



HAL
open science

A reduction technique for mistuned bladed disks with superposition of large geometric mistuning and small model uncertainties

Vladislav Ganine, Mathias Legrand, Christophe Pierre, Hannah Michalska

► To cite this version:

Vladislav Ganine, Mathias Legrand, Christophe Pierre, Hannah Michalska. A reduction technique for mistuned bladed disks with superposition of large geometric mistuning and small model uncertainties. 12th International Symposium on Transport Phenomena and Dynamics of Rotating Machinery, Feb 2008, Honolulu, United States. hal-01393687

HAL Id: hal-01393687

<https://hal.science/hal-01393687v1>

Submitted on 12 Nov 2016

HAL is a multi-disciplinary open access archive for the deposit and dissemination of scientific research documents, whether they are published or not. The documents may come from teaching and research institutions in France or abroad, or from public or private research centers.

L'archive ouverte pluridisciplinaire **HAL**, est destinée au dépôt et à la diffusion de documents scientifiques de niveau recherche, publiés ou non, émanant des établissements d'enseignement et de recherche français ou étrangers, des laboratoires publics ou privés.



Distributed under a Creative Commons Attribution 4.0 International License

A REDUCTION TECHNIQUE FOR MISTUNED BLADED DISKS WITH SUPERPOSITION OF LARGE GEOMETRIC MISTUNING AND SMALL MODEL UNCERTAINTIES

^aVladislav Ganine*, ^aMathias Legrand, ^aChristophe Pierre, and ^bHannah Michalska

^aStructural Dynamics & Vibration Laboratory, McGill University Montreal QC H3A 2K6, Canada

^bDepartment of Electrical, & Computer Engineering, McGill University, Montreal QC H3A 2A7, Canada

vladislav.ganine@mail.mcgill.ca, mathias.legrand@mcgill.ca,

christophe.pierre@mcgill.ca, hannah.michalska@mcgill.ca

ABSTRACT

A new method for vibration analysis of mistuned bladed disks is presented. The method combines two previously reported modeling techniques in order to study the effects of small random parameter variation on geometrically mistuned bladed disks. It is based on the observation that the nominal projections usually possess a certain degree of robustness tolerating small perturbations. Hence the subspace spanned by a sufficient number of compensated tuned system normal modes can be used to repeatedly project models with small random parameter variation. The method is validated numerically on an industrial bladed disk model, both free and forced responses are compared with full model finite element analysis. The benchmark cases show acceptable accuracy while retaining low computational cost to build and evaluate obtained ROM.

INTRODUCTION

Bladed disk assemblies belong to a class of rotationally periodic systems in which cyclic symmetry is usually exploited to predict vibrational response because it reduces the size of a problem to a single sector. However, the cyclic symmetry is destroyed by blade-to-blade geometrical and/or mechanical properties variations due to manufacturing tolerances and operational wear and tear, which is commonly referred to as mistuning. The impact of mistuning on a periodic system sums down to strong mode localization and significantly larger amplitudes of vibration response unlike

those of the nominal system and as a consequence their inadequate prediction when using cyclic symmetry. The stochastic nature of blade to blade properties variations requires Monte Carlo simulation to quantitatively assess the effect of mistuning. In this connection model order reduction of mistuned bladed disks plays an important role to save the computational effort.

The problem of approximating high order finite element models by lower order ones received a substantial amount of interest over the years. Most of the reduction methods for bladed rotors involve projection into a lower order subspace. Methods that follow that general structure [Castanier et al., 1997, Bladh et al., 2001a, Bladh et al., 2001b, Bladh et al., 2002, Lim et al., 2003, Lim et al., 2004, Yang and Griffin, 2001] constitute modal truncation, where state transformation into a realization in which the state variables can be ranked according to some measure of importance, i.e. natural frequencies falling into expected excitation frequency band, followed by truncation of the least important ones. Unlike others, [Petrov et al., 2002] used Woodbury formula to calculate the inverse of modified by mistuning symmetric FRF (Frequency Response Function) matrix.

Beyond the reduction approach the models can also be classified in a way they introduce mistuning. Here the substructuring of a system into blade and disk components is a primary technique to conveniently introduce measurable blade to blade properties variations. [Castanier et al., 1997] applied component mode synthesis technique modifying blade natural frequencies that appear explicitly in the synthesized matrices. [Bladh et al., 2001a, Bladh et al.,

*Address all correspondence to this author.

2001b, Bladh et al., 2002] projected mistuning onto a selected set of tuned blade-alone modes, which allowed to capture nonproportional blade to blade variations in a computationally efficient way. [Lim et al., 2003] extended this approach and obtained a more compact ROM (Reduced Order Model) with CMM (Component Mode Mistuning) technique without component based representation of the reduced system where mistuning is projected directly on to a set of tuned system modes through mode participation factors. Although [Yang and Griffin, 2001] were first to express mistuned system normal modes as a linear combination of tuned ones, their approach required the knowledge of mistuned system stiffness/mass matrices. Thus they only simulated proportional mistuning by perturbing the Young's moduli of individual blades.

Even if using this projection yields accurate low order models, the number of tuned system modes required to capture the effects of large geometric mistuning (e.g. blade geometry change) render them computationally impractical. For that reason [Lim et al., 2004] developed a new SMC technique that generates models of commensurable size in the presence of large geometric mistuning. Later [Sinha, 2007] presented MMDA (Modified Modal Domain Analysis) method to deal with geometric mistuning problem yet his ROM are of significantly higher order - number of tuned system modes kept times number of POD features.

The current paper combines the results of two publications [Lim et al., 2003, Lim et al., 2004] and addresses the issue of how to efficiently carry out a Monte Carlo simulation in order to investigate the small mistuning-dependent performances with a reduced order model retaining the same set of mistuning parameters as those in the original FEM in the presence of deterministic geometric mistuning. We extend the original SMC (Static Mode Compensation) formulation to a non symmetric multiple blade geometry mistuning and demonstrate that CMM and SMC techniques can be effectively combined in a computationally inexpensive way to address the needs of design community.

This paper is organized as follows. The theoretical basis for the reduced order model formulation is presented in Section 2. In Section 3, the extensive numerical studies with an industrial bladed disk FEM are given. In particular, a numerical comparison with an exact solution using full FEM with different geometrical mistuning patterns is carried out (eigenvalues, mode shapes and forced response). The last section of the paper presents a comprehensive assessment of the numerical properties of the method proposed here.

SELECTED NOMENCLATURE

Λ^S	Diagonal matrix of nominal system eigenvalues
μ_I^{syn}	Reduced mass matrix with large mistuning only
Φ^S	Nominal modal matrix
Φ_n^B	Cantilevered-blade normal modes of the n -th blade
$\Psi_n^{B,M}$	Modes that describe n -th blade motion at the bound-

	ary, mass mistuning
ω_c^2	Eigenvalue shift
\otimes	Kronecker product
\mathbf{F}^Q	Forcing terms due to mistuning
$\mathbf{G}_{\Gamma_k}^Q$	k -th blade portion of equivalent forcing terms due to mistuning
\mathbf{K}	Nominal stiffness matrix
\mathbf{M}	Nominal mass matrix
$\mathbf{M}_n^{\delta L}$	n -th blade mass matrix deviation due to large mistuning
$\mathbf{M}_n^{\delta S}$	n -th blade mass matrix deviation due to small mistuning
$\mathbf{q}_{\Phi,n}^M$	Participation factors of the cantilevered-blade modes in the n -th blade portion of compensated nominal modes, mass mistuning
$\mathbf{q}_{\Phi,n}^M$	Participation factors of the boundary modes in the n -th blade portion of compensated nominal modes, mass mistuning
$\tilde{\mathbf{B}}\text{diag}[\cdot]$	A pseudo-block-diagonal matrix, where the size of $h=1,\dots,H$ a block changes according to harmonic h
$\hat{\Phi}$	Mistuned system modal matrix
$\hat{\lambda}_i$	Mistuned system eigenvalue
$\tilde{\Psi}_{\Gamma}^{S,Q}$	Blade portion of nominal quasi-static modes matrix in cyclic coordinates
\mathcal{F}_k	k -th row of the real-valued Fourier matrix

THEORY

Consider a finite element model of an undamped structure around a position of equilibrium in frequency domain

$$(-\omega^2 \mathbf{M} + \mathbf{K})\mathbf{x}(\omega) = \mathbf{f}(\omega) \quad (1)$$

where $\mathbf{x} = (x_1, \dots, x_n)$ is the displacement vector of n degrees of freedom, $\mathbf{f} = (f_1, \dots, f_n)$ is a vector of harmonic excitation forces, \mathbf{M} and \mathbf{K} are real symmetric definite mass and stiffness matrices, respectively.

The objective of the modal superposition reduction procedure is to determine an r -th order reduced-space basis, i.e. a set of modeshapes corresponding to contiguous set of natural frequencies within expected excitation frequency band, onto which the state vector can be projected

$$\begin{aligned} (-\omega^2 \mathbf{M}_r + \mathbf{K}_r)\mathbf{x}_r(\omega) &= \mathbf{f}_r(\omega) \\ \mathbf{M}_r &= \Phi^{S^T} \mathbf{M} \Phi^S, & \mathbf{K}_r &= \Phi^{S^T} \mathbf{K} \Phi^S \\ \mathbf{f}_r(\omega) &= \Phi^{S^T} \mathbf{f}(\omega), & \mathbf{x}_r(\omega) &= \Phi^{S^T} \mathbf{x}(\omega) \end{aligned} \quad (2)$$

Typically, the modal truncation model reduction approach is accompanied by projection of the mistuned matrices onto the same truncated set of nominal system normal modes Φ^S ,

which leads to a mistuned ROM eigenvalue problem:

$$\Phi^{S^T} (\mathbf{K} + \mathbf{K}^\delta) \Phi^S \mathbf{y} = \Lambda^m \Phi^{S^T} (\mathbf{M} + \mathbf{M}^\delta) \Phi^S \mathbf{y} \quad (3)$$

Although the procedure is relatively straightforward yielding very compact low order models when applied to bladed disk assemblies, it does not extend to the large geometric mistuning problem. Essentially, the precision in that case is limited by the amplitude of variation in mass and stiffness matrices \mathbf{M}^δ and \mathbf{K}^δ due to mistuning. If it is small compared to nominal properties, the reduced model natural frequencies and modeshapes will be close to those of the full system. It is well known [Stewart and Sun, 1990] that the spectrum of the matrix pencil

$$(\Phi^{S^T} (\mathbf{K} + \mathbf{K}^\delta) \Phi^S, \Phi^{S^T} (\mathbf{M} + \mathbf{M}^\delta) \Phi^S)$$

perturbed from (Λ^S, \mathbf{I}) is equivalent to that of the matrix pencil

$$(\Phi^{S^T} (\mathbf{K} + \mathbf{K}^\delta) \Phi^S + O(\Delta^2), \Phi^{S^T} (\mathbf{M} + \mathbf{M}^\delta) \Phi^S + O(\Delta^2))$$

up to the terms in $\Delta^2 = \|\mathbf{K}^\delta\|^2 + \|\mathbf{M}^\delta\|^2$. To address this issue, [Yang and Griffin, 2001] discuss the possibility of using the modes of a geometrically mistuned system as nominal modes to project additional random mistuning. Clearly, for very large systems that will be computationally infeasible, given that one must repeatedly solve the generalized eigenvalue problem for a full non symmetrical system. Here we present an alternative approach that does not require full system analysis: by using SMC method we calculate the compensated tuned system normal modes that accurately approximate those of a full system featuring large geometric mistuning, upon which we project small random mistuning using CMM.

Algorithm Formulation

The combined model reduction algorithm can be summarized as follows:

1. Use SMC method to obtain a set of compensated basis vectors $(\Phi_\Gamma^S - \Psi_\Gamma^{S,Q} \mathbf{G}_\Gamma^Q)$.
2. Project large mistuning $\mathbf{K}^{\delta L}$ and $\mathbf{M}^{\delta L}$ onto the subspace spanned by the basis vectors.
3. Calculate mode participation factors of CMM method using $(\Phi_\Gamma^S - \Psi_\Gamma^{S,Q} \mathbf{G}_\Gamma^Q)$.
4. Project small mistuning $\mathbf{K}^{\delta s}$ and $\mathbf{M}^{\delta s}$.
5. Assemble total reduced order model mass and stiffness matrices $\mu_L^{syn} + \mu_S^{syn}$ and $\kappa_L^{syn} + \kappa_S^{syn}$.
6. Solve the reduced system.

Calculation of the compensated basis vectors

[Lim et al., 2004] used Modal Acceleration Method to modify tuned system normal modes in such a way so they are closer to the mistuned system modes. In so doing one needs much fewer compensated modes to approximately span the eigenspace of geometrically mistuned system

$$\Phi^S - ((\mathbf{K} + \mathbf{K}^{\delta L}) - \omega_c^2 (\mathbf{M} + \mathbf{M}^{\delta L}))^{-1} \mathbf{F}(\omega) = \sum_{i=1}^N \left(\frac{\omega^2 - \omega_c^2}{\hat{\lambda}_i - \omega_c^2} \right) \frac{\hat{\Phi}_i^T \mathbf{F}(\omega)}{\hat{\lambda}_i - \omega^2} \hat{\Phi}_i \quad (4)$$

Notice that ω correspond to nominal system natural frequencies Λ^S and $\mathbf{F}(\omega)$ is an equivalent to geometric mistuning forcing excitation matrix defined as

$$\mathbf{F}^Q = (-[\Lambda^S - \omega_c^2 \mathbf{I}] [\mathbf{M} + \mathbf{M}^{\delta L}] + [\mathbf{K} + \mathbf{K}^{\delta L}] - \omega_c^2 [\mathbf{M} + \mathbf{M}^{\delta L}]) \Phi^S = \left\{ \begin{array}{c} \mathbf{0} \\ (-[\Lambda^S - \omega_c^2 \mathbf{I}] \mathbf{M}^{\delta L} + \mathbf{K}^{\delta L} - \omega_c^2 \mathbf{M}^{\delta L}) \Phi_\Gamma^S \end{array} \right\} \quad (5)$$

Rather than performing full matrix inversion, the quasi-static compensation terms can be obtained much more efficiently by taking advantage of the number of zero terms in the perturbation matrices $\mathbf{M}^{\delta L}$ and $\mathbf{K}^{\delta L}$

$$(\mathbf{K} + \mathbf{K}^{\delta L} - \omega_c^2 (\mathbf{M} + \mathbf{M}^{\delta L}))^{-1} = [\mathbf{K} - \omega_c^2 \mathbf{M}]^{-1} (\mathbf{I} + [\mathbf{K}^{\delta L} - \omega_c^2 \mathbf{M}^{\delta L}] [\mathbf{K} - \omega_c^2 \mathbf{M}]^{-1})^{-1} \quad (6)$$

Then the equivalent force can be expressed as

$$\mathbf{G}_\Gamma^Q = \left\{ \begin{array}{c} \mathbf{0} \\ (\mathbf{I} + [\mathbf{K}^{\delta L} - \omega_c^2 \mathbf{M}^{\delta L}] \Psi_\Gamma^{S,Q})^{-1} \mathbf{F}_\Gamma^Q \end{array} \right\} \quad (7)$$

In order to efficiently calculate the quasi-static modes $\Psi_\Gamma^{S,Q}$, we use the cyclic symmetry domain, where $[\mathbf{K} - \omega_c^2 \mathbf{M}]^{-1}$ is inverted blockwise

$$\tilde{\Psi}_\Gamma^{S,Q} = \tilde{\mathbf{B}} \text{diag} [(\tilde{\mathbf{K}}_h - \omega_c^2 \tilde{\mathbf{M}}_h)_\Gamma^{-1}]_{h=1, \dots, H} \quad (8)$$

Clearly, in physical domain we only need those partitions of $\Psi_\Gamma^{S,Q}$ that correspond to mistuned DOF of each blade

$$\Psi_{k,m}^{S,Q} = (\mathcal{F}_k \otimes \mathbf{I}) \tilde{\Psi}_\Gamma^{S,Q} (\mathcal{F}_m^T \otimes \mathbf{I}) \quad (9)$$

Tuned blades are numbered $k = 1, \dots, N$, mistuned $m = 1, \dots, K$, where N and K denoting the total number and the number of mistuned blades correspondingly. Note that \mathcal{F}_k is referred to k -th row of real-valued Fourier matrix.

Large mistuning projection

The reduced order model mass and stiffness matrices due to geometric mistuning are given by

$$\begin{aligned}\boldsymbol{\mu}_L^{syn} &= (\boldsymbol{\Phi}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q)^T (\mathbf{M} + \mathbf{M}^{\delta_L}) (\boldsymbol{\Phi}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q) \\ &= \mathbf{I} + \boldsymbol{\mu}_1^{syn} - \boldsymbol{\mu}_2^{syn} - \boldsymbol{\mu}_2^{synT} + \boldsymbol{\mu}_3^{syn} \\ \boldsymbol{\kappa}_L^{syn} &= (\boldsymbol{\Phi}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q)^T (\mathbf{K} + \mathbf{K}^{\delta_L}) (\boldsymbol{\Phi}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q) \\ &= \mathbf{I} + \boldsymbol{\kappa}_1^{syn} - \boldsymbol{\kappa}_2^{syn} - \boldsymbol{\kappa}_2^{synT} + \boldsymbol{\kappa}_3^{syn}\end{aligned}\quad (10)$$

Those products can be taken on a sector-by-sector basis i.e.,

$$\begin{aligned}\boldsymbol{\mu}_1^{syn} &= \sum_{k=1}^K \boldsymbol{\Phi}_{\Gamma_k}^{ST} \mathbf{M}_k^{\delta_L} \boldsymbol{\Phi}_{\Gamma_k}^S \\ \boldsymbol{\mu}_2^{syn} &= \sum_{k=1}^K \left((\boldsymbol{\Lambda}^S - \omega_c^2 \mathbf{I})^{-1} \boldsymbol{\Phi}_{\Gamma_k}^{ST} \mathbf{G}_{\Gamma_k}^Q + \boldsymbol{\Phi}_{\Gamma_k}^{ST} \mathbf{M}_k^{\delta_L} \sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right) \\ \boldsymbol{\mu}_3^{syn} &= \sum_{k=1}^K \left(\left[\sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right]^T \mathbf{M}_k^{\delta_L} \sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right. \\ &\quad \left. + \sum_{m=1}^K \mathbf{G}_{\Gamma_m}^{QT} [\boldsymbol{\Psi}^{S,QT} \mathbf{M} \boldsymbol{\Psi}^{S,Q}]_{m,k} \mathbf{G}_{\Gamma_k}^Q \right) \\ \boldsymbol{\kappa}_1^{syn} &= \sum_{k=1}^K \boldsymbol{\Phi}_{\Gamma_k}^{ST} \mathbf{K}_k^{\delta_L} \boldsymbol{\Phi}_{\Gamma_k}^S \\ \boldsymbol{\kappa}_2^{syn} &= \sum_{k=1}^K \left(\boldsymbol{\Lambda}^S (\boldsymbol{\Lambda}^S - \omega_c^2 \mathbf{I})^{-1} \boldsymbol{\Phi}_{\Gamma_k}^{ST} \mathbf{G}_{\Gamma_k}^Q + \boldsymbol{\Phi}_{\Gamma_k}^{ST} \mathbf{K}_k^{\delta_L} \sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right) \\ \boldsymbol{\kappa}_3^{syn} &= \sum_{k=1}^K \left(\mathbf{G}_{\Gamma_k}^Q \sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right. \\ &\quad \left. + \omega_c^2 \left[\sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right]^T \mathbf{M}_k^{\delta_L} \sum_{m=1}^K \boldsymbol{\Psi}_{k,m}^{S,Q} \mathbf{G}_{\Gamma_k}^Q \right. \\ &\quad \left. + \sum_{m=1}^K \mathbf{G}_{\Gamma_m}^{QT} [\boldsymbol{\Psi}^{S,QT} \mathbf{M} \boldsymbol{\Psi}^{S,Q}]_{m,k} \mathbf{G}_{\Gamma_k}^Q \right)\end{aligned}$$

Again, $\boldsymbol{\Psi}^{S,QT} \mathbf{M} \boldsymbol{\Psi}^{S,Q}$ terms are calculated using cyclic symmetry in the similar vein

$$\tilde{\boldsymbol{\Psi}}_\Gamma^{S,QT} \tilde{\mathbf{M}} \tilde{\boldsymbol{\Psi}}_\Gamma^{S,Q} = \tilde{\mathbf{B}} \text{diag} [\tilde{\boldsymbol{\Psi}}_{\Gamma,h}^{S,QT} \tilde{\mathbf{M}}_h \tilde{\boldsymbol{\Psi}}_{\Gamma,h}^{S,Q}] \quad (11)$$

$$(\boldsymbol{\Psi}^{S,QT} \mathbf{M} \boldsymbol{\Psi}^{S,Q})_{k,m} = (\mathcal{F}_k \otimes \mathbf{I}) (\tilde{\boldsymbol{\Psi}}_\Gamma^{S,QT} \tilde{\mathbf{M}} \tilde{\boldsymbol{\Psi}}_\Gamma^{S,Q}) (\mathcal{F}_m^T \otimes \mathbf{I}) \quad (12)$$

Small mistuning projection

Following calculation of basis vectors and contribution of large geometric mistuning to the reduced matrices, we can obtain the total reduced mass and stiffness matrices $\boldsymbol{\mu}^{syn}$ and

$\boldsymbol{\kappa}^{syn}$ by projecting small random blade-to-blade variations with CMM method [Lim et al., 2003]

$$\begin{aligned}\boldsymbol{\mu}^{syn} &= \boldsymbol{\mu}_L^{syn} + (\boldsymbol{\Phi}_\Gamma^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q)^T \mathbf{M}^{\delta_S} (\boldsymbol{\Phi}_\Gamma^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q) \\ &= \boldsymbol{\mu}_L^{syn} + \sum_{n=1}^N \mathbf{q}_n^{MT} \mathbf{U}_n^{MT} \mathbf{M}_n^{\delta_S} \mathbf{U}_n^M \mathbf{q}_n^M \\ \boldsymbol{\kappa}^{syn} &= \boldsymbol{\kappa}_L^{syn} + (\boldsymbol{\Phi}_\Gamma^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q)^T \mathbf{K}^{\delta_S} (\boldsymbol{\Phi}_\Gamma^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q) \\ &= \boldsymbol{\kappa}_L^{syn} + \sum_{n=1}^N \mathbf{q}_n^{KT} \mathbf{U}_n^{KT} \mathbf{K}_n^{\delta_S} \mathbf{U}_n^K \mathbf{q}_n^K\end{aligned}\quad (13)$$

where the blade portion of compensated normal modes is expressed as

$$(\boldsymbol{\Phi}_\Gamma^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_\Gamma^Q) = \tilde{\mathbf{B}} \text{diag} [\mathbf{U}_n] \mathbf{q} \quad (14)$$

Note that matrices \mathbf{U}_n and modal participation factors \mathbf{q}_n , which represent the motion of n -th blade in the coordinates of basis vectors and described by cantilevered-blade normal modes together with boundary modes, are calculated for both mass and stiffness mistuning

$$\mathbf{U}_n^M = \begin{bmatrix} \boldsymbol{\Phi}_n^B & \boldsymbol{\Psi}_n^{B,M} \\ \mathbf{0} & \mathbf{I} \end{bmatrix} = \begin{bmatrix} \boldsymbol{\Phi}_n^B - (\mathbf{M}_{n,\Gamma_{ii}} + \mathbf{M}_{n_{ii}}^{\delta_L})^{-1} (\mathbf{M}_{n,\Gamma_{ib}} + \mathbf{M}_{n_{ib}}^{\delta_L}) & \\ \mathbf{0} & \mathbf{I} \end{bmatrix} \quad (15)$$

$$\mathbf{U}_n^K = \begin{bmatrix} \boldsymbol{\Phi}_n^B & \boldsymbol{\Psi}_n^{B,K} \\ \mathbf{0} & \mathbf{I} \end{bmatrix} = \begin{bmatrix} \boldsymbol{\Phi}_n^B - (\mathbf{K}_{n,\Gamma_{ii}} + \mathbf{K}_{n_{ii}}^{\delta_L})^{-1} (\mathbf{K}_{n,\Gamma_{ib}} + \mathbf{K}_{n_{ib}}^{\delta_L}) & \\ \mathbf{0} & \mathbf{I} \end{bmatrix} \quad (16)$$

where the subscripts i and b indicate that each n -th blade component is partitioned into internal and boundary DOFs.

$$\mathbf{q}_n^M = \begin{bmatrix} \mathbf{q}_{\Phi,n}^M \\ \mathbf{q}_{\Psi,n}^M \end{bmatrix} = \begin{bmatrix} \boldsymbol{\Phi}_n^{BT} (\mathbf{M}_{n,\Gamma} + \mathbf{M}_{n_{ii}}^{\delta_L}) (\boldsymbol{\Phi}_{n,\Gamma}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_{n,\Gamma}^Q) \\ (\boldsymbol{\Phi}_{n,\Gamma}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_{n,\Gamma}^Q)_b \end{bmatrix} \quad (17)$$

$$\mathbf{q}_n^K = \begin{bmatrix} \mathbf{q}_{\Phi,n}^K \\ \mathbf{q}_{\Psi,n}^K \end{bmatrix} = \begin{bmatrix} \boldsymbol{\Lambda}_n^{B-1} \boldsymbol{\Phi}_n^{BT} (\mathbf{K}_{n,\Gamma} + \mathbf{K}_{n_{ii}}^{\delta_L}) (\boldsymbol{\Phi}_{n,\Gamma}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_{n,\Gamma}^Q) \\ (\boldsymbol{\Phi}_{n,\Gamma}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_{n,\Gamma}^Q)_b \end{bmatrix} \quad (18)$$

subscript b denotes blade boundary DOF partition of compensated modes $(\boldsymbol{\Phi}_{n,\Gamma}^S - \boldsymbol{\Psi}^{S,Q} \mathbf{G}_{n,\Gamma}^Q)$. Unlike CMM based on tuned system nominal modes, the cantilevered-blade normal and boundary modes of each blade with rogue geometry must also be precalculated in addition to those of a tuned blade.

SMC Convergence

The CMM concept extends readily to the combined large plus small mistuning case; however, it is important to accurately approximate the eigenspace of the system undergoing large mistuning before any small mistuning projection. Therefore, it is useful to revisit the classical MAM formulation given in [Akgun, 1993]. If the mid-frequency truncation scheme with eigenvalue shifting is applied, the FRF can be expressed as

$$\mathbf{x}(\omega) \simeq \left(\sum_{j=1}^J \left((\omega^2 - \omega_c^2) (\mathbf{K} - \omega_c^2 \mathbf{M})^{-1} \mathbf{M} \right)^{j-1} (\mathbf{K} - \omega_c^2 \mathbf{M})^{-1} \mathbf{f}(\omega) \right. \\ \left. + \sum_{i=N_{low}}^{N_{high}} \left(\frac{\omega^2 - \omega_c^2}{\lambda_i - \omega_c^2} \right)^J \frac{\Phi_i^T \mathbf{f}(\omega)}{\lambda_i - \omega^2} \Phi_i \right) \quad (19)$$

where J is a positive integer denoting the level of acceleration, N_{low} and N_{high} are indexes of the first and last kept modes, and the frequency shift is usually taken as $\omega_c^2 = (\lambda_{N_{low}} - \lambda_{N_{high}})/2$, must satisfy $\omega_c^2 \neq \lambda_i$ and, in our case, $\omega_c^2 \neq \omega$. Clearly, the FRF $\mathbf{x}(\omega)$ obtained from (19) reach the exact values when the number of modes included increases. Although the accuracy of the FRF may also be improved by increasing the level of acceleration J , it seems impractical computationally. Since we do not know how many modes are enough to compute the FRFs to the prescribed accuracy, the iterative scheme becomes necessary. Also, a proper selection of ω_c^2 closer to some harder to approximate λ_i could help to accelerate the convergence to that eigenpair.

NUMERICAL STUDIES

The theory presented in the last section was implemented in FORTRAN code using high performance Intel MKL numerical library. The program incorporates mistuned system modes calculation and forced response analysis. In this section the performance of the algorithm will be illustrated using an industrial bladed disk model. First the natural frequencies and mode shapes of geometrically mistuned disc are analyzed. Second a realistic small mistuning is added on top of geometrical mistuning and forced response is calculated. Both mistuned modes and dynamic response are also computed with a full finite element model using the commercially available ANSYS FE code, which serves as a benchmark to assess the accuracy of ROM solution.

Large Mistuning

Consider the finite element model of an industrial 29 blade rotor depicted in Fig. 1. The model is build with standard linear eight node brick elements totaling 126846 DOF. In order to provide a simple test case the bladed disk was mistuned by significantly changing geometry of three arbitrarily selected blades with the assumption that mass density and Young's modulus are unchanged. We construct ROM

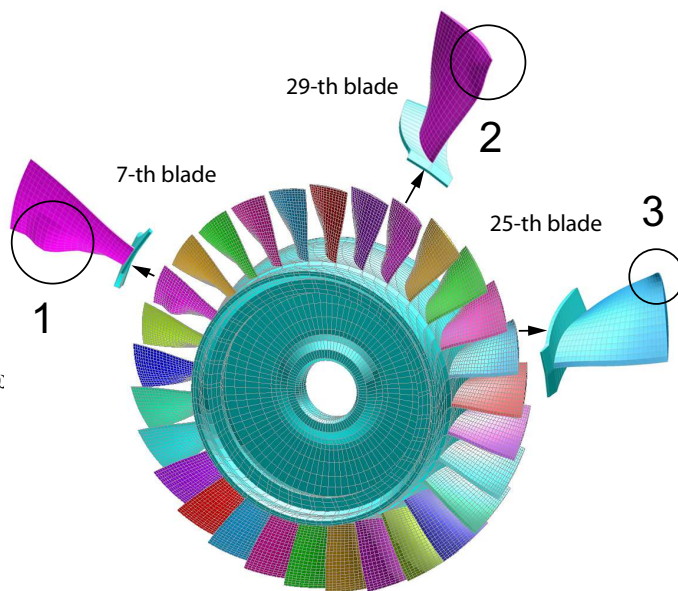


Figure 1. Finite element model of mistuned bladed disk.

for two frequency ranges 33 – 36 KHz and 14.5 – 16.5 KHz. The first band corresponds to $3T$ torsion and $2S$ stripe mode families, and the second to $2T/2F$ mixed torsion - flexural bending family of modes, which is shown in Fig. 2. The frequency bands of interest are marked by two horizontal lines.

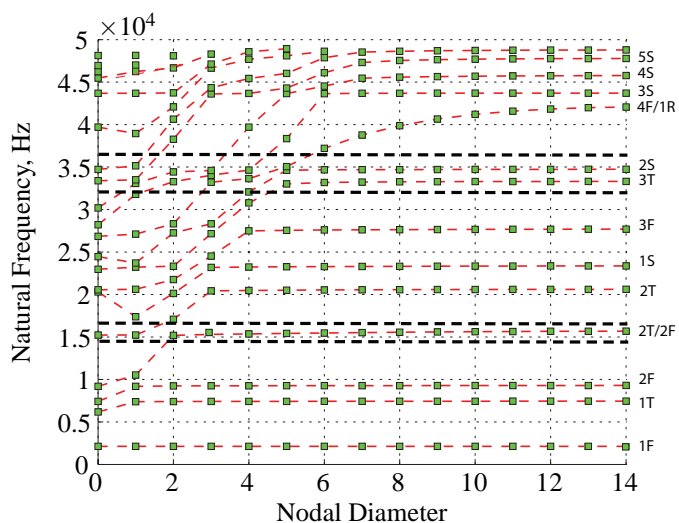


Figure 2. Natural frequencies versus nodal diameters for the tuned FEM

The selection of $2S$ and $3T$ families is a good candidate for a benchmark test because the amount of interaction between them would typically require to include both in ROM. $2T/2F$ mode family is relatively isolated, therefore it would provide an alternative test case for ROM.

To build the reduced order model, the following finite

element models are required:

- A single sector 4372 DOF model of the tuned system on which we run ANSYS modal analysis in cyclic symmetry and obtain system normal modes with harmonic blocks of mass and stiffness matrices in real-cyclic coordinates.
- Tuned blade 2496 DOF model on which we run ANSYS modal analysis to obtain the cantilevered-blade modes (used later when small mistuning is projected) and blade alone mass and stiffness matrices.
- For each blade with geometrical mistuning we run ANSYS modal analysis on 2496 DOF model to obtain the cantilevered-blade modes (used later to project small mistuning) and mistuned blade mass and stiffness matrices.

Fig. 3 and Fig. 4 depict the mistuned frequencies errors and MAC ratio between mistuned mode shapes predicted by ROM and ANSYS for the 33-36 KHz frequency band calculated with $\omega_c = 32400$ KHz. 71 natural frequencies that exist in that region are chosen for comparison with full FEM results. Note that even for the smallest model of 71 DOF

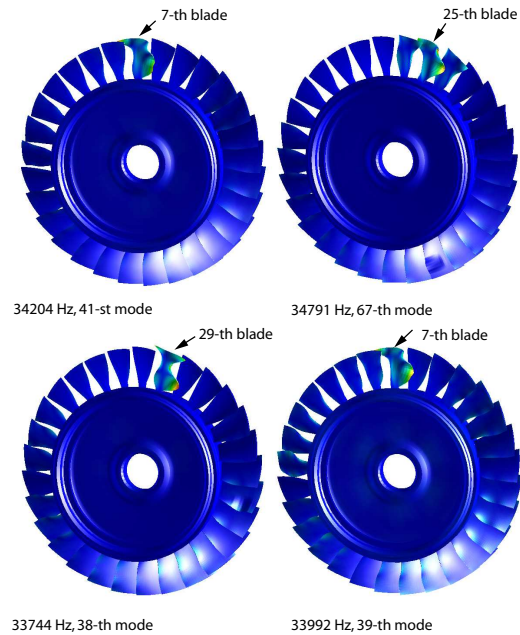


Figure 5. Highly localized modes.

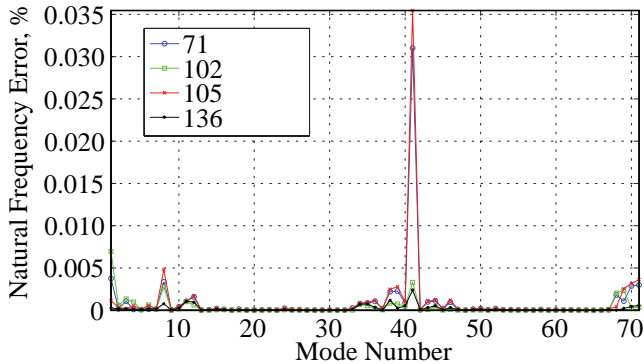


Figure 3. Mistuned natural frequencies errors for 33 – 36 KHz region

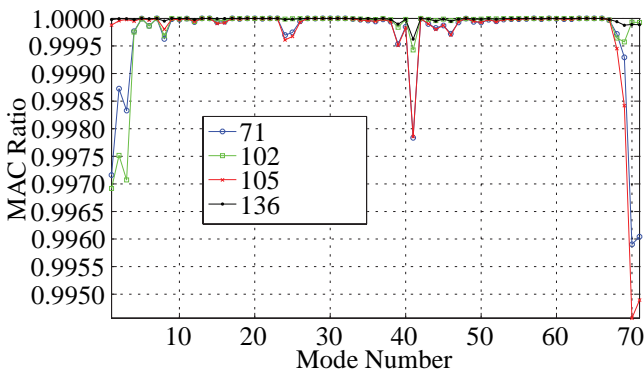


Figure 4. Comparison of MAC ratio between FEM and ROM mistuned mode shapes for 33 – 36 KHz region

the natural frequency error for all approximated modes is below 0.04% and MAC value above 0.995, which demonstrates that 71 DOF ROM is sufficiently accurate to capture the dynamics of mistuned system in the range 33 – 36 KHz. The models of greater size are shown to demonstrate the convergence towards FEM results. Here the modal basis was gradually augmented by including $3F$ and $4F/1R$ families of modes, which resulted in ROM of the size up to 136 DOF. Problematic for the proper development of an iterative algorithm is the fact that in some cases the asymmetric inclusion of mode families might potentially increase the error of approximation for the modes closer to the opposite side the studied frequency band, which is the case for the models of 102 and 105 DOF.

It should also be pointed out that the modes with higher error of approximation and slower convergence rate are those strongly localized to a single blade. Some of them that appear in 33 – 36 KHz frequency range are depicted in Fig. 5. In particular note 41-st mode located in the middle of the frequency range and localized to 7-th blade. The errors at the end of the analyzed frequency range are typical for any modal projection approach with modal truncation [Yang and Griffin, 2001] because the modes in the middle are more accurately represented than those at the end.

Fig. 6 and Fig. 7 depict the natural frequency errors and MAC ratio in the 14.5-16.5 KHz frequency band calculated with $\omega_c = 15390$ KHz. 29 modes are chosen for comparison with full FEM results. The error in the frequencies for the smallest model is less than 0.01 percent and the worst MAC ratio is 0.9995. Again the maximum error was observed for the 29-th mode, which is a highly localized to 7-th blade mode. Clearly, the error has been significantly reduced by

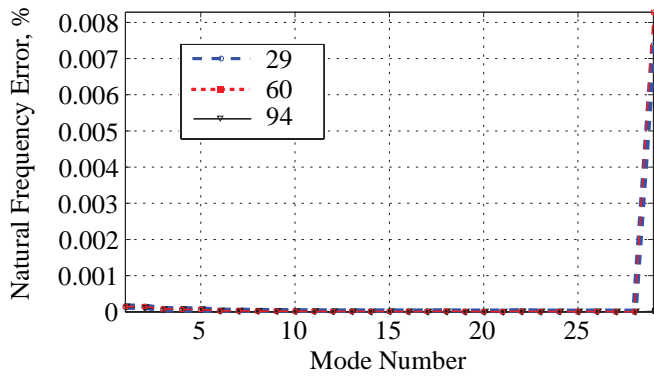


Figure 6. Mistuned natural frequencies errors for 14.5 – 16.4 KHz region

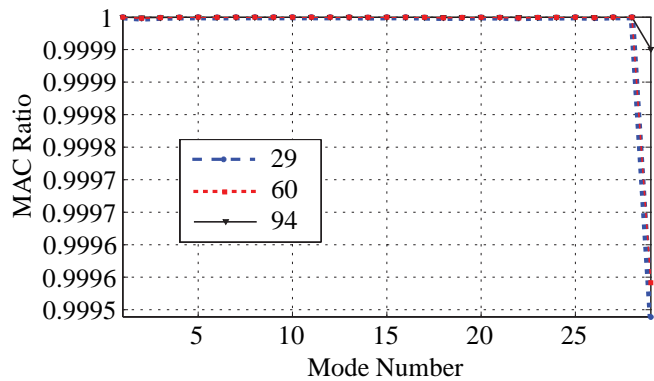


Figure 7. Comparison of MAC ratio between FEM and ROM mistuned mode shapes for 14.5 – 16.5 KHz region

increasing the number of modes included in the basis. The important point to note is that SMC better approximates the lower frequency modes and those of belonging to an isolated family, which results in very compact models of the number of blades order.

Combined Large and Small Mistuning

In this section SMC and CMM techniques are used to compute the dynamic response of the geometrically mistuned bladed disk system, Fig. 1, with the nominal Young's modulus of the n -the blade mistuned as

$$E_n = E_0(1 + \delta_n^e) \quad (20)$$

where E_0 is the nominal Young's modulus and δ_n^e is a non-dimensional mistuning value. The specific pattern used in this case is shown in Tab. 1. Although the proportional blade to blade stiffness variation introduced by changing Young's modulus of the blades is a very rough way to model random small mistuning phenomena, it has been chosen for simplicity to validate the accuracy of the developed method. In this study the structural damping coefficient is set to 0.006, the modal participation factors are calculated using

Table 1. Eigenvalue mistuning pattern

Blade	δ_n^e	Blade	δ_n^e
1	0.05704	16	0.04934
2	0.01207	17	0.04479
3	0.04670	18	0.03030
4	-0.01502	19	0.00242
5	0.05969	20	0.01734
6	-0.03324	21	0.02919
7	-0.00078	22	-0.00328
8	-0.01688	23	0.00086
9	0.00242	24	-0.03654
10	-0.02747	26	-0.03631
11	-0.03631	26	-0.01665
12	-0.03570	27	0.00783
13	-0.03631	28	-0.01169
14	-0.03631	29	-0.01332
15	0.00242		

30 cantilevered-blade modes, with only a third of them dominant. Despite the large changes in geometry the aerodynamic mistuning is neglected; the forcing function applied corresponds to the 3 and 5 nodal diameter excitation for 71 DOF ROM in the 33 – 36 KHz region, and 2 and 3 engine order excitation for 29 DOF ROM in the 14.6 – 16.6 KHz region. An arbitrary load is applied at a tip of each blade. Figures 8 and 9 show the Euclidean displacement norms

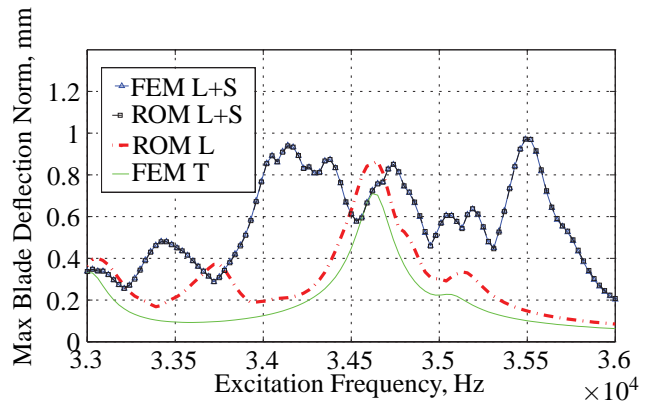


Figure 8. Forced response in 33 – 36 KHz region due to engine order 5 excitation

for maximum responding blade versus frequency of excitation for 33 – 36 KHz region EO 5 and EO 3 excitation respectively. The results for 14.5 – 16.5 KHz region under EO 3 and EO 2 excitation are depicted in Fig. 10 and Fig. 11. ANSYS results are marked as FEM L+S and those of re-

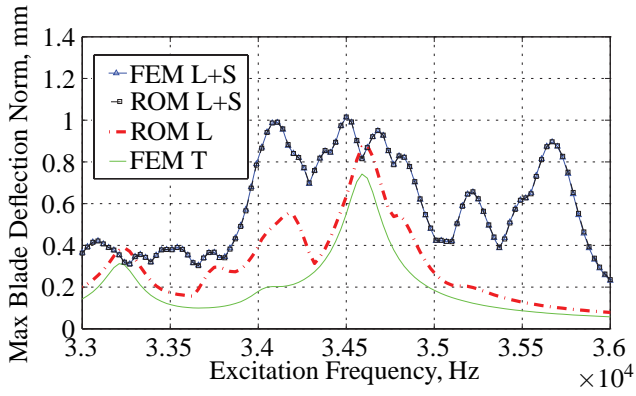


Figure 9. Forced response in 33 – 36 KHz region due to engine order 3 excitation

duced model as ROM L+S. The responses of the tuned system denoted as FEM T and geometrically mistuned system denoted as ROM L are presented for comparison. Observe

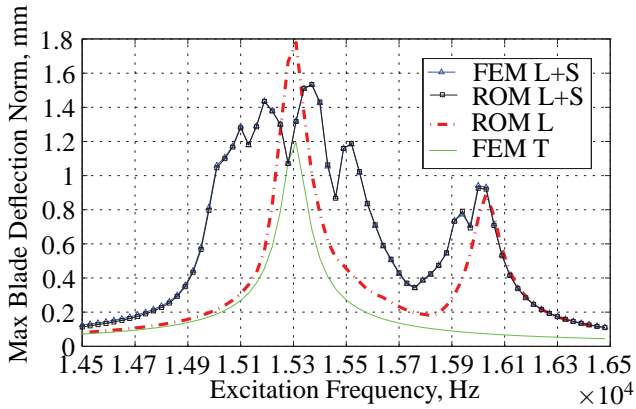


Figure 10. Forced response in 14.5 – 16.5 KHz region due to engine order 3 excitation

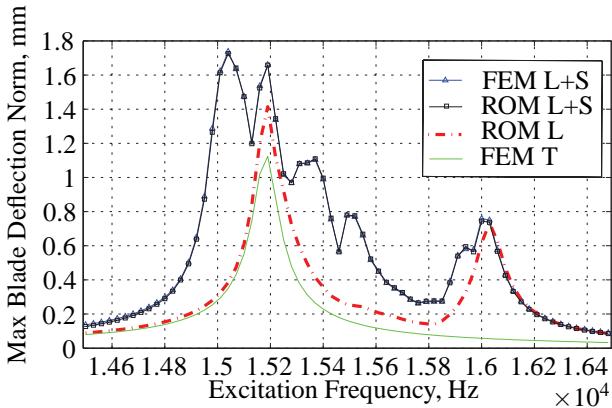


Figure 11. Forced response in 14.5 – 16.5 KHz region due to engine order 2 excitation

that the ROM in both frequency bands provide an accurate prediction of the mistuned systems forced response to engine order excitation in the tuned system veering regions as compared to the results computed by ANSYS. Even at the natural frequencies of slower to converge highly localized modes 71 and 29 DOF ROM are able to match FEM maximum forced response.

DISCUSSION

In developing ROM there is an initial computational overhead of calculating tuned system quasi-static modes $\Psi^{S,Q}$ that are used as input to SMC reduction algorithm. Since they need to be calculated only once for the selected frequency band and mistuned DOF distribution, the real computational advantage in using SMC occurs when simulating a large number of geometrically mistuned bladed disks either to analyze physical blade damage phenomena or for design optimization purposes. Thus we must solve $N_{harmonics}$ times a symmetric sparse system of the order of $2N_{DOF}$ of a basic sector. Additionally there is a small cost associated with calculating $\Psi^{S,Q^T} \mathbf{M} \Psi^{S,Q}$ matrix product for each harmonic. For the example presented here with 4372 DOF per sector model the time of calculation on a 2.66 GHz XEON Quad-Core with 4 GB of RAM is of the order of 8 minutes, however this time is dominated by disk I/O operations and is amenable to further optimization. The derivation of compensated basis vectors incurs the solution of a fully populated system in equation (7) of the order of total number of geometrically mistuned DOF. If a large number of DOF need to be mistuned, then the SMC reduction approach will be prohibitively expensive. The projection of large mistuning is carried out in an efficient, potentially parallelizable way, because the perturbation matrices corresponding to the geometric mistuning of blades have block-diagonal structure. The total time of calculation of 71 geometrically mistuned system mode shapes for the test case model with 3 mistuned blades, 594 mistuned DOF per blade, is 45 seconds. The same task with FEM requires ~ 5 minutes of ANSYS analysis running on the same platform, which demonstrates the potential of a substantial analysis cost reduction.

Depending on the applied geometric mistuning and the frequency band of interest we can obtain a very compact ROM of the order of number of blades. If system dynamics cannot be accurately captured by a reasonable number of modes while the smallest possible model is desired, it might be more efficient to first solve the reduced system to find mode shapes of a geometrically mistuned system and then use the approximated modes in CMM projection.

Throughout the algorithm, there are several decisions to be made. First, the centering frequency ω_c must be carefully selected not to cross the tuned system and potential geometrically mistuned system natural frequencies. Second, the number of cantilevered-blade normal modes to accurately represent small mistuning have to be determined given the

tuned and mistuned blades geometry. Finally, we have to know how many compensated normal modes should be retained in the reduced order model based on the expected error of approximation. Work is currently underway to develop an inexpensive iterative approach to better select the projection basis as well as to assess the perturbation bounds.

CONCLUSION

Motivated by the practical needs of the design community to assess the effect of random mistuning on geometrically mistuned bladed disks a new reduced order modeling method has been presented. By the use of two previously reported techniques, SMC and CMM, compact ROM were efficiently calculated that accurately capture the mistuned system dynamics with a very low number of states over a broad range of excitation frequencies and different geometry mistuning patterns. It was demonstrated with numerical examples that this method can accurately predict mistuned frequencies, mistuned mode shapes and forced response of a bladed disk with non symmetric geometric mistuning and simultaneous small perturbations in both mass and stiffness matrices of blades. The required amount of computation is significantly lower than that of the projection method using nominal geometrically mistuned system modes calculated by FEM analysis. In this light the developed numerical tool is very promising for future investigations of interaction of large deterministic and small random mistuning, as well as intentional mistuning.

REFERENCES

- [Akgun, 1993] Akgun, M. (1993). A new family of mode-superposition methods for response calculations. *Journal of Sound and Vibration*.
- [Bladh et al., 2001a] Bladh, R., Castanier, M., and Pierre, C. (2001a). Component-mode-based reduced order modeling techniques for mistuned bladed disks - part 1: Theoretical models. *ASME Journal of Engineering for Gas Turbines and Power*.
- [Bladh et al., 2001b] Bladh, R., Castanier, M., and Pierre, C. (2001b). Component-mode-based reduced order modeling techniques for mistuned bladed disks - part 2: Application. *ASME Journal of Engineering for Gas Turbines and Power*.
- [Bladh et al., 2002] Bladh, R., Castanier, M., Pierre, C., and Kruse, M. (2002). Dynamic response predictions for a mistuned industrial turbomachinery rotor using reduced order modeling. *ASME Journal of Engineering for Gas Turbines and Power*.
- [Castanier et al., 1997] Castanier, M., Ottarsson, G., and Pierre, C. (1997). A reduced-order modeling technique for mistuned bladed disks. *Journal of Vibration and Acoustics*.
- [Lim et al., 2003] Lim, S., Bladh, R., Castanier, M., and Pierre, C. (2003). A compact, generalized component mode mistuning representation for modeling bladed disk vibration. In *Proceedings of the 45-th AIAA/ASME/ASCE/AHS/ASC Structures, Structural Dynamics and Material Conference, paper 2003-1545*, Palm Springs, California, USA.
- [Lim et al., 2004] Lim, S., Castanier, M., and Pierre, C. (2004). Vibration modeling of bladed disks subject to geometric mistuning and design changes. In *Proceedings of the 45-th AIAA/ASME/ASCE/AHS/ASC Structures, Structural Dynamics and Material Conference, paper 2004-1686*, Palm Springs, California, USA.
- [Petrov et al., 2002] Petrov, E., Sanliturk, K., and Ewins, D. (2002). A new method for dynamic analysis of mistuned bladed disks based on the exact relationship between tuned and mistuned systems. *ASME Journal of Engineering for Gas Turbines and Power*.
- [Sinha, 2007] Sinha, A. (2007). Reduced-order model of a bladed rotor with geometric mistuning.
- [Stewart and Sun, 1990] Stewart, G. and Sun, J. (1990). *Matrix Perturbation Theory*. Academic Press, San Diego CA.
- [Yang and Griffin, 2001] Yang, M. and Griffin, J. (2001). A reduced order model of mistuning using a subset of nominal system modes. *ASME Journal of Engineering for Gas Turbines and Power*.