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# Experimental investigation of the influence of internal frames on the vibroacoustic behavior of a stiffened cylindrical shell using wavenumber analysis

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## Abstract

The understanding of the influence of non-axisymmetric internal frames on the vibroacoustic behavior of a stiffened cylindrical shell is of high interest for the naval or aeronautic industries. Several numerical studies have shown that the non-axisymmetric internal frame can increase the radiation efficiency significantly in the case of a mechanical point force. However, less attention has been paid to the experimental verification of this statement. That is why this paper proposes to compare the radiation efficiency estimated experimentally for a stiffened cylindrical shell with and without internal frames. The experimental process is based on scanning laser vibrometer measurements of the vibrations on the surface of the shell. A transform of the vibratory field in the wavenumber domain is then performed. It allows estimating the far-field radiated pressure with the stationary phase theorem. An increase of the radiation efficiency is observed in the low frequencies. Analysis of the velocity field in the physical and wavenumber spaces allows highlighting the coupling of the circumferential orders at the origin of the increase in the radiation efficiency.

*Keywords:* Stiffened shell, Vibroacoustics, Non-axisymmetric, Laser vibrometer, wavenumber analysis.

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## 1. Introduction

Many works can be found in the literature dealing with the vibroacoustic behavior of cylindrical shells reinforced by circumferential stiffeners [1, 2, 3]. As a matter of fact, many structures in the aeronautical or naval industry can be modeled by stiffened cylindrical shells. Williams *et al.* [4] measured the nearfield pressure on a submerged cylinder excited by a point force. They deduced the vibration and radiation from the cylinder thanks to a technique called the generalized nearfield acoustical holography. In a subsequent study, the authors

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analyzed the results in the wavenumber domain [5]. This analysis shows a great  
10 interest to give a physical insight of the vibroacoustic behavior of the cylindrical  
shell. More particularly, wavenumber diagrams can be plotted and the waves  
contributing to the far-field radiation can be identified. The technique is later  
used by Photiadis *et al.* to investigate the response of a stiffened cylinder in  
the wavenumber domain [6]. The wavenumber diagrams show more complex  
15 patterns, due to the stiffeners periodicity and the propagation of Bloch-Floquet  
waves. Slowness surfaces, which are given by the normal to the curves of large  
amplitude and large radius of curvature on the wavenumber diagrams, are used  
to predict the dominant directions of wave propagation.

The studies mentioned previously are limited to axisymmetric systems. How-  
20 ever, in industrial applications, internal structures such as floor partitions or  
engine foundations can have an influence on the vibroacoustic behavior of the  
stiffened cylindrical shell. Experimental work has been carried out for a stiff-  
ened cylindrical shell coupled to numerous oscillators, and the influence on the  
scattered or radiated pressure has been highlighted [7, 8]. The results show  
25 that adding internal degrees of freedom reduces scattering from flexural Bloch-  
Floquet waves and gives rise to a noisy speckle pattern. Nevertheless, in the case  
of a point mechanical excitation, only a few studies based on numerical simula-  
tion have investigated the effect of non-axisymmetric internal frames. Some of  
them use discretization methods such as Finite Element Method and Boundary  
30 Element Method to predict the radiated pressure by an immersed shell with  
and without internal masses or structures and show the importance of flexible  
mounts [9, 10]. Other studies use substructuring approaches based on the ad-  
mittance method to include internal structures in an axisymmetric cylindrical  
shell [11, 12]. The main results are that more circumferential orders tend to  
35 play a role in the response of the cylindrical shell when it is coupled to a non-  
axisymmetric internal frame. If the problem is decomposed on a modal basis, it  
is equivalent to say that the non-axisymmetry tends to couple the circumferen-  
tial orders [13]. As the structural waves having low wavenumbers are the main  
contributors to the far-field radiated pressure, the coupling with low circumfer-  
40 ential numbers tends to increase the radiation efficiency. Another effect of the  
internal frame is given by the fuzzy internal theory [14, 15]. This theory states  
that if the internal structure has a high modal density and if the impedance  
at the junction with the cylindrical shell is adapted, the addition of internal  
degrees of freedom can give rise to an apparent damping effect.

45 The aim of the present work is to discuss experimental results of a point-  
driven stiffened cylindrical shell with and without a non-axisymmetric internal  
frame. The effect of the internal structure on the vibroacoustic behavior of  
the cylindrical shell, and more especially on the radiation efficiency is inves-  
tigated. The main obstacle for estimating accurately the radiation efficiency  
50 lies in the estimation of the radiated power. Precision methods using sound  
pressure require a large number of microphones and are difficult to set up [16].  
Measurement by scanning using sound intensity [17] is widely used to measure  
the radiated power from machines, but it can be difficult to define an enclos-  
ing surface around the system. To tackle these issues, the methodology of the

55 present work is based on the use of the stationary phase theorem. It relates the far-field radiated pressure to the spectral velocity field of the shell (*i.e.* the velocity field in the wavenumber domain). An advantage of this approach over the measurement with microphones or intensity probes is that the directivity can also be estimated. The spectral velocity field can be obtained with a scanning  
60 laser vibrometer and a 2D Fourier transform. Attention is paid to the definition of the scanning grid in order to avoid aliasing effect.

The paper is organized as follows:

- Background on the estimation of the far-field radiated pressure from the spectral displacements of the shell is given in section 2.
- 65 • The system and the experimental setup are presented in section 3.
- Operational dispersion curves plotted from the spectral velocity field are discussed in section 4.
- In section 5, the stationary phase theorem is used to determine the far-field radiated power and to investigate the influence of the non-axisymmetric frame on the radiation efficiency.  
70
- Velocity fields are compared for the axisymmetric and non-axisymmetric cases in the real space and in the wavenumber domain in section 6.
- Finally, conclusions are drawn in section 7.

## 2. Spectral displacements and far-field pressure

75 Let us consider a cylindrical shell of radius  $R_0$  immersed in an acoustic domain. The shell can either be of infinite length or of finite length extended by a cylindrical baffle. Considering the cylindrical coordinates  $(x, r, \theta)$ ,  $x$  being the axis of the shell,  $W(x, \theta)$  is the radial shell displacements at a given angular frequency  $\omega$ .

80 A two dimensional Fourier transform can be used to transform the spatial variations of a vibratory field into the wavenumber domain. The 2D Fourier transform of  $W(x, \theta)$  is given by

$$W(x, \theta) \mapsto \tilde{W}(k_x, n) = \int_{\mathbb{R}} \int_{\theta=0}^{2\pi} W(x, \theta) e^{-j(k_x x + n\theta)} dx d\theta \quad (1)$$

where  $k_x$  is the axial wavenumber and  $n$  the circumferential order. As the cylindrical shell is  $2\pi$ -periodic along the circumferential coordinate  $\theta$ , then  $n$  is  
85 a relative integer.

The wavenumber analysis gives an opportunity to analyze the wave propagation on the structure and the role played by the contribution of each circumferential order. The wavenumber analysis also gives an insight on the spectral

components contributing to the radiated pressure, through the radiation circle  
 90 defined by the points of coordinates  $(\bar{k}_x, \bar{n})$  in the wavenumber domain

$$\left(\frac{\bar{n}}{R_0}\right)^2 + \bar{k}_x^2 = k_0^2 \quad (2)$$

where  $k_0 = \frac{\omega}{c_0}$  is the acoustic wavenumber and  $c_0 = 340 \text{ m.s}^{-1}$  is the sound  
 velocity in air. For a plate, the radiation circle represents the supersonic acoustic  
 domain, in which the components radiate to the far-field. It remains a good  
 indicator for localizing the radiating waves on a cylindrical shell when the effect  
 95 of the curvature is low [5].

The vibratory field in the wavenumber domain can be used to estimate the  
 far-field radiated pressure. For  $\mathcal{R}$  in the acoustic far-field, *i.e.*  $\mathcal{R} \gg \frac{2\pi}{k_0}$ , the  
 asymptotic expression for the Hankel function and the stationary phase theorem  
 can be used [18]. If  $\xi$  is the angle from the point to the normal axis and  $\theta$  is the  
 100 circumferential angle, the far-field pressure  $p$  can be written as:

$$p(\mathcal{R}, \xi, \theta) = \sum_{n=-\infty}^{+\infty} \frac{j\rho_0\omega^2}{2\pi^2\mathcal{R}k_0 \cos \xi} \frac{\tilde{W}(-k_0 \sin \xi, n)}{H_n^{(2)'}(R_0 k_0 \cos \xi)} e^{-j\mathcal{R}k_0 + jn(\theta + \frac{\pi}{2})} \quad (3)$$

where  $\rho_0 = 1.2 \text{ kg.m}^{-3}$  is the density of air and  $H_n^{(2)'}$  is the derivative of the  
 Hankel function of second kind and order  $n$ . For a cylinder of finite length,  
 the stationary phase theorem is based on the assumption of a baffled cylinder.  
 However, the influence of the baffle on the radiation due to radial vibrations  
 105 is weak when the shell is long compared to the acoustic wavelength [19]. The  
 stationary phase theorem may then be used to predict the radiated pressure from  
 the cylindrical shell. One of the advantages of the wavenumber analysis is that  
 it allows the calculation of the radiated pressure from vibrations measurements  
 only.

110 The far-field radiated power  $P_a$  is obtained by integrating the radiated pressure  
 over the surface of the sphere of radius  $\mathcal{R}$ :

$$P_a = \frac{1}{\rho_0 c_0} \int_{\xi=0}^{\pi} \int_{\theta=0}^{2\pi} p^2(\mathcal{R}, \xi, \theta) \mathcal{R}^2 \sin \theta d\xi d\theta \quad (4)$$

As the acoustic domain is conservative, any value of  $\mathcal{R}$  in the far-field can be  
 chosen to yield  $P_a$ . The radiation efficiency  $\sigma$  of a cylindrical shell is defined  
 by the ratio between the actual far-field radiated power  $P_a$  and the power that  
 115 would be radiated by a cylindrical shell of the same radiating surface  $S =$   
 $2\pi R_0 L$ , having an uniform radial velocity equal to the mean quadratic velocity  
 $\langle \dot{W} \rangle$ :

$$\sigma = \frac{P_a}{\rho_0 c_0 S \langle \dot{W} \rangle^2} \quad (5)$$

The mean quadratic velocity is defined as follows:

$$\langle \dot{W} \rangle^2 = \frac{1}{S} \int_{x=0}^L \int_{\theta=0}^{2\pi} |\dot{W}(x, \theta)|^2 dx R_0 d\theta \quad (6)$$

with  $\dot{W}(x, \theta)$  the point radial velocity at the point of coordinates  $(x, R_0, \theta)$ .

120 **3. Presentation of the experimental setup**

*3.1. Axisymmetric stiffened cylindrical shell*

A stiffened cylindrical shell, which geometrical characteristics are given in table 1, is shown in Fig. 1. The cylindrical shell is made of standard non-alloy steel, for which common values may be considered: Young's modulus  $E = 2.1 \cdot 10^{11}$  Pa, density  $\rho_s = 7800 \text{ kg.m}^{-3}$  and Poisson's coefficient  $\nu = 0.3$ . It has 72 identical stiffeners of rectangular cross-section, which have been machined from a thicker cylinder. The cylindrical shell is divided in five sections with three different stiffeners spacings, as shown in the sketch in Fig. 1c. 15 mm thick caps are screwed at the ends of the cylinder. This stiffened cylindrical shell is referred to as the axisymmetric case.

Table 1: Stiffened cylindrical shell dimensions (mm).

Parameter	Notation	Value
Radius	$R_0$	100
Length	$L$	1500
Shell thickness	$h$	1.5
Stiffeners height	$h_w$	5
Stiffeners thickness	$t_w$	1
Stiffeners spacing 1	$\alpha$	24
Stiffeners spacing 2	$\beta$	20
Stiffeners spacing 3	$\gamma$	17

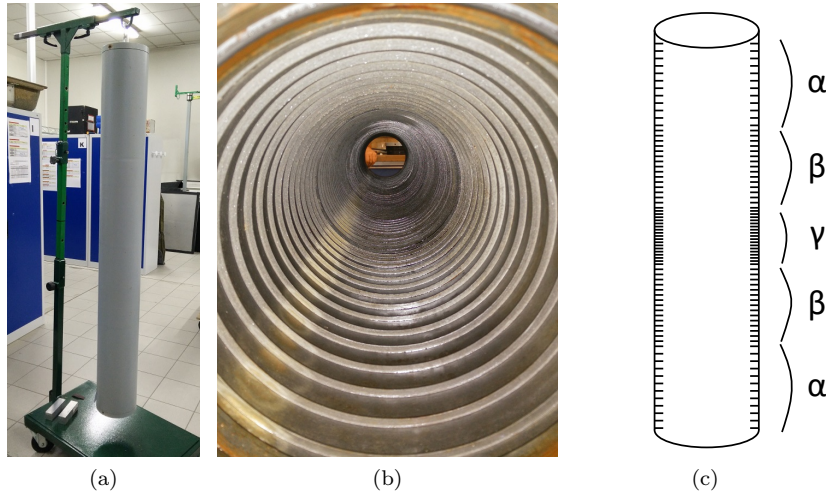


Figure 1: Pictures of the axisymmetric stiffened cylindrical shell: (a) hanging on an arm and (b) interior view. (c) Sketch of the cylindrical shell with the sections of different stiffeners spacings.

The ring frequency  $f_r$ , for which the longitudinal wavelength is equal to the circumference, is defined by the following equation [18]:

$$f_r = \frac{1}{2\pi R_0} \sqrt{\frac{E}{\rho_s(1-\nu^2)}} = 8657 \text{ Hz.} \quad (7)$$

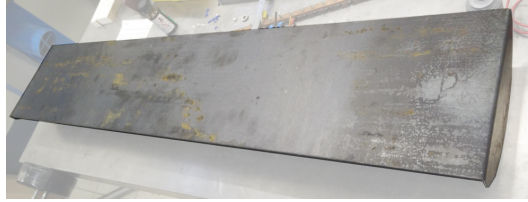
The coincidence frequency, for which the acoustic wavenumber  $k_0$  is equal to the flexural wavenumber in an equivalent plate of thickness  $h$ , can be defined as [18]:

$$f_c = \frac{c_0^2}{2\pi h} \sqrt{\frac{12\rho_s(1-\nu^2)}{E}} = 7812 \text{ Hz.} \quad (8)$$

In general, the radiation efficiency defined in Eq. (5) presents peaks at the ring frequency and at the coincidence frequency [20]. It can be noticed that for this model the ring frequency and the coincidence frequency have close values. The radiation efficiency is expected to be low below these frequencies and to tend to 1 in the supersonic domain above the coincidence frequency.

### 3.2. Non-axisymmetric internal frame

In the second configuration, a non-axisymmetric internal frame is added inside the cylinder described in section 3.1. As shown in Fig. 2, it is a floor which is 869 mm long, 64.3 mm high and 170.6 mm wide. It consists in a 1.5 mm thick plate, for which the ends have been cut along a circular arc of radius 98.5 mm, before being bended to form right angles. The contact between the cylindrical shell and the floor is only on these two circular arcs. A square tube of 20 mm wide and 2mm thick in steel is glued under the horizontal part and reinforces the structure lengthwise.



(a)



(b)

Figure 2: Pictures of the non-axisymmetric internal frame: (a) top view and (b) bottom view.

150 The floor is placed inside the cylindrical shell, with line contacts along the circumference at  $x = 0.315$  m and  $x = 1.185$  m. Several constraints need to be taken into account to assemble the two subsystems:

- the displacements continuity should be verified (*i.e.* rigid connection) and no damping (*i.e.* energy dissipation) should be introduced;
- 155 • the junctions should be strong enough to support the floor when the cylindrical shell is hanged;
- the access inside the cylindrical shell is difficult because of the small radius;
- the process should not damage the cylinder;
- the internal structure could be dismantled once the measurements done.

160 Tests are done on samples made of two steel plates perpendicularly assembled (not presented here for sake of brevity). Several types of glue, as well as brazing are compared. An acrylic glue with activator (Loctite AA 330) is chosen for this application. A picture of the assembled system is shown in Fig. 3.



Figure 3: Interior view of the stiffened cylindrical shell with the non-axisymmetric internal frame.

### 3.3. Experimental setup

165 As shown in Fig. 4, the system is hanged vertically on an arm. A strap is used in order to limit the transfer of vibrations between the system and the arm. A rotating platform allows rotating the cylindrical shell around its axis and identifying precisely the angles. A 12.5 mm diameter patch is glued on the outer surface of the cylindrical shell, and screwed to a shaker. The shaker  
170 is hanged on the rotating platform in order to excite always the same point of coordinate  $(x, r, \theta) = (1.195, 0.1, 0)$  in the cylindrical system (with  $x = 0$  at the bottom of the cylindrical shell). This excitation point is located at the bottom of a stiffener, and lies at 10 mm from one of the floor junctions in



the non-axisymmetric configuration. Two elements come in between the patch  
 175 and the shaker: an impedance head that measures acceleration and force at  
 the excitation point, and a threaded rod that limits the moments and allows  
 assuming a radial excitation in the cylindrical system.

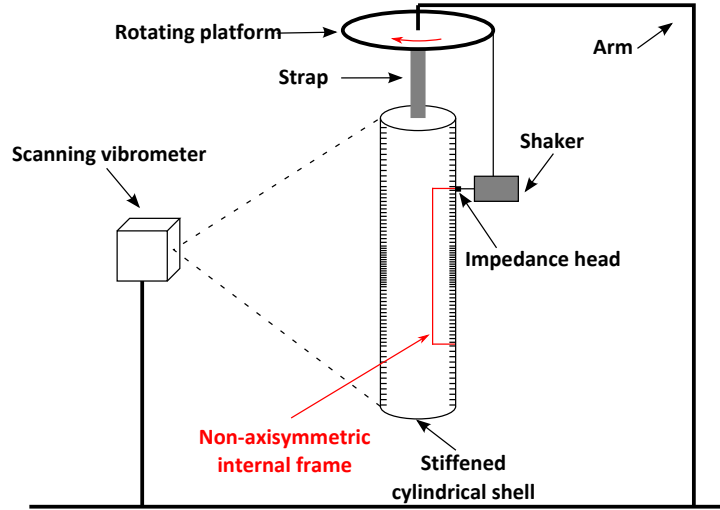


Figure 4: Sketch of the experimental setup.

The measurements are conducted in a semi-anechoic room (8.6 m long, 6.4 m  
 wide, 4 m high and 80 Hz cut-off frequency). The floor is made of concrete and  
 180 a 175 mm space separates it from the bottom of the cylinder. These acoustic  
 conditions are supposed to weakly influence the shell vibrations as the bulk  
 modulus of air is low (*i.e.* weak coupling between the fluid and the structure).  
 The laser vibrometer is placed 2.68 m far from the cylindrical shell's surface  
 as seen in Fig. 5. The vibrometer scans the cylinder from down to up along a  
 185 single generating line. The experiment consists in keeping the vibrometer at a  
 fixed position while rotating the set cylindrical shell-shaker.

The aim is to measure the radial vibrations on the outer surface of the  
 cylindrical shell for the two configurations, that means with and without the  
 non-axisymmetric internal frame. The measurements are made up to 16 kHz.  
 190 The excitation is a chirp (swept-frequency sine), and the output is the transfer  
 function with regard to the excitation force, calculated on 20 averages.

### 3.4. Scanning grid definition

The grid definition in the physical space determines the definition of the  
 wavenumber domain (see the properties of discrete Fourier transform [21]). In  
 195 order to examine the vibrations results in the wavenumber domain, the scanning  
 grid is defined as a function of the flexural wavenumber  $k_f$  of an equivalent plate

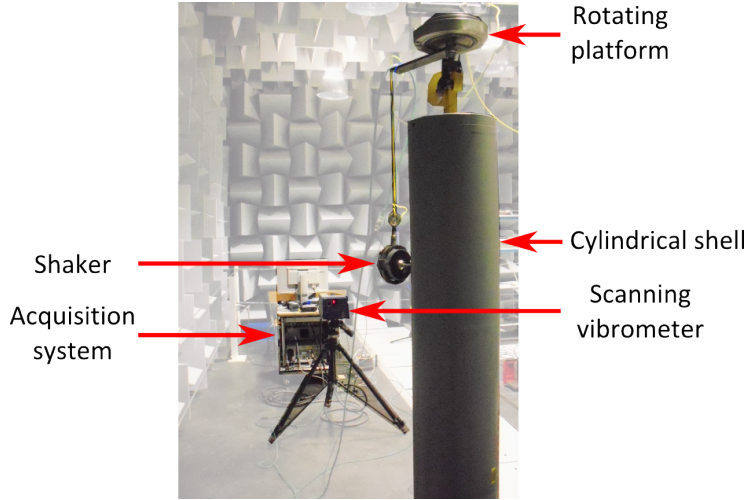


Figure 5: Picture of the experimental setup

(in steel, of thickness  $h = 1.5$  mm) at the highest frequency [18]:

$$k_f = \left( \frac{\omega}{h} \sqrt{\frac{12(1-\nu^2)\rho_s}{E}} \right)^{1/2} \quad (9)$$

To avoid aliasing effect, the maximum axial wavenumber is  $k_x^{max} = k_f \simeq 207 \text{ m}^{-1}$  and the maximum circumferential number is  $n_{max} = k_f R_0 \simeq 20$ .  
 200 For a cylindrical shell, as the system is stiffer than a plate because of the curvature, the natural wavenumbers of the propagative waves are lower than  $k_f$ . The criteria are thus stricter than they could be. The maximum distance  $d_{max}$  between two consecutive measurement points is then:

$$d_{max} = \frac{2\pi}{2k_x^{max}} \quad (10)$$

The numerical application yields  $d_{max} = 15$  mm. A scanning line with  $N_x =$   
 205 101 points along the cylinder is thus defined. The maximum angle  $\theta_{max}$  between two consecutive measurements is given similarly by:

$$\theta_{max} = \frac{2\pi}{2n_{max}} \quad (11)$$

The present case gives  $\theta_{max} = 9^\circ$  and results in  $N_\theta = 41$  lines to scan. The system being symmetric, it is sufficient to measure between 0 and  $180^\circ$ . The experiment consists then in 21 linear scans, where the system is successively  
 210 rotated by  $9^\circ$ .

Similarly, the resolution in the wavenumber domain  $\delta k_x$  is linked to the length of the cylindrical shell:

$$\delta k_x = \frac{2\pi}{L} \simeq 4.2 \quad (12)$$

The axial wavenumber resolution  $\delta k_x$  can be enhanced using zero padding, but results show that the resolution is fine enough to observe the variations with the axial wavenumber.

#### 4. Operational dispersion curves for the axisymmetric system

The 2D Fourier transform defined in Eq. (1) is applied to evaluate experimentally the velocity of the shell in the wavenumber domain. In practice, as the velocities have been measured on a regularly spaced grid, the integrals of Eq. (1) are approximated by the sum of all the points on the grid, weighted by the exponential functions:

$$\tilde{W}(k_x, n) \simeq \sum_{i=1}^{N_x} \sum_{j=1}^{N_\theta} \dot{W}(x_i, \theta_j) e^{-j(k_x x_i + n \theta_j)} d_{max} \theta_{max} \quad (13)$$

where  $(x_i, \theta_j)$  is the coordinate of a point on the scanning grid.

Operational dispersion curves are constructed by plotting the amplitude of the spectral radial velocity on the surface of the axisymmetric cylindrical shell as a function of the frequency and the axial wavenumber, for a given circumferential order. The plots in Fig. 6 show the dispersion curves for the circumferential orders  $n = 0$ ,  $n = 1$ ,  $n = 4$  and  $n = 6$  for the axisymmetric case. Theoretical models are presented in the left column and compared to the experimental results in the right column. The theoretical model uses the Fluegge equations for an infinite cylindrical shell with stiffeners included in the dynamic model using smeared theory [22, 23]. An average stiffeners spacing of 20 mm is considered.

On the theoretical curves, the solid white line represents the flexural wavenumber for an equivalent plate projected on the circumferential order  $n$ :

$$k_{xf}(n) = \sqrt{k_f^2 - \left(\frac{n}{R_0}\right)^2} \quad (14)$$

where  $k_f$  is the flexural wavenumber for a plate defined in Eq. (9). The dashed white line represents the acoustical wavenumber projected on the circumferential order  $n$ :

$$k_{x0}(n) = \sqrt{k_0^2 - \left(\frac{n}{R_0}\right)^2} \quad (15)$$

The area under this dashed curve represents the supersonic domain, where the structural waves are strongly coupled to the acoustic domain and radiate efficiently. For the circumferential orders  $n = 0$  and  $n = 1$ , the dashed-dotted line represents the longitudinal wavenumbers projected on the circumferential order  $n$  given by:

$$k_{xl}(n) = \sqrt{k_l^2 - \left(\frac{n}{R_0}\right)^2} \quad (16)$$

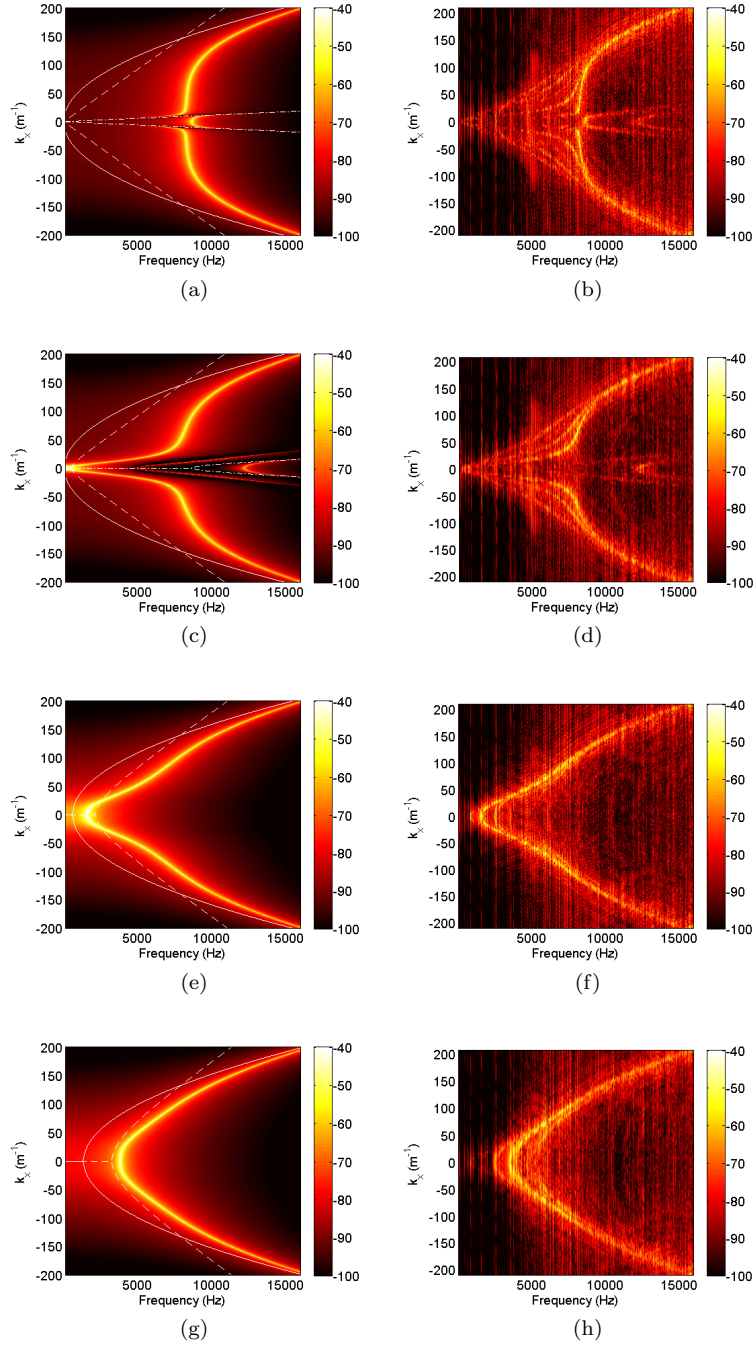


Figure 6: Operational dispersion curves plotted from the spectral radial velocities (dB ref  $1 \text{ m}\cdot\text{s}^{-1}$ ) in the axisymmetric case. (a) Theoretical,  $n = 0$ . (b) Experimental,  $n = 0$ . (c) Theoretical,  $n = 1$ . (d) Experimental,  $n = 1$ . (e) Theoretical,  $n = 4$ . (f) Experimental,  $n = 4$ . (g) Theoretical,  $n = 6$ . (h) Experimental,  $n = 6$ .

where  $k_l = \omega \sqrt{\rho_s(1 - \nu^2)/E}$  is the longitudinal wavenumber. For the circumferential orders  $n = 4$  and  $n = 6$ ,  $k_{xl}$  is not real and is not represented on the plots.

245 For the theoretical model in Fig. 6a, the radial breathing mode  $n = 0$  is plotted. As expected, one can see that the model for a plate is not adapted at low frequencies (the solid white line does not fit the curve), meaning that the curvature of the cylindrical shell has an important effect on the vibroacoustic behavior. The trend for the plate model fits better at higher frequencies (above  
250 12 kHz), where the curvature can be neglected [24]. The breathing mode appears around 8.5 kHz, at the ring frequency [see Eq. (7)]. The coincidence frequency for the equivalent plate is given by the intersection of the flexural wavenumber curve with the acoustical wavenumber [see Eq. (8)] and reads around 7.9 kHz. This value is however not relevant for the present case because of the curvature.  
255 Indeed, as the operational dispersion curve from the cylindrical shell model lies always under the dashed curve, it shows that the breathing mode always radiates efficiently. In Fig. 6a and 6c, the curves seen close to the axis  $k_x = 0$  are propagating with high velocities. They are well approached by the dashed-dotted lines and can be identified as longitudinal waves.

260 For the circumferential orders  $n = 1$  and  $n = 4$  in Fig. 6c and 6e respectively, the effect of curvature can also be seen. The dispersion curve have a S-shape instead of the square-root shape for the flexural waves in a plate. The coincidence frequency for the cylindrical shell model can then be read around 2.7 kHz, instead of 8 kHz for the plate. Cut-on frequencies can also be seen on  
265 the operational dispersion curves. For instance, it can be seen that the waves cannot propagate for the circumferential order  $n = 4$  below 1500 Hz. It can be checked that the cut-on frequency increases with the circumferential order. For the circumferential order  $n = 6$  in Fig. 6g, the cut-on frequency is around 3700 Hz. For the circumferential order  $n = 6$ , the effect of curvature is less  
270 important and the dispersion curve is closer to the plate model. However, it can be seen that the flexural waves are faster than the plate model shown by the white solid line, due to added stiffness because of the curvature and the stiffeners. Consequently, the whole dispersion curve is in the supersonic domain and this circumferential order always radiates efficiently.

275 Comparing the experimental and theoretical operational dispersion curves in Fig. 6 shows a good agreement between the curves. On the experimental plots, spots of high amplitudes can be seen instead of continuous lines. These spots correspond to axial resonances due to the fact that the experimental cylindrical shell is of finite length. However, the comparison also shows that it is difficult to  
280 isolate the contribution from only one circumferential order in the experimental case. Additional dispersion curves can indeed be seen, even if their levels are lower than the main contribution. Three explanations are considered:

- As the cylindrical shell has sections with three different stiffeners spacings (see Fig. 1c), it could explain why several dispersion curves can be seen  
285 for one circumferential order. However, numerical tests show that the influence of the stiffeners spacing is weak on the dispersion curves and can

only be seen at high frequencies.

- Uncertainties in the position of the points on the scanning grid can occur. These uncertainties can break the regularity of the measured velocity field and couple the circumferential orders. However, these uncertainties have been numerically tested on an infinite shell model and a equally distributed law on the points of the scanning grid (not shown here for the sake of conciseness). It can be seen that their influence is weak on the coupling of the circumferential orders.
- Although the particular experimental care, the system and the excitation are not perfectly axisymmetric. Indeed, it is really hard in practice to be sure that the shaker only excites the cylindrical shell in a perfectly radial direction. Besides there may be imperfections on the cylindrical shell geometry because of the machining (0.1 mm tolerance on the shell thickness for instance) and the damages during transportation. It is known that non-axisymmetries tend to couple circumferential orders [13].

This suggests that the additional dispersion curves seen in the right column of Fig. 6 are a coupling of the circumferential orders due to imperfections in the axisymmetry of the system.

## 5. Influence of the non-axisymmetric internal frame

### 5.1. Mean quadratic velocity

The mean quadratic velocity defined in Eq. (6) is written in a discrete form as

$$\langle \dot{W} \rangle^2 = \frac{1}{S} \sum_{i=1}^{N_x} \sum_{j=1}^{N_\theta} |\dot{W}(x_i, \theta_j)|^2 d_{max} R_0 \theta_{max} \quad (17)$$

The mean quadratic velocity is plotted as a function of the frequency in Fig. 7a for the axisymmetric case. Only the frequency range between 0 and 2500 Hz is shown in this plot in order to clearly see the resonance peaks. By looking at the velocity repartition on the surface of the cylindrical shell for the resonance peaks, one can notice that the circumferential modes appear by groups. This phenomenon is linked to the fact that there is a cut-on frequency below which the circumferential orders cannot appear, as explained in [25] and on the dispersion curves in section 4.

Before placing the non-axisymmetric internal frame in the cylinder, it was hanged with flexible strings and an accelerometer was placed at an arbitrary spot in order to measure the frequency response to an impact hammer. The transfer function is plotted in Fig. 7b. This plot allows estimating qualitatively the number of resonances of the isolated non-axisymmetric internal frame, which appears high in comparison with the number of resonances of the axisymmetric cylindrical shell in the frequency range considered (*cf.* Fig. 7a).

The comparison of the mean quadratic velocities for the two configurations (with and without the non-axisymmetric internal frame) is plotted as a function

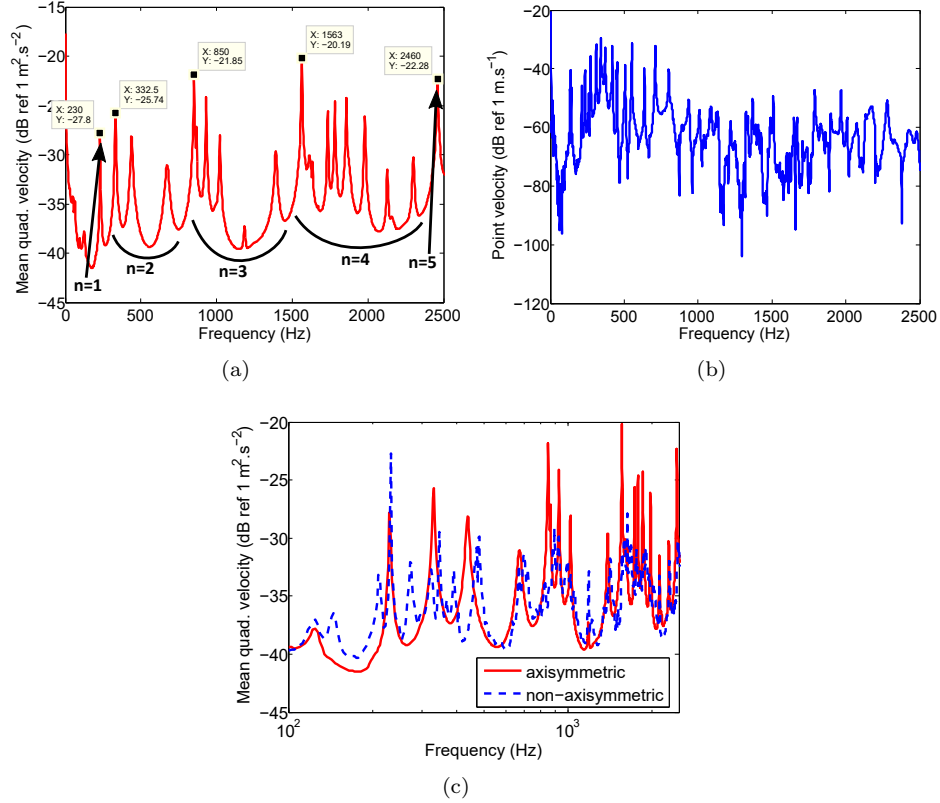


Figure 7: (a) Mean quadratic velocity as a function of the frequency for the axisymmetric stiffened cylindrical shell. (b) Transfer function between an impact point on the top of the floor and an accelerometer placed on the side. (c) Comparison of the mean quadratic velocities for the two configurations.

of the frequency in Fig. 7c. The first observation is that the vibration level is lower in the non-axisymmetric case, in particular above 500 Hz. The cylindrical shell and the internal structure have the same thickness and are made from the same material. We can expect that the mechanical impedances of the two structures are similar and then adapted to exchange vibratory energy from one structure to another. Besides, the modal density of the internal structure is higher than the one of the cylindrical shell. The combination of these two elements leads to an apparent damping effect on the shell because of energy being trapped in the internal frame. Moreover, new resonance peaks can be observed, in particular at low frequency between 200 and 500 Hz. The peaks frequencies (at 213, 234 and 273 Hz for instance) are directly linked to the resonance of the non-axisymmetric internal frame seen in Fig. 7b.

## 5.2. Radiated power and radiation efficiency

The stationary phase theorem given in Eq. (3) is used to calculate the pressure radiated in the far-field using the vibrations measurements on the cylindrical shell in the wavenumber domain. In practice, the sum is calculated for  $n$  between  $-n_{max}$  and  $n_{max}$ . The far-field radiated power is deduced by discretization of Eq. (4):

$$P_a = \frac{1}{\rho_0 c_0} \sum_{i_\xi=1}^{N_\xi} \sum_{j_\theta=1}^{N_\theta} p^2(\mathcal{R}, \xi_i, \theta_j) \mathcal{R}^2 \sin \theta_j \delta \xi \theta_{max} \quad (18)$$

where  $(\xi_i)_{i=1,2,\dots,N_\xi}$  is the vector of azimuthal angles and  $\delta \xi$  the step between two consecutive angles. For this example  $\xi_i$  is varying between  $-187^\circ$  and  $+187^\circ$  with a step of  $\delta \xi = 3^\circ$

The radiated power calculated from the spectral displacements and the stationary phase theorem is plotted as a function of the frequency for the axisymmetric and non-axisymmetric case in Fig. 8a. The influence of the non-axisymmetric internal frame is mainly seen in the low frequencies (below 1 kHz). The radiation efficiency [see Eq. (5)] is plotted in Fig. 8b. The coincidence frequency for the equivalent plate ( $h = 1.5$  mm) is plotted with a vertical dotted line. It is clear that the non-axisymmetric internal frame tends to increase the radiation efficiency below the coincidence frequency. Differences up to 10 dB can be seen at some frequencies on this configuration. Above the coincidence frequency, it can be said that the structure radiates efficiently with or without the non-axisymmetry.

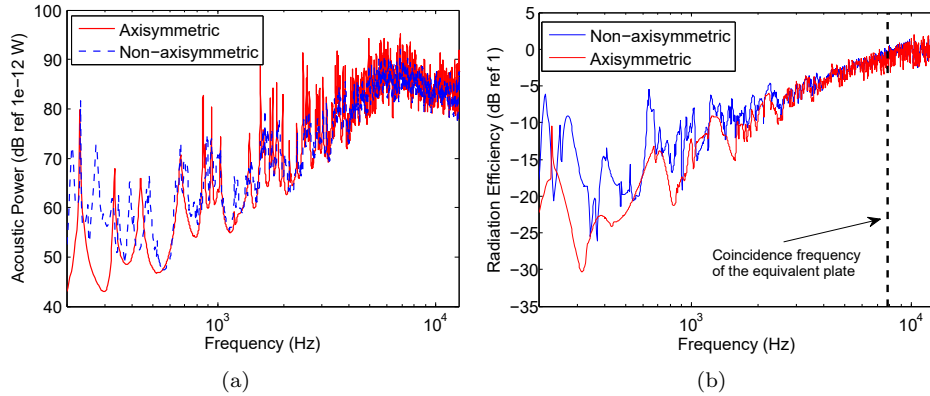


Figure 8: Estimation as a function of the frequency for the two configurations using the stationary phase theorem of (a) the radiated power (dB ref  $1e-12$  W) and (b) the radiation efficiency (dB ref 1).

To explain the increase in the radiation efficiency, a focus is done on the operational dispersion curves in the frequency range up to 3000 Hz, where the radiation efficiency increases in the non-axisymmetric case. The experimental



operational dispersion curves for this frequency range are plotted for the axisymmetric and non-axisymmetric case in Fig. 9 for the circumferential orders  $n = 0$  and  $n = 4$ . It can be seen that in the non-axisymmetric case, the contribution of the circumferential order  $n = 0$  is higher than in the axisymmetric case, while the amplitude of the waves of circumferential order  $n = 4$  is lower. It means that there is a coupling between the circumferential orders because of the internal frame, and that energy is transmitted from higher circumferential orders to lower ones. Besides, it has been shown in Fig. 6a that the circumferential order  $n = 0$  is efficiently radiating, even in the low frequency range. This explains why the cylindrical shell radiates more efficiently when it is coupled to a non-axisymmetric internal frame. It confirms the observations from numerical simulations [12]. In the next section, a closer look is taken at the velocity field to understand the influence of the non-axisymmetric internal frame on the repartition of the vibratory energy.

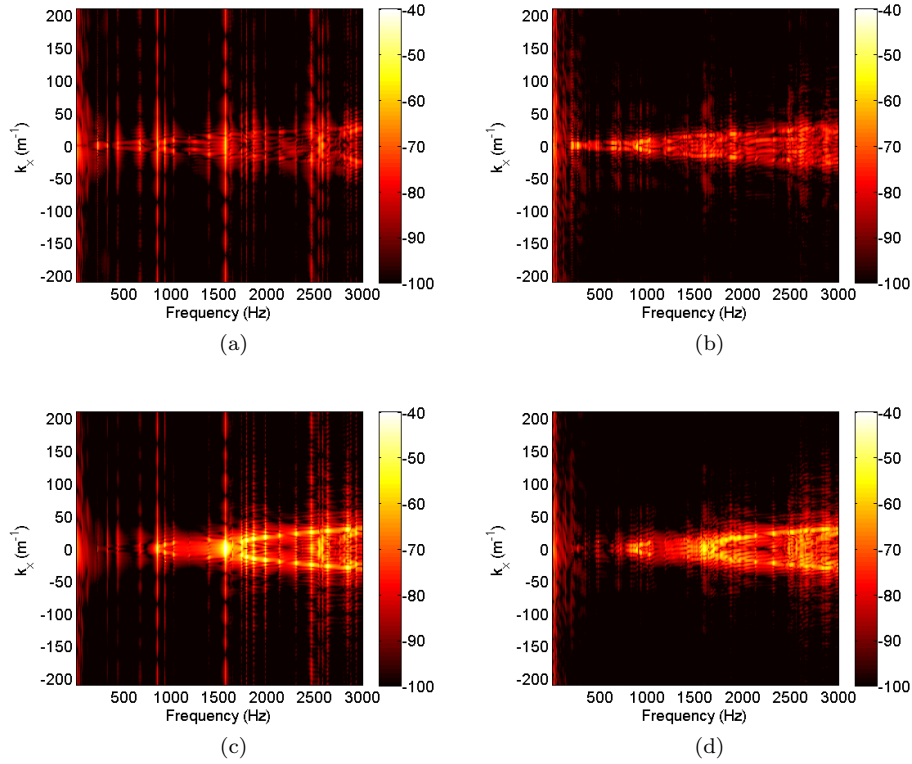


Figure 9: Operational dispersion curves plotted from the spectral radial velocities (dB ref 1  $m.s^{-1}$ ) in the low frequency range. (a) Axisymmetric case,  $n = 0$ . (b) Non-axisymmetric case,  $n = 0$ . (c) Axisymmetric case,  $n = 4$ . (d) Non-axisymmetric case,  $n = 4$

## 375 6. Maps of the radial vibratory field

The velocity maps on the outer surface of the cylindrical shell are plotted in Fig. 10 for the axisymmetric and the non-axisymmetric configurations, at  $f = 1980$  Hz and  $f = 3580$  Hz. These frequencies are chosen because they show an increase in the radiation efficiency in the non-axisymmetric case (2.8 dB at 1980 Hz and 1.9 dB at 3580 Hz) while having a drop in the radiated power (from 82.8 dB to 66.5 dB at 1980 Hz and from 82.5 dB to 80.4 dB at 3580 Hz). Besides, energy transfer through the internal structure can be observed. Looking at the frequency response in Fig. 7a, it can be said that the axisymmetric cylindrical shell has a resonance at 1980 Hz and has no resonance at 3580 Hz (out of the range in Fig. 7a, but not on a resonance peak). The plots in Fig. 10 are functions of the tangential coordinate  $\theta$  in abscissa and the axial coordinate  $x$  in ordinate. The excitation is placed in  $\theta = 0$  and  $x = 1195$  mm. A spot of low velocities can be seen around the excitation, because the vibrometer cannot scan the surface behind the shaker. At  $f = 1980$  Hz, the mode  $(m, n) = (8, 4)$  is observed for the axisymmetric case, where  $m$  is the longitudinal modal index and  $n$  the circumferential order. Adding the non-axisymmetric internal structure has two main influences on the cartography at this frequency: the amplitude decreases and the nodes of vibrations are modified.

In Fig. 10c it can be seen that the point force mainly excites the area close to the excitation point. One of the possible explanations is that the cylindrical shell has sections with different stiffeners spacings. This means that there is a mechanical impedance discontinuity between two consecutive sections, which explains the localization effect seen at  $f = 3580$  Hz. In the non-axisymmetric case (Fig. 10d), the energy can be transmitted through the internal structure to the other end of the cylinder, where the stiffeners spacing is the same than in the section which is excited (see Fig. 1c).

To see the role played by the low circumferential orders, the radial velocities are plotted in the wavenumber space for the two configurations and the two frequencies in Fig. 11. In each plot, the amplitude has been normalized to the highest value in order to compare the repartition of the radial velocities in the wavenumber space. At  $f = 1980$  Hz, the amplitude maxima appear in the axisymmetric case at the wavenumbers corresponding to the modal wavenumbers. The circumferential order  $n = 4$  is playing a leading role in this case. In the non-axisymmetric case, more circumferential orders play a role, especially in the low circumferential orders. The same observations can be made at  $f = 3580$  Hz in Fig. 11c and 11d.

The radiation circle defined by Eq. (2) is plotted in white on the plots in Fig. 11. The radiation circles are elliptical due to the plotting aspect ratio. It can be noticed that the components inside the circle, which contribute to the far-field radiated pressure, have a higher relative level in the non-axisymmetric case. The radiation efficiency is thus expected to increase at these frequencies, as shown in section 5.2.

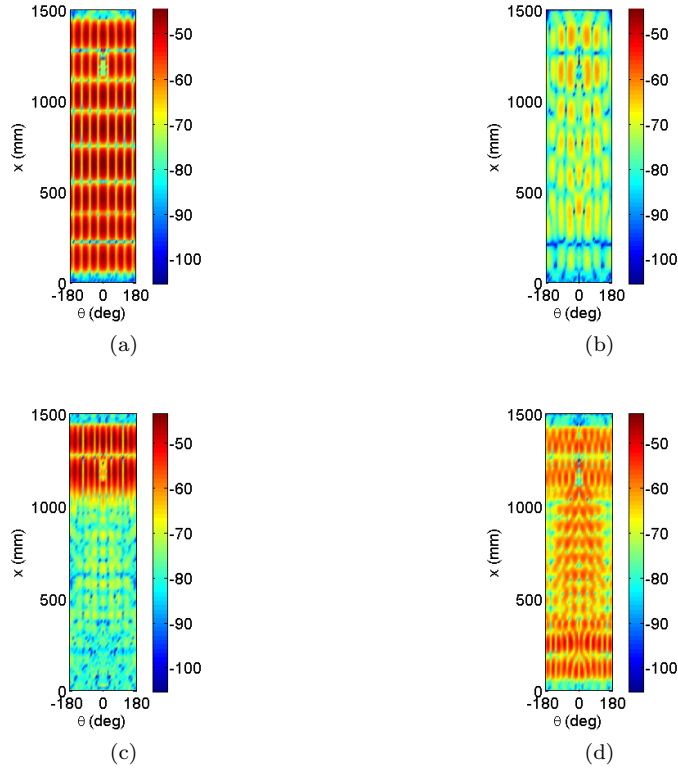


Figure 10: Map of radial velocities (dB ref  $1 \text{ m.s}^{-1}$ ) on the outer surface of the cylindrical shell. (a) Asymmetric case at  $f = 1980 \text{ Hz}$ . (b) Non-axisymmetric at  $f = 1980 \text{ Hz}$ . (c) Asymmetric case at  $f = 3580 \text{ Hz}$ . (d) Non-axisymmetric at  $f = 3580 \text{ Hz}$ .

## 7. Conclusions

Experiments have been conducted on a stiffened cylindrical shell to investigate the effect of a non-axisymmetric internal frame. A scanning laser vibrometer has been used to measure the radial velocities on a grid on the outer surface of the point-driven cylindrical shell, with and without a non-axisymmetric internal frame. A wavenumber analysis of the vibrations of the shell is used to discuss the vibroacoustic behavior of the shell. It shows that more circumferential orders tend to play a role when the cylindrical shell is coupled to the non-axisymmetric internal frame. More particularly energy is transmitted from high to low circumferential orders. As the low wavenumbers are in the supersonic domain, the structure tends to radiate more efficiently. This phenomenon explains the increase in radiation efficiency below the coincidence frequency seen in the non-axisymmetric case.

Further work consists in conducting the same experimental process at higher frequencies. It will allow studying the influence of the non-axisymmetric internal

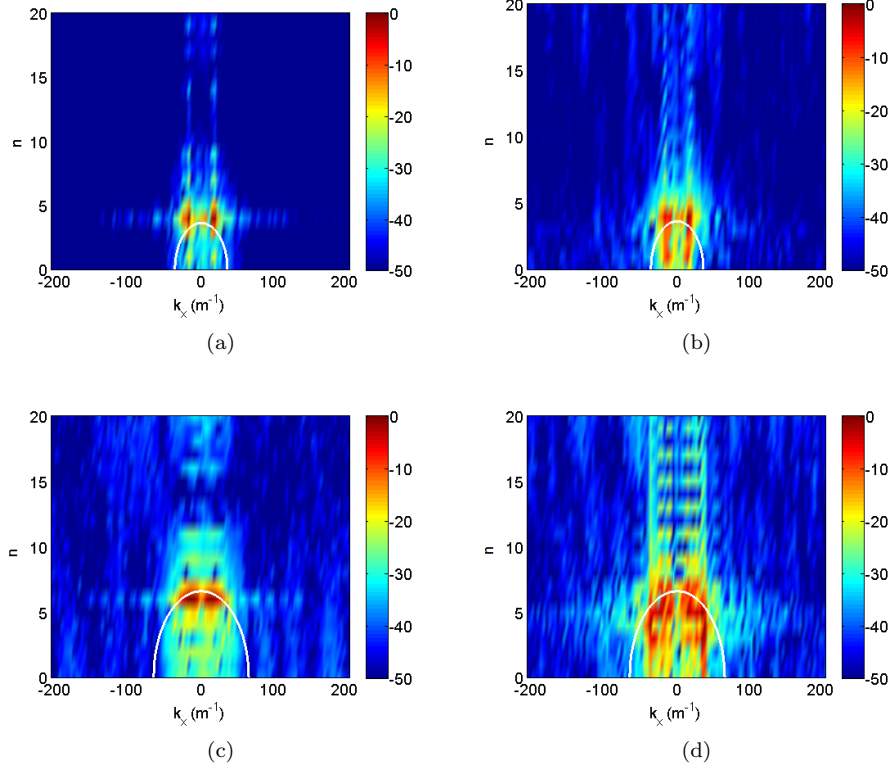


Figure 11: Map of radial velocities (dB ref 1 m.s<sup>-1</sup>) in the wavenumber domain ( $k_x, n$ ). (a) Asymmetric case at  $f = 1980$  Hz. (b) Non-axisymmetric at  $f = 1980$  Hz. (c) Asymmetric case at  $f = 3580$  Hz. (d) Non-axisymmetric at  $f = 3580$  Hz.

frame on the propagation of Bloch-Floquet waves due to periodic stiffeners. Besides, if the model is submerged in water, the coincidence frequency is about  
 435 20 times higher. In this case, it is likely that the effect of the internal frame is seen on a wider frequency range. Moreover, the strong coupling with the fluid can be investigated.

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