their
22 respective characteristic fields. Other strategies have been proposed to separate the convective and acoustic parts
23 of the flux as the one from Zha and Bilgen [8] or more recently from Toro and Vazquez [9] and from Mandal and
24 Panwar [10]. Contact preservation is achieved by modifying the dissipation vector of the pressure flux using isentropic
25 relations [10].

26 Though upwind methods can handle the waves related to the compressible Euler system, it is known that they fail
27 to be accurate with respect to solutions of the asymptotically incompressible Euler system when the Mach number
28 tends to zero. Such a drawback has been extensively studied in the literature. The pioneering work of Guillard and
29 Viozat [11], based on a formal asymptotic decomposition of the discrete solution obtained using the Roe’s scheme,
30 has formally revealed that, a non-dimensional pressure term scaling as $O(M)$ is inherently present in the discrete so-
31 lution. In [12], Guillard and Murrone inject a Mach power expansion inside the exact solution of a Riemann problem
32 related to the isentropic Euler system. They show that the spurious pressure term of order $O(M)$ corresponds to an
33 artificial viscosity stemming from the normal momentum equation of the *acoustic* part of the system. Subsequently,
34 the presence of such a numerical diffusion has been discovered in [13] for a large number of approximate Riemann
35 solvers. Enhancing the understanding of the low Mach number issue, the work of Dellacherie [14] and Dellacherie
36 *et al.* [15] establishes a link between the above artificial acoustic viscosity and the disturbance of the discrete incom-
37 pressible phase space throughout time. Indeed, by studying the *modified equation* obtained from an exact Godunov
38 scheme related to the linear wave system, the authors prove that the presence of the acoustic viscosity in the momen-
39 tum equation “shrinks” the kernel associated with the acoustic operator. As a consequence, the latter kernel is strictly
40 included in the incompressible phase space. This means that after one time-step, initial conditions belonging to the
41 discrete incompressible space might be disturbed by the discrete acoustic operator. As highlighted in [16], this issue
42 disappears in 1-D or in multi-D if a triangular (tetrahedral) mesh is used. In order to partly solve this difficulty, many
43 works [17, 18, 15, 19, 20] propose to weaken the order of the artificial viscosity in the normal momentum equation
44 by multiplying it with a term proportional to the local Mach number. The same procedure is applied to the AUSM
45 scheme leading to the AUSM⁺-up method [21, 22]. In terms of stability, the consequences of the above artificial
46 viscosity reduction, depend on the numerical method at stake. Indeed, in [15] a linear L^2 -stability study shows that
47 Godunov-like solvers remain *a priori* stable under a classical CFL condition based on the biggest Euler eigenvalues.
48 On the contrary, the FVS-type AUSM⁺-up is binded by an extremely severe explicit stability constraint.

49 In the current contribution, the proposed numerical scheme combines the simplicity associated with the FVS
50 methods while keeping a good control on the acoustic operator discretization. Its formulation relies on a convec-
51 tive/acoustic decomposition originally proposed by Zha, Bilgen [8] and Baraille [23] for the complete Euler system
52 and by Baraille *et al.* [24] for the Euler barotropic system. The convective part is evaluated via a simple upwinding.
53 The acoustic part is discretized using a HLLC-like Riemann solver. By doing so, one can easily identify, in the nor-
54 mal contribution of the momentum flux, the artificial viscosity term responsible for the low Mach number accuracy
55 issue. Besides, a formal analysis, based on the *sufficient* criterion related to the discrete incompressible space invari-
56 ance through the application of the discrete acoustic operator is presented here. It suggests that additional low-Mach
57 corrections should be also included inside the *energy* flux of the acoustic part.

58 In order to fully distinguish the numerical treatment of the present FVS scheme in sonic flow regimes from the
59 subsonic ones, a dynamic flow-dependent splitting parameter is introduced. As described recently in [25], such a
60 splitting parameter acts as a weighting factor between the convective and the acoustic part of the flux. Its definition
61 is based on the local Mach number such as for low-Mach configurations the convective and acoustic parts of the
62 flux completely decouple whereas for sonic and highly subsonic configurations, the complete Euler equations are
63 retrieved in the convective part of the considered splitting. This makes the present numerical scheme self-adaptive as
64 the low-Mach regions can be detected in both space and time.

65 The paper is organized as follows. Section 2 is devoted to the governing equations associated to the Euler system.
66 The numerical approaches considered in the present work are then presented in Section 3, in particular the HLLC
67 scheme and the present self-adaptive splitting. Afterwards, focus is given on the construction of low Mach corrections
68 in Section 4. The classical reduction of the numerical diffusion in the momentum equation [14] is recalled and a
69 special formal analysis is carried out on the pressure flux in the acoustic part issued from the present splitting. Three

70 numerical methods, HLLC, HLLC with a standard low-Mach correction and the present FVS approach, are then
 71 assessed on several problems in Section 5. The numerical tests involve highly subsonic configurations, stationary
 72 incompressible test cases as well as simulations where fast acoustic waves and low-Mach number solutions interact.

73 2. Governing equations

The set of compressible Euler equations is here considered for inviscid compressible flows:

$$\partial_t \mathbf{U} + \nabla \cdot \mathbf{F}(\mathbf{U}) = \mathbf{0} \quad \text{with} \quad \mathbf{U} = \begin{pmatrix} \rho \\ \rho \mathbf{u} \\ \rho e \end{pmatrix} \quad \text{and} \quad \mathbf{F}(\mathbf{U}) = \begin{pmatrix} \rho \mathbf{u} \\ \rho \mathbf{u} \otimes \mathbf{u} + p \mathbf{I}_d \\ (\rho e + p) \mathbf{u} \end{pmatrix}, \quad (1)$$

74 where ρ , \mathbf{u} , p and e are respectively the density, the velocity vector, the pressure, the specific total energy and \mathbf{I}_d the
 75 identity tensor. The specific internal energy ε is given by: $\varepsilon = e - \mathbf{u}^2/2$. The set of Eq. (1) is completed by an equation
 76 of state (EOS): $\varepsilon = \varepsilon(\rho, p)$. The speed of sound denoted by c is obtained via the EOS: $c^2 \equiv (p/\rho - \partial_\rho \varepsilon)(\partial_p \varepsilon)^{-1}$. Let
 77 us mention that the formulation of the numerical scheme proposed below is independent of the EOS. However, the
 78 applications presented here only involve ideal gas EOS for which: $\rho \varepsilon = p/(\gamma - 1)$ and $c = \sqrt{\gamma p/\rho}$; γ being the heat
 79 capacity ratio.

80
 81 The model in Eq. (1) is known to be hyperbolic and admits three eigenvalues in 1-D: $\lambda_1 = u - c$, $\lambda_2 = u$ and
 82 $\lambda_3 = u + c$. The field associated with eigenvalue λ_2 is linearly degenerate (LD), other fields are genuinely non-linear
 83 (GNL).

84
 85 In the sequel, the framework related to unstructured multi-D Finite-Volume schemes is introduced. Different
 86 numerical fluxes are also described.

87 3. Numerical approach

The present scheme relies on the integral form of Eq. (1):

$$\frac{d}{dt} \int_C \mathbf{U} dV + \int_{\partial C} \mathbf{F}(\mathbf{U}) \mathbf{n} dS = \mathbf{0}, \quad (2)$$

where C is the control volume, ∂C its boundary whose unit outward normal vector is denoted by \mathbf{n} ; the cell-centered
 Finite-Volume discretisation can be written as:

$$\mathbf{U}_i^{n+1} = \mathbf{U}_i^n - \frac{\Delta t^n}{|C_i|} \sum_{j \in V(i)} \Phi_{i,j} |S_{i,j}|, \quad (3)$$

with \mathbf{U}_i^n the cell average of the conservative variables vector \mathbf{U} on the cell C_i at time t^n and $\Phi_{i,j}$ the time-explicit
 numerical flux-vector at the cell interface $S_{i,j}$ between the cell C_i and the cell C_j . $V(i)$ refers to the neighboring cells
 of C_i . Note that, in Eq. (3), $|C_i|$ stands for the volume of the cell C_i whereas $|S_{i,j}|$ is the surface of the interface between
 C_i and C_j . The time step Δt^n used in Eq. (3) is given by the Courant number C defined as:

$$C = \Delta t^n \max_i \left(\max_{j \in V(i)} \left(\frac{|\mathbf{u}_i^n \cdot \mathbf{n}_{i,j}| + c_i^n}{h_i} \right) \right), \quad (4)$$

88 with h_i the characteristic length of the cell C_i and $\mathbf{n}_{i,j}$ the outward (from C_i to C_j) unit normal vector of a given
 89 interface $S_{i,j}$. Several numerical flux-vectors are considered in the following. Without loss of generality the numerical
 90 fluxes are written using a 2-D framework.

91 *3.1. HLLC scheme*

92 Consider the one-dimensional Riemann problem related to system (1) projected on the direction pointed by the
 93 normal $\mathbf{n}_{i,j}$. Besides, define $\mathbf{t}_{i,j}$, the tangential unit vector related to $\mathbf{n}_{i,j}$. Along the normal direction, let us introduce
 94 the left and right states of the interface denoted by \mathbf{U}_L and \mathbf{U}_R respectively.

The HLLC (Harten, Lax, van-Leer + Contact) scheme [5, 3] is an approximate Riemann solver considering all the waves of the Euler equations. Its numerical flux writes as:

$$\Phi_{i,j}^{\text{HLLC}} = \begin{cases} \Phi_L & \text{if } 0 < S_L, \\ \Phi_L^* & \text{if } S_L \leq 0 < S_M, \\ \Phi_R^* & \text{if } S_M \leq 0 < S_R, \\ \Phi_R & \text{if } S_R \leq 0, \end{cases} \quad (5)$$

where $\Phi_K = \mathbf{F}(\mathbf{U}_K) \mathbf{n}_{i,j}$ with the flux-vector

$$\mathbf{F}(\mathbf{U}) \mathbf{n} = \begin{pmatrix} \rho \mathbf{u} \cdot \mathbf{n} \\ \rho \mathbf{u} \mathbf{u} \cdot \mathbf{n} + p \mathbf{n} \\ (\rho e + p) \mathbf{u} \cdot \mathbf{n} \end{pmatrix}, \quad (6)$$

and

$$\Phi_K^* = \Phi_K + S_K (\mathbf{U}_K^* - \mathbf{U}_K), \quad (7)$$

with,

$$\mathbf{U}_K^* = \rho_K \frac{(S_K - \mathbf{u}_K \cdot \mathbf{n}_{i,j})}{(S_K - S_M)} \begin{pmatrix} 1 \\ S_M \\ \mathbf{u}_K \cdot \mathbf{t}_{i,j} \\ \rho_K e_K + (S_M - \mathbf{u}_K \cdot \mathbf{n}_{i,j}) \left(S_M + \frac{p_K}{\rho_K (S_K - \mathbf{u}_K \cdot \mathbf{n}_{i,j})} \right) \end{pmatrix} \quad \text{where } K = L, R. \quad (8)$$

The pressure p_M is given by:

$$p_M = \rho_L (S_L - \mathbf{u}_L \cdot \mathbf{n}_{i,j}) (S_M - \mathbf{u}_L \cdot \mathbf{n}_{i,j}) + p_L = \rho_R (S_R - \mathbf{u}_R \cdot \mathbf{n}_{i,j}) (S_M - \mathbf{u}_R \cdot \mathbf{n}_{i,j}) + p_R, \quad (9)$$

and the speed S_M is defined as:

$$S_M = \frac{\rho_R \mathbf{u}_R \cdot \mathbf{n}_{i,j} (S_R - \mathbf{u}_R \cdot \mathbf{n}_{i,j}) - \rho_L \mathbf{u}_L \cdot \mathbf{n}_{i,j} (S_L - \mathbf{u}_L \cdot \mathbf{n}_{i,j}) + p_L - p_R}{\rho_R (S_R - \mathbf{u}_R \cdot \mathbf{n}_{i,j}) - \rho_L (S_L - \mathbf{u}_L \cdot \mathbf{n}_{i,j})}. \quad (10)$$

The speed S_L and S_R corresponding to the fastest waves at each side of the interface are computed as proposed in Batten *et al.* [26]:

$$S_L = \min(\mathbf{u}_L \cdot \mathbf{n}_{i,j} - c_L, \hat{\mathbf{u}} \cdot \mathbf{n}_{i,j} - \hat{c}) \quad \text{and} \quad S_R = \max(\mathbf{u}_R \cdot \mathbf{n}_{i,j} + c_R, \hat{\mathbf{u}} \cdot \mathbf{n}_{i,j} + \hat{c}), \quad (11)$$

with $\hat{\mathbf{u}}$ and \hat{c} the Roe average [4] of the velocity \mathbf{u} and speed of sound c variables defined as:

$$\hat{f} = \frac{\sqrt{\rho_L} f_L + \sqrt{\rho_R} f_R}{\sqrt{\rho_L} + \sqrt{\rho_R}} \quad (12)$$

95 This numerical scheme was previously used in two-dimensional Euler equations [5], as well as in steady supersonic
 96 flow regimes [27]. More recently, it has been applied to fast transient two-phase flows [28, 29].

97 3.2. Flux-vector splitting approach

Based on the work of Iampietro *et al.* [25, 30], the flux-vector is split into a convective part and a pressure part as:

$$\mathbf{F}(\mathbf{U})\mathbf{n} = \mathbf{C}(\mathbf{U})\mathbf{n} + (1 - \mathcal{E}_0^2)\mathbf{P}(\mathbf{U})\mathbf{n} \quad \text{with} \quad \mathbf{C}(\mathbf{U})\mathbf{n} = \mathbf{U}\mathbf{u} \cdot \mathbf{n} + \mathcal{E}_0^2\mathbf{P}(\mathbf{U})\mathbf{n} \quad \text{and} \quad \mathbf{P}(\mathbf{U})\mathbf{n} = \begin{pmatrix} 0 \\ p\mathbf{n} \\ \rho\mathbf{u} \cdot \mathbf{n} \end{pmatrix}, \quad (13)$$

where \mathcal{E}_0 is a dynamic flow-dependent splitting parameter which is detailed in the following. Note that the specific choice $\mathcal{E}_0 = 0$ corresponds to the splitting proposed by Baraille *et al.* in [24] for the Euler barotropic system and by Zha and Bilgen in [8] for the whole Euler system. Both convective and acoustic subsystems issued from the present splitting have been theoretically studied in [25]. The pressure part of the flux-vector is evaluated based on the acoustic solver [3]. Its expression is similar to the one of a HLLC-type solver related to the system:

$$\partial_t \mathbf{U} + \nabla \cdot \mathbf{P}(\mathbf{U}) = \mathbf{0}. \quad (14)$$

It simply writes:

$$\mathbf{\Pi}_{i,j} = \begin{pmatrix} 0 \\ p^*\mathbf{n}_{i,j} \\ p^*u^* \end{pmatrix} \quad \text{with} \quad \begin{cases} p^* = \frac{\mathcal{S}_L p_R + \mathcal{S}_R p_L - \mathcal{S}_R \mathcal{S}_L (\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j})}{\mathcal{S}_R + \mathcal{S}_L}, \\ u^* = \frac{\mathcal{S}_R \mathbf{u}_R \cdot \mathbf{n}_{i,j} + \mathcal{S}_L \mathbf{u}_L \cdot \mathbf{n}_{i,j} - (p_R - p_L)}{\mathcal{S}_R + \mathcal{S}_L}, \end{cases} \quad (15)$$

with for $K = L, R$, the acoustic impedance \mathcal{S}_K being defined as:

$$\mathcal{S}_K = \rho_K c_K. \quad (16)$$

In Figure 1, the wave structure as well as the intermediate states associated with the HLLC approximate solver of

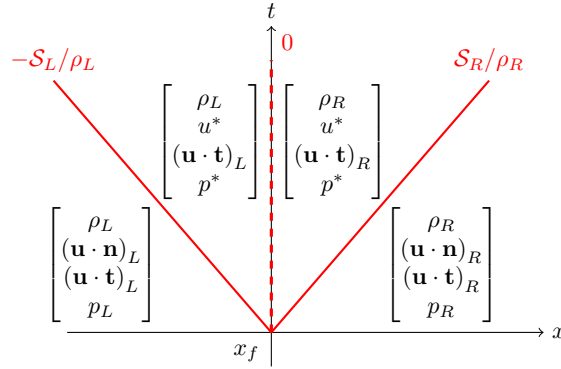


Figure 1: HLLC approximate Riemann solver for the pressure system: waves and states.

98 system (14) are displayed. Let us mention that, once again, \mathbf{t} stands for the tangential unit vector orthogonal to \mathbf{n} in a
99 2-D unstructured mesh.
100

The convective part of the flux-vector is then evaluated using the following upwinding:

$$\mathbf{\Gamma}_{i,j} = \frac{1}{2} [u^* \mathbf{W}_R + u^* \mathbf{W}_L - |u^*| (\mathbf{W}_R - \mathbf{W}_L)] + \mathcal{E}_0^2 \begin{pmatrix} 0 \\ p^* \mathbf{n}_{i,j} \\ 0 \end{pmatrix}, \quad (17)$$

with for $K = L, R$,

$$\mathbf{W}_K = \begin{pmatrix} \rho_K \\ (\rho\mathbf{u})_K \\ (\rho\ell)_K + \mathcal{E}_0^2 p_K \end{pmatrix}. \quad (18)$$

As it can be noticed, the upwinding leans on the intermediate velocity u^* in the acoustic subsystem and is applied to each flux term driven by $\mathbf{u} \cdot \mathbf{n}$. As for the residual $\mathcal{E}_0^2 \nabla p$ of the pressure gradient in the momentum equation, it is discretized using the intermediate acoustic pressure p^* . Finally, the numerical flux-vector is given by:

$$\Phi_{i,j}^{\text{FVS}} = \Gamma_{i,j} + (1 - \mathcal{E}_0^2) \Pi_{i,j}, \quad (19)$$

where

$$\mathcal{E}_0 = \max(\mathcal{E}_{\text{inf}}, \min(1, \tilde{M})), \quad 0 < \mathcal{E}_{\text{inf}} \ll 1. \quad (20)$$

Here, \tilde{M} stands for the local Mach number built using the HLLC material velocity, S_M , at cell interface $S_{i,j}$ given in Eq. (10). For example, in 2-D, one can consider v^* such that:

$$v^* = \begin{cases} \mathbf{u}_L \cdot \mathbf{t}_{i,j} & \text{if } 0 < S_M, \\ \mathbf{u}_R \cdot \mathbf{t}_{i,j} & \text{if } 0 \geq S_M. \end{cases} \quad (21)$$

Introducing the local speed of sound at the interface:

$$\tilde{c} = \max(c_L, c_R), \quad (22)$$

the local Mach number is then defined as:

$$\tilde{M} = \sqrt{(S_M)^2 + (v^*)^2} / \tilde{c}. \quad (23)$$

If one considers the underlying convective and pressure sub-systems associated with the flux decomposition (13), it is known [24] that when $\mathcal{E}_0 = 0$, the convective sub-system

$$\partial_t \mathbf{U} + \nabla \cdot \mathbf{C}(\mathbf{U}) = \mathbf{0}, \quad (24)$$

101 is no longer hyperbolic. Hence, in order to prevent issues in zero material-velocity regions, the strictly positive lower
102 bound \mathcal{E}_{inf} has been introduced in the definition of the parameter \mathcal{E}_0 . In the following, \mathcal{E}_{inf} is taken equal to 10^{-8} .

103
104 After having completely defined the present numerical scheme, let us stress that the choice of p^* to discretize
105 the pressure gradient residual contribution in the momentum flux of $\mathbf{C}(\mathbf{U}) \mathbf{n}$ is crucial. Indeed, when the local Mach
106 number of the flow goes over one, the parameter \mathcal{E}_0 tends towards one and the pressure part $(1 - \mathcal{E}_0^2) \mathbf{P}(\mathbf{U}) \mathbf{n}$ of the
107 splitting cancels out. Hence, in the momentum equation, the contribution related to the pressure gradient exclusively
108 stems from the convective part of the flux. The non-centered part of p^* then improves the stability of the present
109 approach when highly compressible shock waves propagate through the flow.

110
111 Besides when the local Mach number tends towards zero, \mathcal{E}_0 also tends towards zero and the splitting completely
112 decouples convection from acoustic wave production. In such a low Mach number regime, the ability of approximate
113 Riemann solvers to be accurate with respect to the solutions of the incompressible Euler system requires to apply
114 *low-Mach corrections*. In the following, this issue is briefly presented and low Mach corrections met in many works
115 of the literature are described. Then, the low Mach correction strategy of the present approach is detailed.

116 4. Low-Mach correction

117 4.1. Low-Mach correction based on the work of Dellacherie et al.

118 In the very complete work of Dellacherie et al. [15], the authors study the behavior of Godunov-like schemes
119 applied to a linear wave equation obtained from the Euler barotropic system. They prove that, for 2-D cartesian
120 meshes, the kernel of the acoustic operator associated with the linear wave equation is strictly smaller than the discrete
121 space related to incompressible solutions. As a consequence, after one time-step, initial solutions that belong to this
122 discrete incompressible space might be distorted by the acoustic operator and acoustic modes might grow.

123 The authors then show that such a lack of accuracy is caused by the numerical diffusion of the non-centered part
124 of the momentum flux scaling as $O(\Delta x/M)$. In addition, a simple *All-Mach* number cure is proposed. In the subsonic

125 case, it simply consists in reducing the prohibitive momentum numerical diffusion by multiplying the non-centered
 126 part of the momentum flux by a term denoted by θ proportional to the local flow Mach number. Subsequently we will
 127 refer to this term as “low-Mach correction”.

Following the work of Dellacherie *et al.* [15], the HLLC scheme described previously is modified for low-Mach number considerations as:

$$\Phi_{i,j}^{\text{HLLC-Corr}} = \Phi_{i,j}^{\text{HLLC}} + (1 - \theta_{i,j}) \frac{\tilde{c}}{2} \begin{pmatrix} 0 \\ (\rho_R \mathbf{u}_R \cdot \mathbf{n}_{i,j} - \rho_L \mathbf{u}_L \cdot \mathbf{n}_{i,j}) \mathbf{n}_{i,j} \\ 0 \end{pmatrix}, \quad (25)$$

with,

$$\theta_{i,j} = \min(1, \tilde{M}). \quad (26)$$

128 Let us mention that similar low-Mach corrections holding on the normal component related to the interface $S_{i,j}$ of the
 129 momentum equation have been extensively used in many works of the literature [17, 18, 31].

130 In the next subsection, a formal analysis inspired from discrete space invariance arguments already presented in
 131 [16, 15] for the linear wave equation is performed. Such an analysis provides elements of justification for the different
 132 low-Mach corrections applied to the non-linear acoustic operator of the present approach. In the framework of low
 133 Mach number *stationary* flows, the need for an additional low-Mach correction in the energy flux is notably studied.
 134

135 4.2. Low-Mach correction of the present approach

Consider a two-dimensional regular cartesian mesh made of $N_x \times N_y = N$ cells of length Δx . A part of such a mesh is displayed in Figure 2. Let us also introduce \mathcal{I}_N , the discrete space of *non-dimensional* solutions such that:

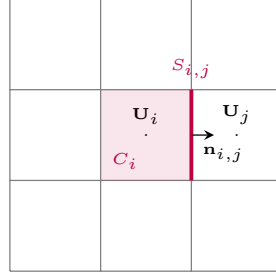


Figure 2: Two-dimensional regular cartesian mesh

$$\mathcal{I}_N = \left\{ \mathbf{V}_i = \begin{bmatrix} \rho \\ \mathbf{u} \\ p \end{bmatrix}_i, i \in [1, N], \text{ such that } : \begin{array}{l} \sum_{j \in V(i)} \frac{\mathbf{u}_i + \mathbf{u}_j}{2} \cdot \mathbf{S}_{i,j} = 0, \\ p_i = p_i^{(0),(1)} + O(M^2), \sum_{j \in V(i)} \frac{p_i^{(0),(1)} + p_j^{(0),(1)}}{2} \mathbf{S}_{i,j} = \mathbf{0}. \end{array} \right\}, \quad (27)$$

with $\mathbf{S}_{i,j} = |S_{i,j}| \mathbf{n}_{i,j}$. Considering the above cartesian mesh, space \mathcal{I}_N is a first-order discrete approximation of the incompressible space characterized by a divergence-free velocity field and a zero pressure gradient at zeroth and first order with respect to the Mach number. One can underline that, in [16], a more specific definition based on the Hodge decomposition of the above discrete incompressible space is given. Let us now introduce the kernel of the *dimensional* discrete acoustic operator $(1 - \mathcal{E}_0^2) \mathbf{P}(\mathbf{U}) \mathbf{n}$ detailed in equations (13) and (15):

$$\{\text{Ker } \mathbf{P}\}_N = \left\{ \mathbf{V}_i = \begin{bmatrix} \rho \\ \mathbf{u} \\ p \end{bmatrix}_i, i \in [1, N], \text{ such that } : \begin{array}{l} \sum_{j \in V(i)} (1 - (\mathcal{E}_0)_{i,j}^2) p_{i,j}^{*\theta} \mathbf{S}_{i,j} = \mathbf{0}, \\ \sum_{j \in V(i)} (1 - (\mathcal{E}_0)_{i,j}^2) p_{i,j}^{*\theta} \mathbf{u}_{i,j}^{*\beta} |S_{i,j}| = 0 \end{array} \right\}. \quad (28)$$

with,

$$\begin{cases} (\mathcal{E}_0)_{i,j} &= \max(\mathcal{E}_{\text{inf}}, \min(1, \tilde{M}_{i,j})), \\ p_{i,j}^{*,\theta} &= \frac{\mathcal{S}_L p_R + \mathcal{S}_R p_L - \theta_{i,j} \mathcal{S}_R \mathcal{S}_L (\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j})}{\mathcal{S}_R + \mathcal{S}_L}, \\ \mathbf{u}_{i,j}^{*,\beta} &= \frac{\mathcal{S}_R \mathbf{u}_R \cdot \mathbf{n}_{i,j} + \mathcal{S}_L \mathbf{u}_L \cdot \mathbf{n}_{i,j} - \beta_{i,j} (p_R - p_L)}{\mathcal{S}_R + \mathcal{S}_L}. \end{cases} \quad (29)$$

It can be noticed that the intermediate pressure and velocity formulas proposed in (29) are identical to the these written in (15) if and only if $(\theta_{i,j}, \beta_{i,j}) = (1, 1)$. These coefficients can be assimilated to potential low-Mach corrections acting on the non-centered part of the momentum and the energy fluxes of the acoustic operator. Indeed, in the case of the acoustic contribution to the momentum flux, one can write

$$p_{i,j}^{*,\theta} = p_{i,j}^* + (1 - \theta_{i,j}) \frac{\mathcal{S}_R \mathcal{S}_L}{\mathcal{S}_R + \mathcal{S}_L} (\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j}). \quad (30)$$

136 The second term is then similar to the θ -correction of Dellacherie *et al.* [15] presented in Eq. (25).

137

In the sequel, by performing a non-dimensional Mach number expansion on the equalities given in the discrete acoustic kernel sub-space defined in (28), one tries to derive sufficient conditions on the scaling of the couple $(\theta_{i,j}, \beta_{i,j})$ for which

$$\mathcal{I}_N \subset \{\text{Ker } \mathbf{P}\}_N, \quad (31)$$

is guaranteed. Inclusion (31) formally states that the discrete acoustic operator will not perturb initial discrete incompressible solutions after one time-step. For the sake of simplicity, let us introduce some notations:

$$\begin{aligned} \omega_{i,j}^+ &= \frac{\mathcal{S}_R}{\mathcal{S}_R + \mathcal{S}_L}, \quad \omega_{i,j}^- = \frac{\mathcal{S}_L}{\mathcal{S}_R + \mathcal{S}_L}, \\ \mathcal{D}_{i,j}^p &= \frac{\mathcal{S}_R \mathcal{S}_L}{\mathcal{S}_R + \mathcal{S}_L}, \quad \mathcal{D}_{i,j}^u = \frac{1}{\mathcal{S}_R + \mathcal{S}_L}. \end{aligned} \quad (32)$$

The pressure $p^{*,\theta}$ and the velocity $\mathbf{u}^{*,\beta}$ can thus be rewritten as

$$\begin{cases} p^{*,\theta} &= \frac{p_R + p_L}{2} - (\omega_{i,j}^+ - 1/2)(p_R - p_L) - \theta_{i,j} \mathcal{D}_{i,j}^p (\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j}), \\ \mathbf{u}^{*,\beta} &= \frac{\mathbf{u}_R \cdot \mathbf{n}_{i,j} + \mathbf{u}_L \cdot \mathbf{n}_{i,j}}{2} - (\omega_{i,j}^- - 1/2)(\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j}) - \beta_{i,j} \mathcal{D}_{i,j}^u (p_R - p_L). \end{cases} \quad (33)$$

138 In order to exhibit the scaling sufficient conditions contributing to inclusion (31), one first needs to derive the non-
139 dimensional version of $\{\text{Ker } \mathbf{P}\}_N$, and decompose its constitutive relations in powers of the Mach number. Such a work
140 is performed in the two following paragraphs in which the momentum and the energy discrete equalities are treated
141 separately.

142 4.2.1. Correction arising from the discrete momentum equation

In the sequel, for the sake of simplicity, the non-dimensional variables are written the same way that the dimensional variables. In definition (28) of the discrete acoustic kernel, the non-dimensional equality related to the momentum equation with $\mathbf{U}_L = \mathbf{U}_i$ and $\mathbf{U}_R = \mathbf{U}_j$ writes:

$$\sum_{j \in V(i)} (1 - M^2 (\mathcal{E}_0)_{i,j}^2) \left[\frac{1}{M^2} \left\{ \frac{p_j + p_i}{2} - (\omega_{i,j}^+ - 1/2)(p_j - p_i) \right\} - \frac{\theta_{i,j}}{M} \mathcal{D}_{i,j}^p (\mathbf{u}_j \cdot \mathbf{n}_{i,j} - \mathbf{u}_i \cdot \mathbf{n}_{i,j}) \right] \mathbf{S}_{i,j} = \mathbf{0}. \quad (34)$$

Recall that in equality (34), each non-dimensional term is *a priori* of order one and can be decomposed in Mach powers:

$$\forall \phi \in \{\rho, \mathbf{u}, p\}, \quad \phi = \phi^{(0)} + M \phi^{(1)} + M^2 \phi^{(2)} + O(M^3). \quad (35)$$

Assuming that equality (34) implies that each contribution of the different Mach powers is null, one can write that for the $1/M^2$ and the $1/M$ terms:

$$\sum_{j \in V(i)} \left\{ \frac{p_j^{(0)} + p_i^{(0)}}{2} - ((\omega_{i,j}^+)^{(0)} - 1/2)(p_j^{(0)} - p_i^{(0)}) \right\} \mathbf{S}_{i,j} = \mathbf{0}, \quad (36a)$$

$$\sum_{j \in V(i)} \left\{ \frac{p_j^{(1)} + p_i^{(1)}}{2} - ((\omega_{i,j}^+)^{(0)} - 1/2)(p_j^{(1)} - p_i^{(1)}) - (\omega_{i,j}^+)^{(1)}(p_j^{(0)} - p_i^{(0)}) - \theta_{i,j}^{(0)} (\mathcal{D}_{i,j}^p)^{(0)} (\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} - \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}) \right\} \mathbf{S}_{i,j} = \mathbf{0}. \quad (36b)$$

Hence, if one imposes $\forall (i, j)$

$$\omega_{i,j}^+ = \omega_{i,j}^- = \frac{1}{2}, \quad (37a)$$

$$\theta_{i,j}^{(0)} = 0, \quad (37b)$$

a discrete solution belonging to \mathcal{I}_N satisfies automatically relations (36). Let us underline that there is no need to impose condition (37a) if $\forall i$

$$p_i = p^{(0),(1)}(t) + O(M^2), \quad (38)$$

with $p^{(0),(1)}$ a time-dependent constant. However, along the calculation, there might exist discrete solutions fulfilling

$$\sum_{j \in V(i)} \frac{p_i^{(0),(1)} + p_j^{(0),(1)}}{2} \mathbf{S}_{i,j} = \mathbf{0}, \quad (39)$$

without guaranteeing that $\forall (i, j)$, $p_i^{(0),(1)} = p_j^{(0),(1)}$. These discrete solutions are called *checkerboard modes* and their appearance and amplification are studied in the framework of a linear wave equation in [32]. Condition (37a) then contributes to the checkerboard modes invariance (and thus their non-amplification) through the momentum equation of the discrete acoustic operator. Let us notice that condition (37a) holds by taking for $K = L, R$,

$$\mathcal{S}_K = \mathcal{S} = \max(\rho_L c_L, \rho_R c_R). \quad (40)$$

143 Equality (40) corresponds exactly to the subcharacteristic condition proposed in [31, 33] involved in the acoustic
144 subsystem discretization of a Lagrange-Projection method. Condition (37b) is more classical and has been already
145 exhibited by the same kind of calculations based on a two-dimensional Roe scheme in [11, 17].

146

147

In the following paragraph, equality (40) is fulfilled.

148

4.2.2. Correction arising from the discrete energy equation

Under equality (40), the intermediate pressure and velocity now simply write:

$$\begin{cases} p^{*,\theta} &= \frac{p_R + p_L}{2} - \theta_{i,j} \frac{\mathcal{S}(\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j})}{2}, \\ u^{*,\beta} &= \frac{\mathbf{u}_R \cdot \mathbf{n}_{i,j} + \mathbf{u}_L \cdot \mathbf{n}_{i,j}}{2} - \beta_{i,j} \frac{p_R - p_L}{2\mathcal{S}}. \end{cases} \quad (41)$$

Let us mention that, in the specific case where $(\theta_{i,j}, \beta_{i,j}) = (1, 1)$, relations (41) are identical to these obtained in [31, 33] in the context of Lagrange-Projection methods. In this section, attention is paid to the non-dimensional equality related to the energy equation in the definition of the discrete acoustic kernel (28). Such an equality reads:

$$\sum_{j \in V(i)} (1 - M^2 (\mathcal{E}_0)_{i,j}^2) \left[\frac{p_j + p_i}{2} - M \theta_{i,j} \frac{\mathcal{S}_{i,j}}{2} (\mathbf{u}_j \cdot \mathbf{n}_{i,j} - \mathbf{u}_i \cdot \mathbf{n}_{i,j}) \right] \times \left[\frac{\mathbf{u}_j \cdot \mathbf{n}_{i,j} + \mathbf{u}_i \cdot \mathbf{n}_{i,j}}{2} - \frac{\beta_{i,j}}{M} \frac{1}{2\mathcal{S}_{i,j}} (p_j - p_i) \right] |\mathbf{S}_{i,j}| = 0. \quad (42)$$

149 After having decomposed equality (42) in powers of the Mach number, one obtains:

Order $1/M$:

$$\sum_{j \in V(i)} \left[\frac{p_j^{(0)} + p_i^{(0)}}{2} \right] \times \left[\beta_{i,j}^{(0)} \frac{1}{2 \mathcal{S}_{i,j}^{(0)}} (p_j^{(0)} - p_i^{(0)}) \right] |S_{i,j}| = 0. \quad (43)$$

Order 1:

$$\begin{aligned} & \sum_{j \in V(i)} \left[\frac{p_j^{(1)} + p_i^{(1)}}{2} - \theta_{i,j}^{(0)} \frac{\mathcal{S}_{i,j}^{(0)}}{2} (\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} - \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}) \right] \times \left[-\beta_{i,j}^{(0)} \frac{1}{2 \mathcal{S}_{i,j}^{(0)}} (p_j^{(0)} - p_i^{(0)}) \right] |S_{i,j}| \\ & + \sum_{j \in V(i)} \left[\frac{p_j^{(0)} + p_i^{(0)}}{2} \right] \times \left[\frac{\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} + \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}}{2} - \left(\beta_{i,j} \frac{1}{2 \mathcal{S}_{i,j}} (p_j - p_i) \right)^{(1)} \right] |S_{i,j}| = 0. \end{aligned} \quad (44)$$

Order M :

$$\begin{aligned} & \sum_{j \in V(i)} -(\mathcal{E}_0)_{i,j}^2 \left[\frac{p_j^{(0)} + p_i^{(0)}}{2} \right] \times \left[-\beta_{i,j}^{(0)} \frac{1}{2 \mathcal{S}_{i,j}^{(0)}} (p_j^{(0)} - p_i^{(0)}) \right] |S_{i,j}| \\ & + \sum_{j \in V(i)} \left[\frac{p_j^{(2)} + p_i^{(2)}}{2} - \left(\theta_{i,j} \frac{\mathcal{S}_{i,j}}{2} (\mathbf{u}_j \cdot \mathbf{n}_{i,j} - \mathbf{u}_i \cdot \mathbf{n}_{i,j}) \right)^{(1)} \right] \times \left[-\beta_{i,j}^{(0)} \frac{1}{2 \mathcal{S}_{i,j}^{(0)}} (p_j^{(0)} - p_i^{(0)}) \right] |S_{i,j}| \\ & + \sum_{j \in V(i)} \left[\frac{p_j^{(1)} + p_i^{(1)}}{2} - \theta_{i,j}^{(0)} \frac{\mathcal{S}_{i,j}^{(0)}}{2} (\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} - \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}) \right] \times \left[\frac{\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} + \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}}{2} - \left(\beta_{i,j} \frac{1}{2 \mathcal{S}_{i,j}} (p_j - p_i) \right)^{(1)} \right] |S_{i,j}| \\ & + \sum_{j \in V(i)} \left[\frac{p_j^{(0)} + p_i^{(0)}}{2} \right] \times \left[\frac{\mathbf{u}_j^{(1)} \cdot \mathbf{n}_{i,j} + \mathbf{u}_i^{(1)} \cdot \mathbf{n}_{i,j}}{2} - \left(\beta_{i,j} \frac{1}{2 \mathcal{S}_{i,j}} (p_j - p_i) \right)^{(2)} \right] |S_{i,j}| = 0. \end{aligned} \quad (45)$$

Consider a discrete solution that belongs to \mathcal{I}_N . Assume additionally that the pressure decomposition (38) holds. As a consequence one can observe that equality (43) is automatically fulfilled. Besides, equality (44) becomes:

$$p^{(0),(1)}(t) \sum_{j \in V(i)} \frac{\mathbf{u}_j^{(0)} + \mathbf{u}_i^{(0)}}{2} \cdot \mathcal{S}_{i,j} = 0. \quad (46)$$

Equality (46) holds automatically as it corresponds exactly to the discrete divergence-free constraint which partly defines \mathcal{I}_N . Note that, if the fluid is endowed with an ideal gas equation of state (EOS) such that

$$\rho \varepsilon = \frac{p}{\gamma - 1}, \quad \rho c^2 = \gamma p, \quad (47)$$

with γ the specific heat ratio, the discrete terms

$$\sum_{j \in V(i)} \left[\frac{p_j^{(0)} + p_i^{(0)}}{2} \right] \times \left[\frac{\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} + \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}}{2} \right] |S_{i,j}|, \quad (48)$$

in equality (44), are consistent (up the γ factor) with

$$(\rho c^2)^{(0)} \nabla \cdot \mathbf{u}^{(0)}, \quad (49)$$

which intervenes in the zeroth-order continuous asymptotic pressure equation related to the Euler system:

$$\partial_t p^{(0)} + \mathbf{u}^{(0)} \cdot \nabla p^{(0)} + (\rho c^2)^{(0)} \nabla \cdot \mathbf{u}^{(0)} = 0. \quad (50)$$

Besides, the most intricate equality (45) simplifies as:

$$\begin{aligned} & \sum_{j \in V(i)} \left[p^{(0),(1)}(t) - \theta_{i,j}^{(0)} \frac{\mathcal{S}_{i,j}^{(0)}}{2} (\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} - \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}) \right] \times \left[\frac{\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} + \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}}{2} \right] |S_{i,j}| \\ & + \sum_{j \in V(i)} \left[p^{(0),(1)}(t) \right] \times \left[\frac{\mathbf{u}_j^{(1)} \cdot \mathbf{n}_{i,j} + \mathbf{u}_i^{(1)} \cdot \mathbf{n}_{i,j}}{2} - \beta_{i,j}^{(0)} \frac{1}{2\mathcal{S}_{i,j}^{(0)}} (p_j^{(2)} - p_i^{(2)}) \right] |S_{i,j}| = 0. \end{aligned} \quad (51)$$

Using the discrete divergence-free relation, equality (46) reduces to

$$\begin{aligned} & \sum_{j \in V(i)} -\theta_{i,j}^{(0)} \frac{\mathcal{S}_{i,j}^{(0)}}{2} (\mathbf{u}_j^{(0)} \cdot \mathbf{n}_{i,j} - \mathbf{u}_i^{(0)} \cdot \mathbf{n}_{i,j}) \times \left[\frac{\mathbf{u}_j^{(0)} + \mathbf{u}_i^{(0)}}{2} \right] \cdot \mathbf{S}_{i,j} \\ & - p^{(0),(1)}(t) \sum_{j \in V(i)} \beta_{i,j}^{(0)} \frac{1}{2\mathcal{S}_{i,j}^{(0)}} (p_j^{(2)} - p_i^{(2)}) |S_{i,j}| = 0. \end{aligned} \quad (52)$$

One can observe that the remaining terms in equality (52) appear only because of the diffusive part of the intermediate pressure and velocity definition (41). As a consequence, by imposing that $\forall (i, j)$

$$\theta_{i,j}^{(0)} = 0, \quad (53a)$$

$$\beta_{i,j}^{(0)} = 0, \quad (53b)$$

150 one ensures that equality (52) holds. The above formal analysis suggests that, in the context of *stationary* discrete
 151 solutions, low-Mach corrections applied to the non-centered part of the intermediate pressure and velocity of the
 152 energy flux ensure that, up to the order $O(M^2)$, the discrete acoustic operator does not perturb discrete incompressible
 153 solutions. Conversely, if $(\theta_{i,j}^{(0)}, \beta_{i,j}^{(0)}) \neq (0, 0)$, equality (52) does not necessarily hold.

154 In the case of a fluid endowed with an ideal gas EOS, the non-dimensional energy equation can be formally read,
 155 after having discarded the kinetic energy terms scaling as $O(M^2)$, as a non-dimensional pressure equation. Then,
 156 in the context of stationary flows, the energy flux contributions associated with equality (52) *might* distort an in-
 157 compressible discrete solution by creating a non-dimensional pressure of order $O(M)$. Let us mention that in the
 158 references [11, 17] that focus on the same topic, no low-Mach correction is applied to the non-dimensional energy
 159 flux of the Godunov-like schemes at stake. Indeed, in their approach, the formal expansion in Mach powers related
 160 to the energy PDE is only carried out up to terms scaling as $O(1)$. The authors notably ensure that, for ideal EOS,
 161 the appropriate zeroth-order incompressible pressure equation is retrieved at discrete level. However, nothing is told
 162 about the possibility, for their scheme, to transfer energy flux contributions scaling as $O(M)$ into the non-dimensional
 163 pressure variable from one time-step to another.

164

Finally, the proposed numerical flux at interface $S_{i,j}$ reads:

$$\Phi_{i,j}^{\text{FVS}} = \Gamma_{i,j} + (1 - (\mathcal{E}_0)_{i,j}^2) \Pi_{i,j}, \quad (54)$$

with for $K = L, R$,

$$\begin{cases} \Gamma_{i,j} = \frac{1}{2} [u^* \mathbf{W}_R + u^* \mathbf{W}_L - |u^*| (\mathbf{W}_R - \mathbf{W}_L)] + (\mathcal{E}_0)_{i,j}^2 \begin{pmatrix} 0 \\ p^{*,\theta} \mathbf{n}_{i,j} \\ 0 \end{pmatrix}, & \mathbf{W}_K = \begin{pmatrix} \rho_K \\ (\rho \mathbf{u})_K \\ (\rho e)_K + \mathcal{E}_0^2 p_K \end{pmatrix}, \\ \Pi_{i,j} = \begin{pmatrix} 0 \\ p^{*,\theta} \mathbf{n}_{i,j} \\ p^{*,\theta} u^* \end{pmatrix}, \end{cases} \quad (55)$$

and,

$$\begin{cases} p^{*,\theta} = \frac{p_R + p_L}{2} - \theta_{i,j} \frac{\mathcal{S}_{i,j} (\mathbf{u}_R \cdot \mathbf{n}_{i,j} - \mathbf{u}_L \cdot \mathbf{n}_{i,j})}{2}, \\ u^* = \frac{\mathbf{u}_R \cdot \mathbf{n}_{i,j} + \mathbf{u}_L \cdot \mathbf{n}_{i,j}}{2} - \frac{p_R - p_L}{2\mathcal{S}_{i,j}}. \end{cases} \quad (56)$$

165 Besides, $\mathcal{S}_{i,j} = \max(\rho_{RCR}, \rho_{LCL})$, $\theta_{i,j} = \tilde{M}_{i,j}$ with $\tilde{M}_{i,j}$ the local Mach number defined in Eq. (23) for a two-
 166 dimensional flow. The splitting parameter is set to be $(\mathcal{E}_0)_{i,j} = \max(\mathcal{E}_{\text{inf}}, \min(1, \tilde{M}_{i,j}))$ with $\mathcal{E}_{\text{inf}} = 10^{-8}$. Although
 167 the above formal low-Mach number analysis compels to apply a low-Mach correction to the non-centered part of the
 168 intermediate velocity u^* in the energy flux, the numerical anti-diffusion that it brings endangers the stability of the
 169 proposed flux vector splitting approach. As a consequence, it has been removed from the present numerical scheme.

170
 171 All of the computations presented in the following have been performed with the fast transient dynamics software
 172 for fluids and structures *Europlexus* [34].

173
 174 In the sequel, numerical test cases are provided in order to assess the accuracy of the proposed flux vector splitting
 175 scheme.

176 5. Numerical tests

177 The numerical results presented below go through a wide variety of flow regimes depending on the maximal value
 178 of the Mach number. The first test-case is the one-dimensional Sod's shock tube [35]. It aims at testing the capacity
 179 of the present scheme to capture fast compressible shock and rarefaction waves in regions where the Mach number is
 180 close to one. Then, another Riemann problem whose analytical solution is a stationary contact discontinuity. Here,
 181 one checks the ability of the proposed flux vector splitting scheme to exactly capture, on a large convective time-scale,
 182 a solution of the compressible Euler system associated with the slow material-wave dynamics.

183 Subsequently, one is interested in approximating, with a sufficient accuracy, stationary solutions related to the
 184 two-dimensional *incompressible* Euler system. In that case, the physical time of simulation is systematically based on
 185 the material velocity and the characteristic length-scale of the non-dimensional cartesian mesh is larger than the Mach
 186 number. In this context, the capacity of the present scheme to tackle the low Mach number accuracy issue is assessed.
 187 The first 2-D test-case is a stationary Gresho vortex [36]. The second one is a low Mach number flow in a channel
 188 with a bump [14].

189 Finally, a case composed of a fast acoustic wave, associated with a Mach number of order one, crossing a stationary
 190 Gresho vortex is studied. Here, the capacity of the proposed splitting to adapt locally according to the variations of
 191 the Mach number so that to either capture compressible waves or preserve stationary incompressible solutions is at
 192 stake.

193 Three numerical schemes are here compared: the classical HLLC scheme (given in Eq. (5)), the modified HLLC
 194 scheme labelled "HLLC-Corr" in the following (see Eq. (25)) and the present flux-vector splitting (detailed in Eq.
 195 (54)).

196 Let us end this test-case presentation by saying that in every simulation, the discrete time-step Δt^n is built such
 197 that the Courant number C based on the Euler eigenvalues in Eq. (4) is equal to 0.45.

198 5.1. Test 1: Sod's shock tube

The initial conditions related to this one-dimensional Riemann problem are displayed in Table 1. The solution is

Left state	Right state
$\rho_L^0 = 1 \text{ kg.m}^{-3}$	$\rho_R^0 = 0.125 \text{ kg.m}^{-3}$
$p_L^0 = 1 \text{ bar}$	$p_R^0 = 0.1 \text{ bar}$
$u_L^0 = 0 \text{ m.s}^{-1}$	$u_R^0 = 0 \text{ m.s}^{-1}$

Table 1: Sod's shock tube: initial conditions

199 made of a left-going rarefaction wave, a right-going contact discontinuity and a right-going shock wave. The maximal
 200 Mach number, reached in the tail of the rarefaction wave is $M \approx 0.92$. The time of the simulation, based on the shock
 201 wave speed, is $T_{\text{end}} = 4.5 \times 10^{-4}$ s. In Figure 3, the pressure, velocity and density profiles of the computed solutions
 202 for the three mentioned schemes are plotted. The mesh is made of 10^3 cells. It turns out that the proposed FVS scheme
 203 manages to capture the fast compressible rarefaction and shock waves.
 204

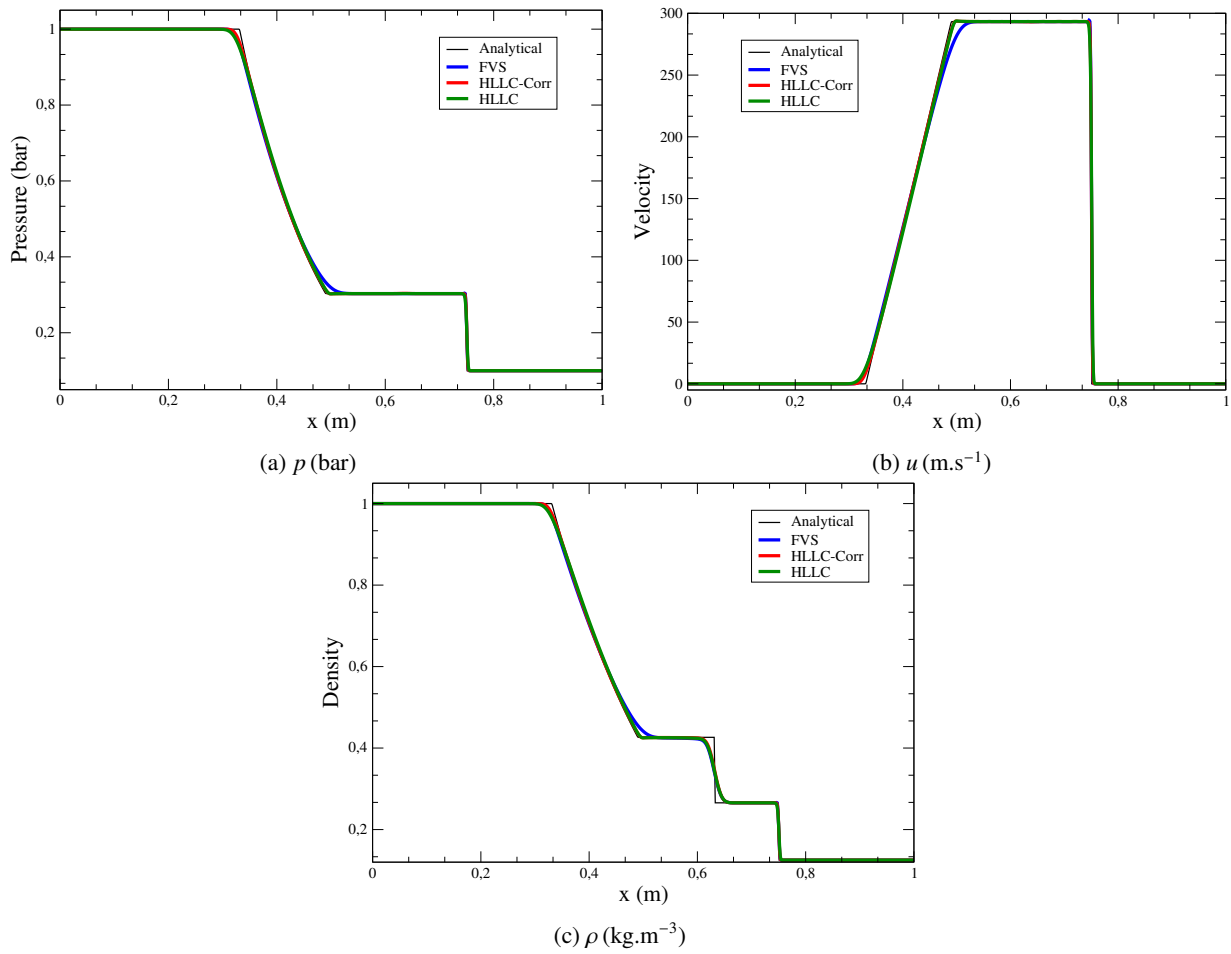


Figure 3: Profiles of (a) the pressure, (b) the velocity and (c) the density for the simulation of the Sod's shock tube: mesh made of 10^3 cells, Courant number $C = 0.45$; comparison between the analytical and the computed solutions.

205 The profiles obtained are equivalent. The proposed FVS scheme is slightly more diffusive at the tail of the rarefac-
 206 tion wave compared to the other HLLC schemes. Indeed, in this region, the Mach number is almost equal to one and
 207 the pressure part, $(1 - \mathcal{E}_0^2) \mathbf{\Pi}_{i,j}$, of the present splitting cancels out. The resulting scheme is then based on a simple
 208 upwinding that, contrary to the two HLLC-type schemes, is not designed to approximate the fastest waves appearing
 209 in the flow.

210 5.2. Test 2: Stationary contact discontinuity

211 In this test-case, initial conditions written in Table 2 are such that pressure is constant and velocity is null in the
 212 overall computational domain. Hence, the analytical solution is made of a single stationary contact wave. The time
 of the simulation is $T_{\text{end}} = 10^{-1}$ s. Similarly to the previous case, pressure, velocity and density profiles at final time

Left state	Right state
$\rho_L^0 = 10 \text{ kg.m}^{-3}$	$\rho_R^0 = 1 \text{ kg.m}^{-3}$
$p_L^0 = 1 \text{ bar}$	$p_R^0 = 1 \text{ bar}$
$u_L^0 = 0 \text{ m.s}^{-1}$	$u_R^0 = 0 \text{ m.s}^{-1}$

Table 2: Stationary contact wave: initial conditions

213 are displayed in Figure 4, for a 10^3 cells mesh. It turns out that the proposed FVS approach is able to preserve exactly
 214 the constant pressure and velocity profile as well as the initial stationary density jump. Similarly to [9], this property
 215 holds because of the pressure flux discretization proposed in Eq. (56). Indeed, it relies on an acoustic approximate
 216 Riemann solver such that the non-centered part of the produced intermediate velocity (respectively the intermediate
 217 pressure) is a function of the pressure difference (respectively of the velocity difference). Let us mention that in this
 218 test-case, the parameter \mathcal{E}_0 is equal to its inferior bound $\mathcal{E}_{\text{inf}} = 10^{-8}$ because of the zero-velocity field. The splitting
 219 thus decouples convection at material velocity from acoustic wave production.
 220

221 5.3. Test 3: 2-D stationary vortex

The third test-case is a stationary version of the vortex proposed by Gresho and Chan [36, 37] and also considered
 in [17]. At the initial time $t = 0$, a vortex of radius $R = 0.4$ m whose the center is located at the position $(x_0, y_0) = (1, 1)$
 is prescribed:

$$\begin{cases} \rho &= \rho_0, \\ u &= -\frac{y - y_0}{r} \delta u, \\ v &= \frac{x - x_0}{r} \delta u, \\ p &= p_0 + \delta p, \end{cases} \quad (57)$$

with

$$\delta u = M_0 c_0 \begin{cases} 2r/R & \text{if } 0 \leq r < R/2, \\ 2(1 - r/R) & \text{if } R/2 \leq r < R, \\ 0 & \text{if } R \leq r, \end{cases}$$

and

$$\delta p = p_0 M_0^2 \begin{cases} 2(r/R)^2 & \text{if } 0 \leq r \leq R/2, \\ 2(r/R)^2 + 4(1 - 2r/R + \log(2r/R)) & \text{if } R/2 < r \leq R, \\ -2 + 4 \log(2) & \text{if } R < r, \end{cases}$$

where $\rho_0 = 1 \text{ kg.m}^{-3}$, $p_0 = 1 \text{ bar}$, $M_0 = 10^{-2}$ with $c_0 = \sqrt{\gamma p_0 / \rho_0}$ and $r = \sqrt{(x - x_0)^2 + (y - y_0)^2}$. Let us underline
 that the velocity and pressure fields are smooth solutions of the stationary incompressible Euler system:

$$\begin{cases} \nabla \cdot \mathbf{u} &= 0 \\ \rho_0 \mathbf{u} \cdot \nabla \mathbf{u} + \nabla p &= \mathbf{0}. \end{cases} \quad (58)$$

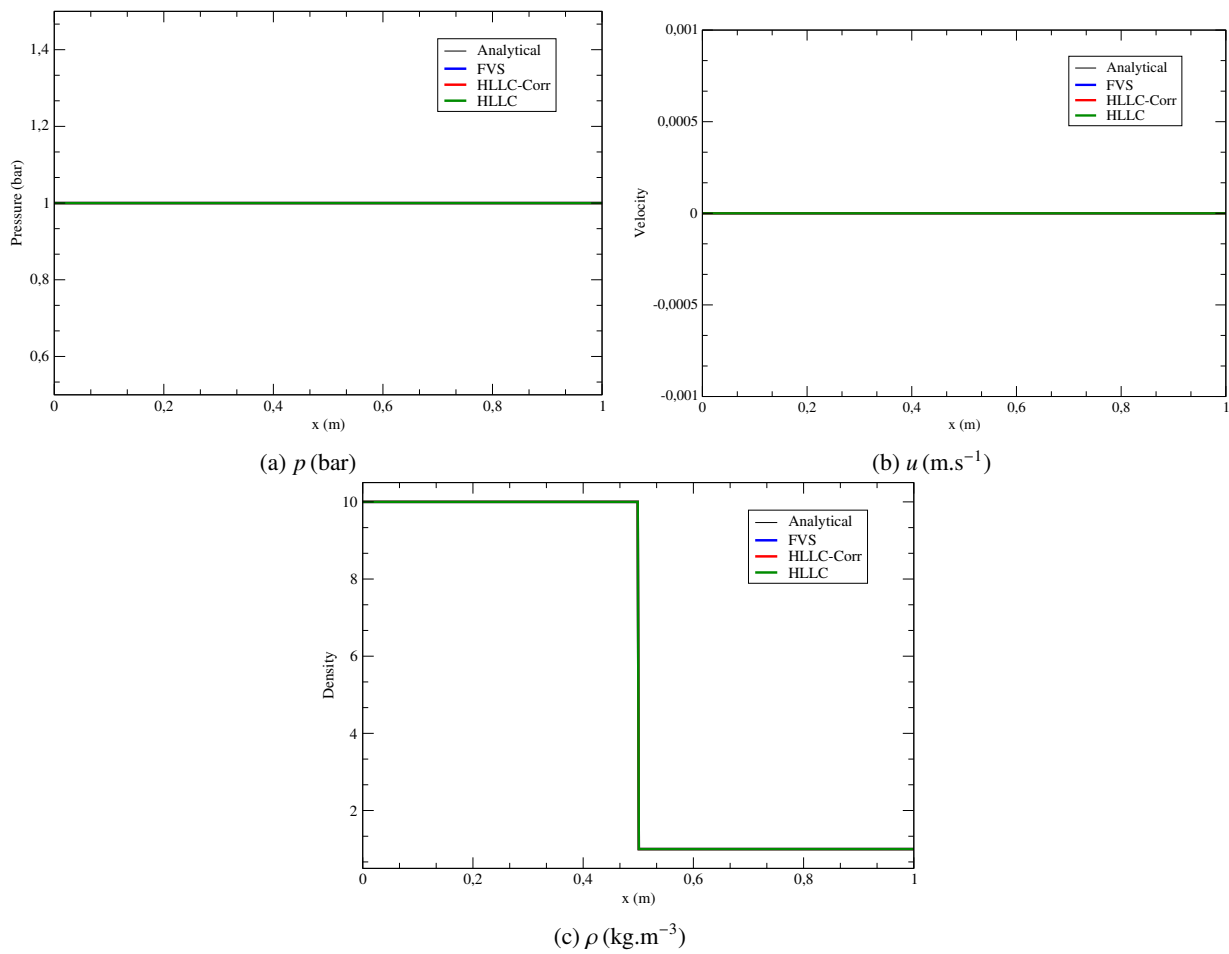


Figure 4: Profiles of (a) the pressure, (b) the velocity and (c) the density for the simulation of the stationary contact wave: mesh made of 10^3 cells, Courant number $C = 0.45$; comparison between the analytical and the computed solutions.

222 The 2-D computation domain corresponds to $[0, L] \times [0, L]$ with $L = 2$ m. All the computations are done with 50×50
 223 cells. The final time of the computation is $t = 0.5$ s corresponding approximately to the convective time scale L/u_0
 224 with $u_0 = M_0 c_0$. Transmissive boundary conditions are used for the inlet, outlet, top and bottom boundaries.

225 The numerical solutions obtained with the HLLC scheme, the modified HLLC solver and the present FVS ap-
 226 proach are compared in Figure 5 and Figure 6. The profiles of the Mach number, the velocity norm $|\mathbf{u}|$ as well as
 the non-dimensional incompressible pressure component $(p - p_0)/p_0$ are displayed. The numerical results obtained

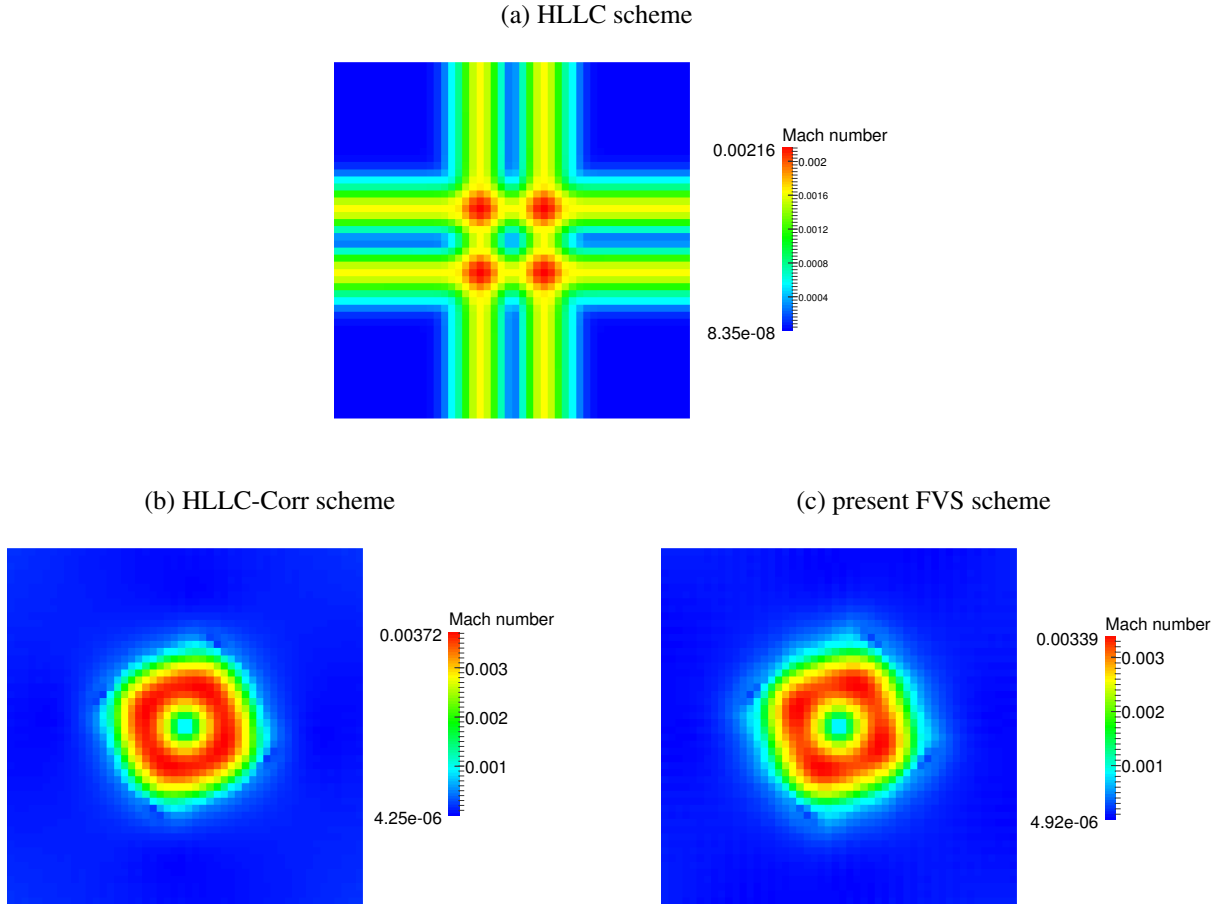


Figure 5: Mach number for simulations of the stationary vortex: $M_0 = 10^{-2}$, mesh made of 50×50 cells, Courant number $C = 0.45$; with (a) the HLLC scheme, (b) the modified HLLC scheme and (c) the present approach.

227 with the HLLC solver deteriorate the vortex leading to non-cylindrical structures as observed in Figure 5-(a) on the
 228 Mach number profile. In contrast, the present FVS approach gives similar results as the modified HLLC scheme with
 229 a strongly improved resolution of the vortex.

230 This is also observed on the velocity norm and non-dimensional incompressible pressure profiles plotted on Fig-
 231 ure 6. Let us finally underline that, for both HLLC-Corr and the present FVS approach, structures looking like checker
 232 board modes seem to appear on the profile of $(p - p_0)/p_0$. A close look on the scale of these modes reveals that their
 233 amplitude is larger in the case of the HLLC-Corr scheme than in the proposed method.
 234

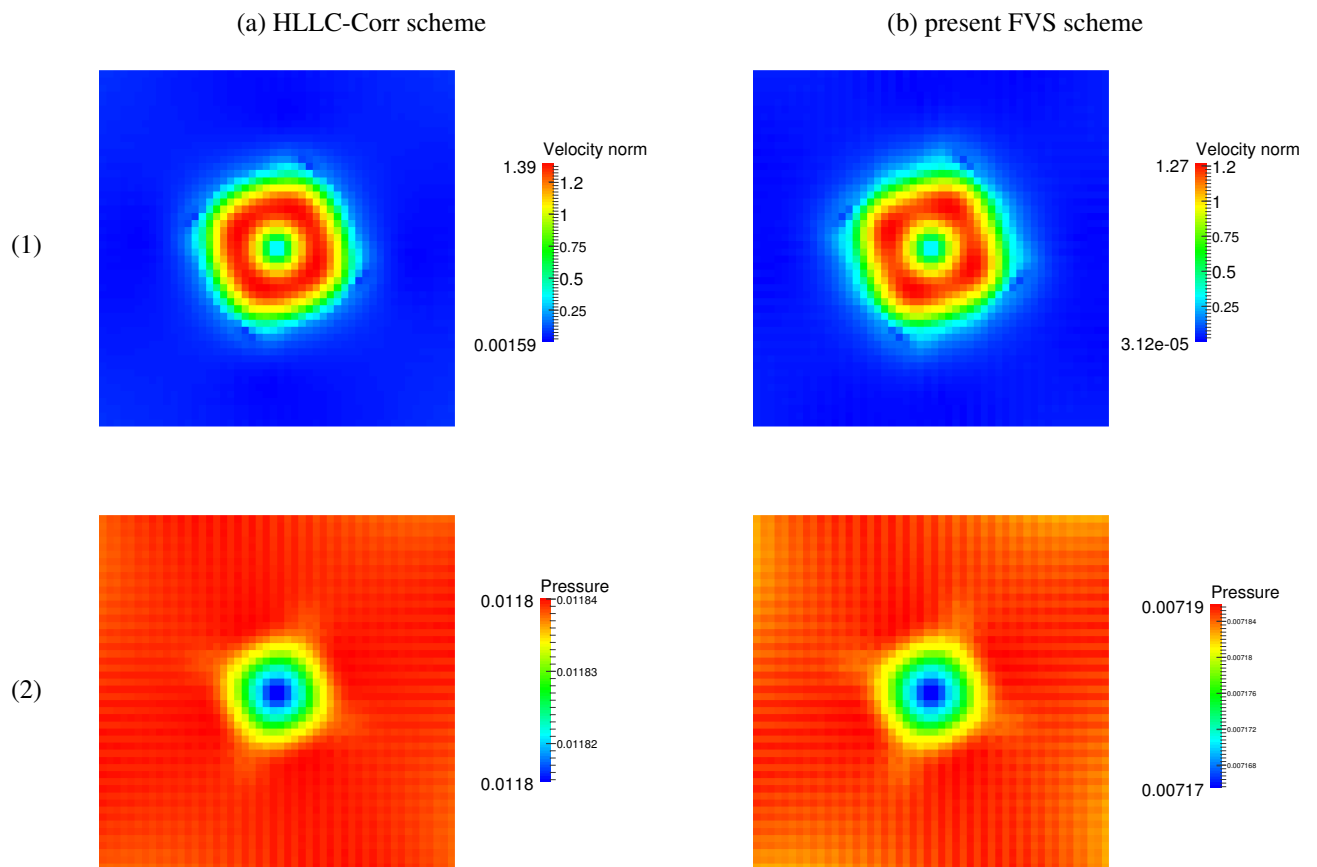


Figure 6: Profiles of (1) the velocity norm $|\mathbf{u}|$, (2) $(p - p_0) / p_0$ for simulations of the stationary vortex: $M_0 = 10^{-2}$, mesh made of 50×50 cells, Courant number $C = 0.45$; with (a) the HLLC scheme, (b) the modified HLLC scheme and (c) the present approach.

235 5.4. Test 4: Subsonic flow in a channel with a bump

The fourth test-case considered here consists of a low-velocity compressible flow in a channel with a bump as proposed in [14]. The initial flow conditions are characterized by:

$$\rho_0 = 1.2 \text{ kg.m}^{-3}, \quad p_0 = 1 \text{ bar} \quad \text{and} \quad u_0 = M_0 c_0, \quad (59)$$

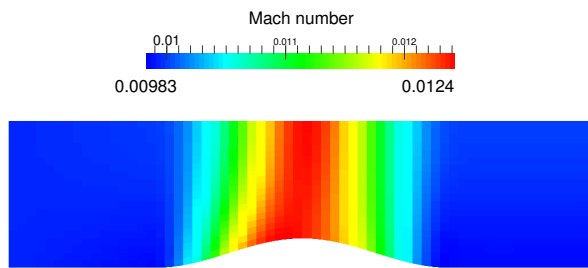
236 with $c_0 = \sqrt{\gamma p_0 / \rho_0}$. The 2-D computation domain corresponds to $[-2L, 2L] \times [0, L]$ with $L = 1$ m. The numerical
 237 solutions obtained with the classical HLLC solver, the modified HLLC scheme and the present approach are compared.
 238 Two set of computations are considered in the following: the first with $M_0 = 10^{-2}$ and the second with $M_0 = 10^{-3}$.
 239 All the computations are done with 64×20 cells and the Courant number $C = 0.45$. For the case $M_0 = 10^{-2}$, the final
 240 time of the computation is $t = 1.17$ s corresponding approximately to the convective time scale $4L/u_0$. It is ten times
 241 longer in the case $M_0 = 10^{-3}$. Inlet and outlet boundary conditions are obtained using the corresponding Riemann
 242 invariants whereas upper and lower boundaries are set to be walls.

243 The numerical results are plotted in Figure 7 and Figure 8. The Mach number, the non-dimensional incompress-
 244 ible pressure $(p - p_0) / p_0$ as well as the density profiles are displayed. First of all, one can observe that the numerical
 245 solution given by the classical HLLC scheme is not correct as observed in the Mach number field. Indeed, as already
 246 proved in [12], the stationary solution captured by the upwind solver corresponds to a *balance of acoustic waves*
 247 instead of an approximation of the incompressible field for which the Bernouilli's principle must hold. The numerical
 248 density fields obtained with the HLLC scheme in conjunction with the classical low-Mach correction on the momen-
 249 tum equation is not correct too. In contrast, the present approach using a correction on both momentum and energy
 250 equation gives satisfactory numerical results on all the variables.

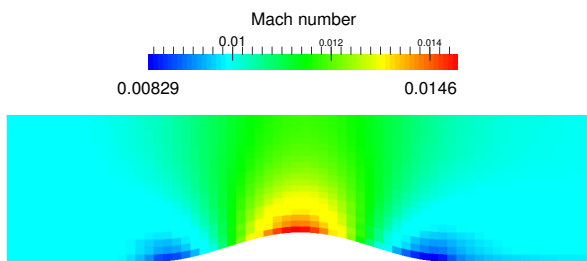
251 In the case $M_0 = 10^{-3}$, the profiles of the Mach number, of $(p - p_0) / p_0$ and of the density are shown in Figure 9.
 252 The present FVS approach is only compared to the HLLC-Corr scheme.

253 The quality of the profiles between both schemes is similar for the Mach number and the non-dimensional incom-
 254 pressible pressure variables. Besides, as already observed previously, a low-Mach number correction holding on the
 255 non-centered part of p^* in the energy flux allows to retrieve an appropriate profile for the density.

(a) HLLC scheme



(b) HLLC-Corr scheme



(c) present FVS scheme

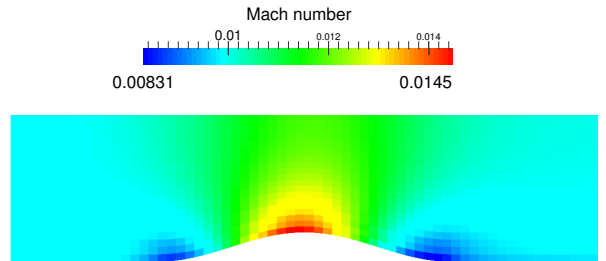


Figure 7: Mach number for simulations of the subsonic flow in a channel with a bump: $M_0 = 10^{-2}$, mesh made of 64×20 cells, Courant number $C = 0.45$; with (a) the HLLC scheme, (b) the modified HLLC scheme and (c) the present approach.

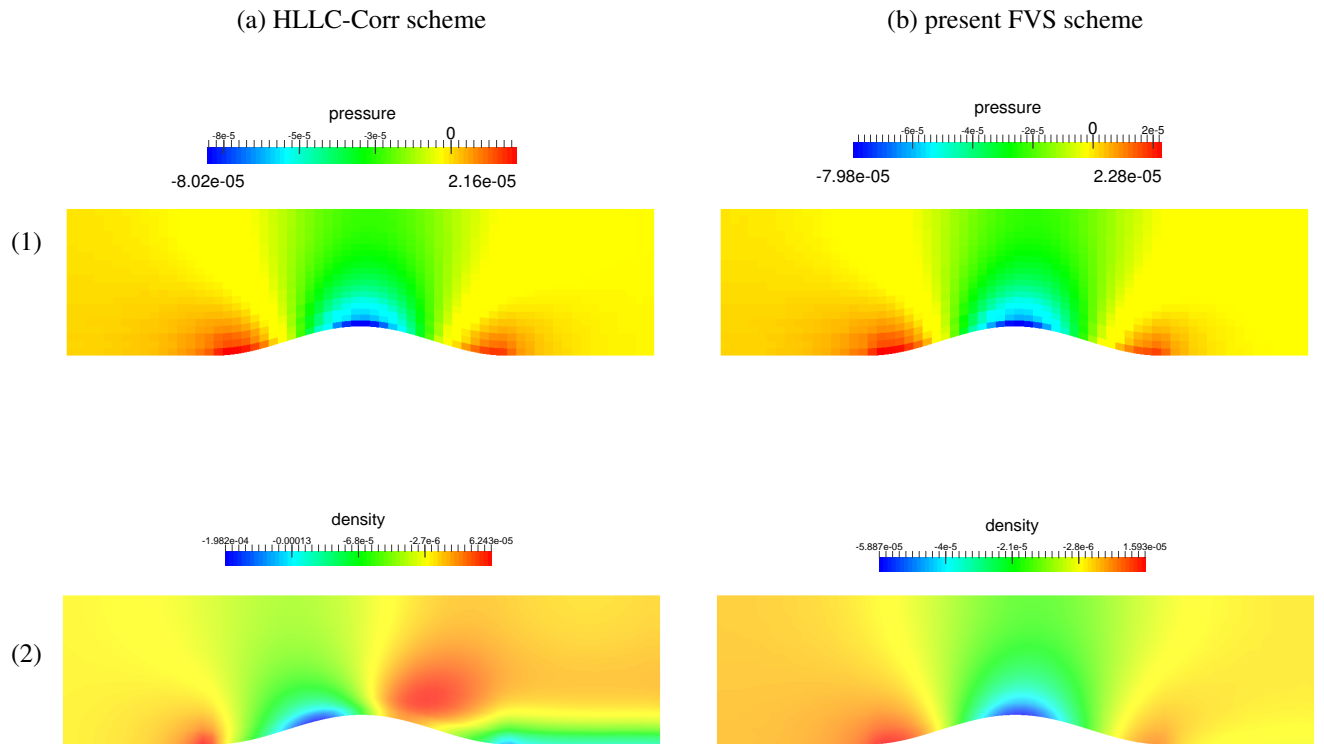


Figure 8: Profiles of (1) $(p - p_0) / p_0$ and (2) $(\rho - \rho_0) / \rho_0$ for simulations of the subsonic flow in a channel with a bump: $M_0 = 10^{-2}$, mesh made of 64×20 cells, Courant number $C = 0.45$; with (a) the HLLC scheme, (b) the modified HLLC scheme and (c) the present approach.

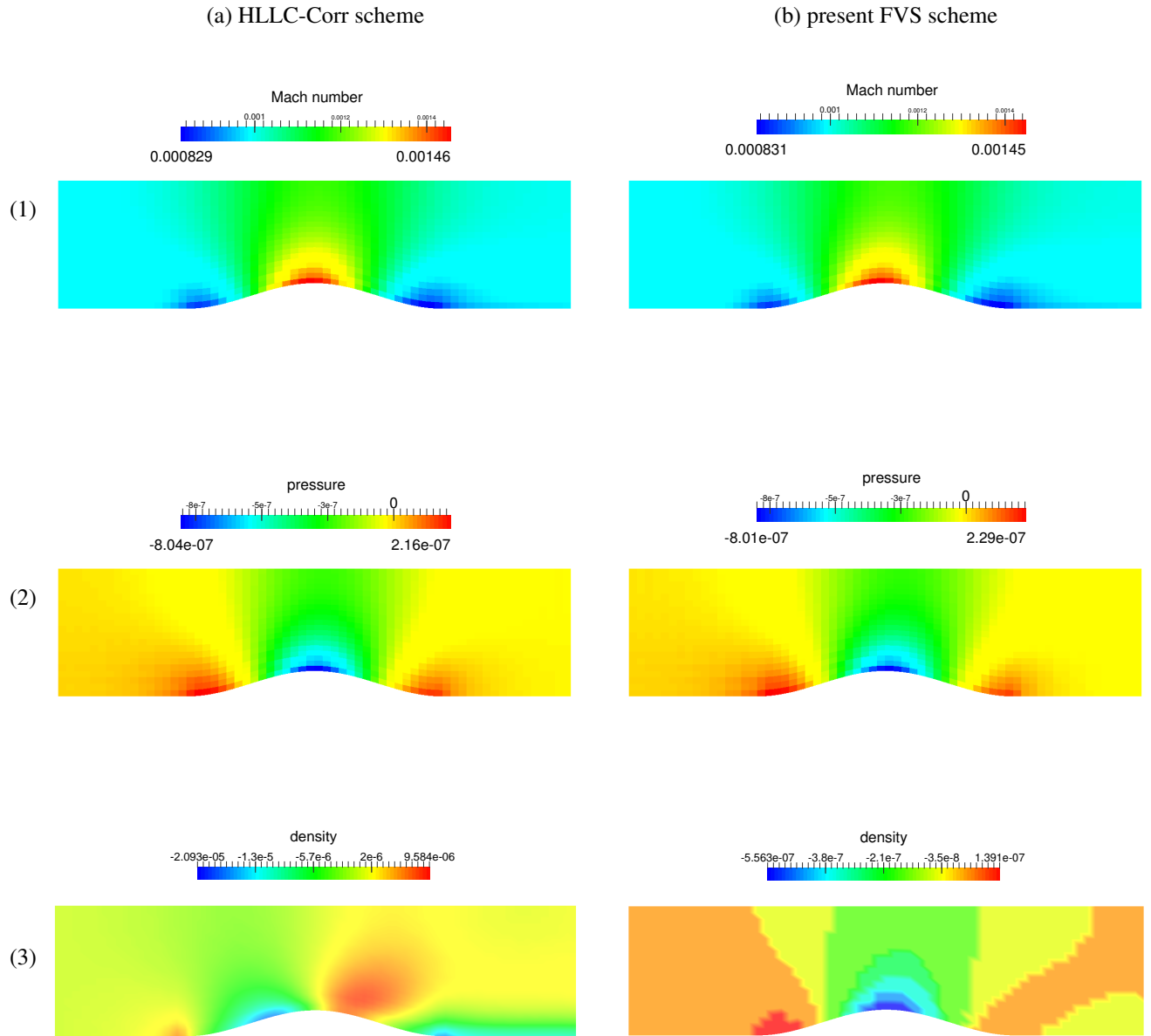


Figure 9: Profiles of (1) the Mach number, (2) $(p - p_0) / p_0$ and (3) $(\rho - \rho_0) / \rho_0$ for simulations of the subsonic flow in a channel with a bump: $M_0 = 10^{-3}$, mesh made of 64×20 cells, Courant number $C = 0.45$; with (a) the modified HLLC scheme, (b) the present approach.

256 *5.5. Test 5: Interaction acoustic wave with a 2-D stationary vortex*

The last test-case has been inspired from the work of Miczek and co-authors [38]. The computational domain represents a pipe section whose diameter is $L = 2$ m and length is $5 \times L = 10$ m. The 2-D cartesian mesh is made of 50×250 cells. At time $t = 0$ s, a stationary vortex similar to the one already introduced in subsection (5.3) is present at $(x_0, y_0) = (7, 1)$. Simultaneously, a smooth but steep pressure pulse is injected along all the pipe section at $x_1 = 5$ m. Such a pulse reads:

$$p = p_0 \left(1 + 9 \times e^{-(x-x_1)^2/2\delta^2} \right), \quad \delta = 1.7 \times 10^{-2} \text{ m}, \quad (60)$$

$$p_0 = 1 \text{ bar}.$$

257 The value of the parameter δ is chosen such that, initially, the width of the acoustic pressure pulse is smaller than the
 258 length of one cell of the considered mesh. Outside the vortex, the other initial conditions are $(u_0, v_0) = (0, 0)$ and
 259 $\rho_0 = 1 \text{ kg.m}^{-3}$. Transmissive boundary conditions hold on the left and right borders of the computational domain.
 260 The top and bottom borders are considered to be walls.

261 At $t = 0$ s, the initial pressure pulse generates symmetric acoustic waves associated with a rise of the velocity be-
 262 hind their front line. In these regions the Mach number is about 0.4 by contrast with the maximal Mach number inside
 263 the vortex which is approximately 10^{-2} showing the wide range of Mach number values involved simultaneously in
 264 the present case. Within an acoustic time-scale, the acoustic wave has interacted and then crossed the vortex.

265 As previously mentioned, the adaptivity of the splitting is assessed in this test-case. Indeed, one would like to
 266 check that the proposed FVS approach is able to locally decouple the convective and pressure parts in low Mach
 267 number regions and retrieve the overall Euler system when the Mach number rises. In the low Mach number regions,
 268 the corrections holding on the pressure part of the flux should allow to be accurate with respect to the incompressible
 269 solutions. In regions where the Mach number scales as $O(1)$, the classical upwinding performed on the convective
 270 flux should allow to capture the fast acoustic wave dynamics.

271 Figure 10 displays the propagation of the acoustic pressure wave at the beginning, half and final time of the
 272 simulation. The present approach is compared to the HLLC scheme. It turns out that the proposed scheme manages to
 273 capture and follow the compressible wave. Because the maximal flow Mach number is about 0.4, the pressure part of
 274 the splitting does not completely cancel and the anti-diffusion brought by low-Mach corrections contribute to preserve
 275 the acoustic wave profile.

276 Finally, in Figure 11, the flow Mach number is plotted using the scaling of the Gresho vortex. One can observe
 277 that after having interacted with the acoustic wave, the structure of the stationary vortex is better preserved in the case
 278 of the present FVS approach than with the HLLC scheme.

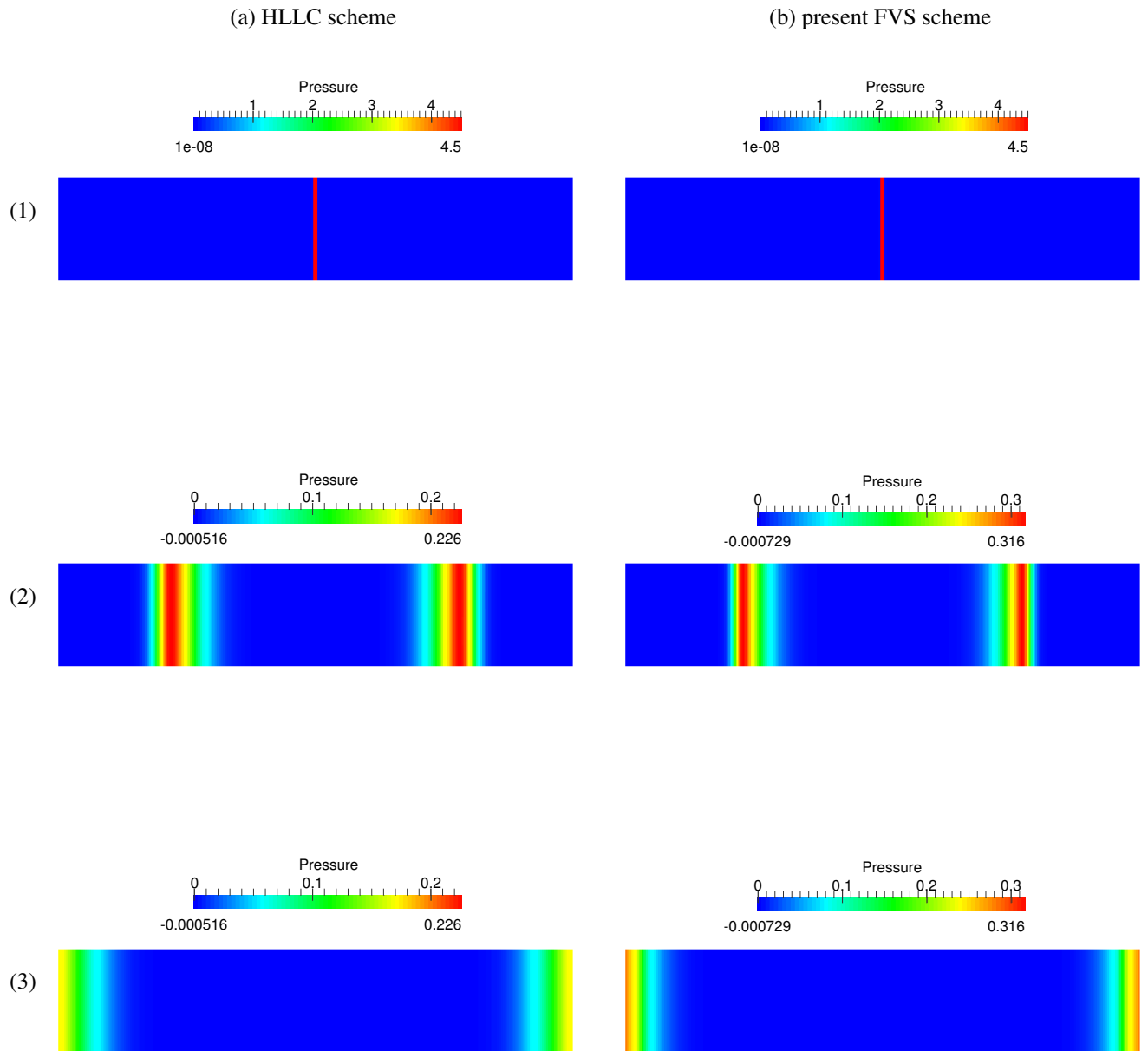


Figure 10: Acoustic pressure waves at time (1) $t = 0$ s, (2) $t = 6 \times 10^{-3}$ s and (3) $t = 1.2 \times 10^{-2}$ s, mesh made of 50×250 cells, Courant number $C = 0.45$; with (a) HLLC scheme, (b) the present approach

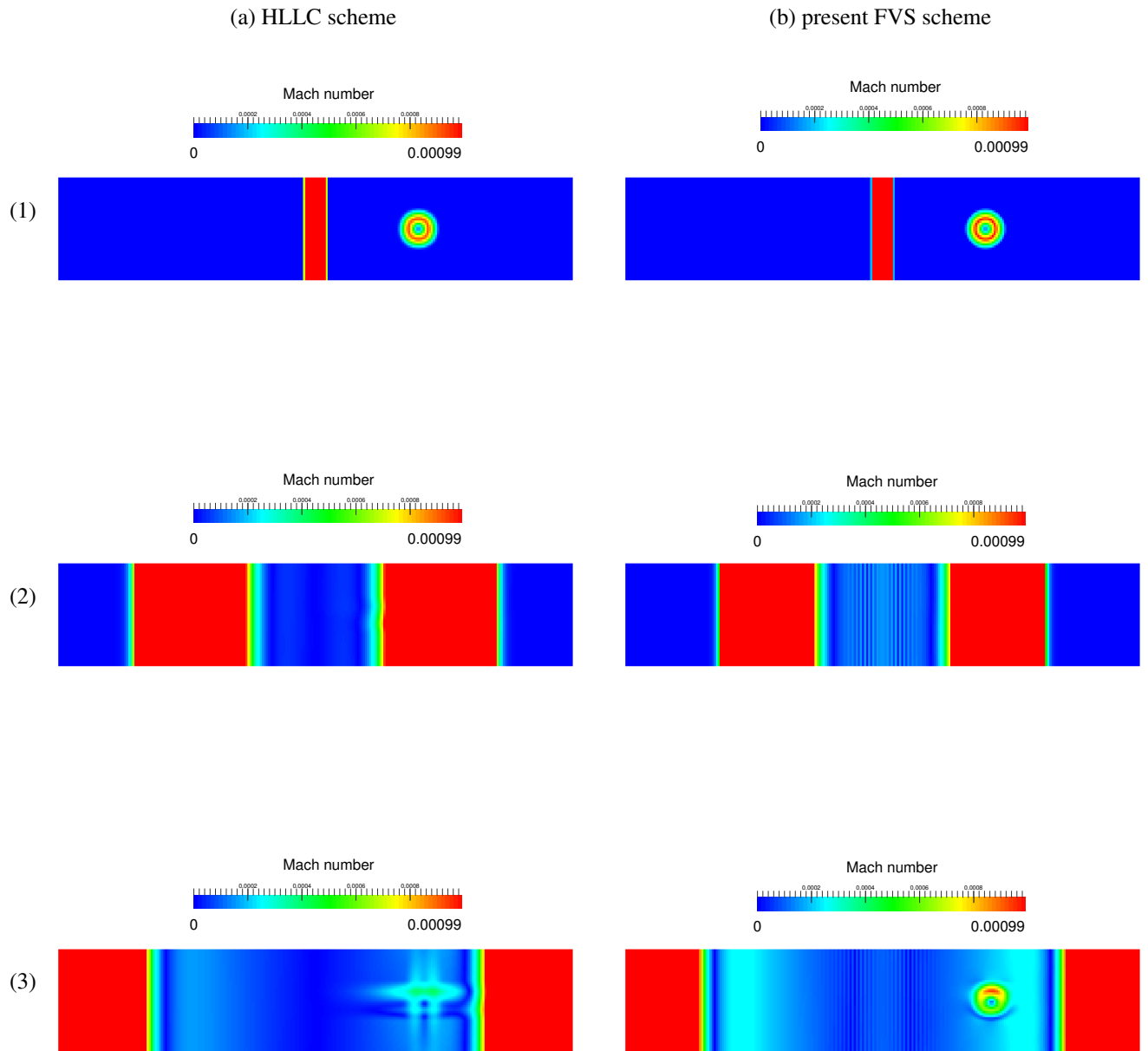


Figure 11: Mach number at time (1) $t \approx 10^{-6}$ s, mesh made of 50×250 cells, Courant number $C = 0.45$; with (a) HLLC scheme, (b) the present approach

279 6. Conclusions and perspectives

280 In the present work, a time-explicit Flux-Vector Splitting solver has been proposed. Its construction relies on a
281 dynamic splitting of the Euler system flux highlighting a convective and a pressure part. The dynamic aspect of the
282 splitting stems from a unique parameter \mathcal{E}_0 which follows the space-and-time variations of the Mach number across the
283 computational domain. In low-Mach number regions, \mathcal{E}_0 tends towards zero, and the whole acoustic wave production
284 is transferred to the pressure part of the splitting. A HLLC-like method applied to the underlying acoustic subsystem
285 is then used to discretize the pressure flux. Besides, in order to handle the low-Mach accuracy issue inherent to any
286 Godunov-like scheme, low-Mach corrections holding on the momentum *and* the energy components of the pressure
287 flux have been added. In sonic or highly subsonic regions, \mathcal{E}_0 is equal to one and the entire compressible Euler flux
288 is retrieved in the convective part of the splitting. In that case a simple upwind scheme, based on the intermediate
289 velocity and pressure produced by the pressure part, is used to discretize the convective flux.

290 A Sod's shock tube has allowed to verify that the proposed approach is able to capture fast compressible shock
291 and rarefaction waves. Besides, it can also preserve exactly stationary contact waves. When tested against stationary
292 incompressible test cases, the low-Mach corrections injected in the pressure flux of the present method contribute
293 to capture the appropriate incompressible solutions. Besides, contrary to other time-explicit solvers, the proposed
294 scheme seems to be stable under a classical CFL condition based on the Euler eigenvalues. However, let us recall that,
295 the formal analysis holding on the invariance principle of the discrete incompressible phase space suggests that a low-
296 Mach correction should also be applied to the non-centered part of the intermediate velocity in the energy component
297 of the pressure flux. Such a correction has been voluntarily avoided as it disturbed the stability of the present method.
298 Hence, a first perspective to the present work could be to carry out a stability study based on a linearized version of
299 the splitting in order to determine the optimal scaling of \mathcal{E}_0 with respect to the different low-Mach corrections. An
300 other perspective could be to derive an implicit-explicit version of the present FVS scheme. Indeed the time-implicit
301 integration of the pressure part of the flux might contribute to increase the range of values of the Courant number for
302 which the method is stable. Eventually, one could also check that such an implicit-explicit version of the scheme is
303 asymptotically preserving towards the Euler incompressible system.

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309 (<http://www-epx.cea.fr/>).

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