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► **To cite this version:**

Christine Chevallereau, Eric Westervelt, Jessy Grizzle. Asymptotically Stable Running for a Five-Link, Four-Actuator, Planar Bipedal Robot. *The International Journal of Robotics Research*, 2005, 24 (6), pp.431-464. hal-00794892

HAL Id: hal-00794892

<https://hal.science/hal-00794892v1>

Submitted on 26 Feb 2013

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Asymptotically Stable Running for a Five-Link, Four-Actuator, Planar Bipedal Robot

C. Chevallereau, E.R. Westervelt, and J.W. Grizzle

Abstract—Provably asymptotically stable running gaits are developed for the five-link, four-actuator bipedal robot, RABBIT. A controller is designed so that the Poincaré return map associated with periodic running gaits can be computed on the basis of a model with impulse-effects that, previously, had been used only for the design of walking gaits. This feedback design leads to the notion of a hybrid zero dynamics for running, which in turn allows the existence and stability of running gaits to be determined on the basis of a scalar map. The main results are illustrated via simulations performed on models with known parameters and on models with parameter uncertainty and structural changes. Animations of the resulting running motions are available on the web.

I. INTRODUCTION

THIS paper addresses the design and analysis of asymptotically stable running gaits for RABBIT, a five-link, four-actuator, planar bipedal robot [1]. Running is defined as *forward motion* with alternation of *single support* (one leg on the ground) and *flight* (no contact with the ground); without forward motion, the robot would be jumping in place, or “jogging” in place. RABBIT was specifically designed to advance the fundamental understanding of *controlled* legged locomotion. In particular, it was designed without feet, and hence a widely-used stability heuristic, called the zero moment point (ZMP) criterion, is not applicable. The robot was designed to be able to walk with an average forward speed of at least 5 km/h and to run at more than 12 km/h.

In a series of papers, the authors and colleagues have developed new feedback control strategies [2]–[7] that achieve provably asymptotically-stable walking gaits in underactuated bipeds, such as RABBIT, and demonstrated many of them experimentally [8], [9]. In regards to running, open-loop trajectories have been studied in [10], [11]. An objective of this paper is to develop a *time-invariant feedback controller* that realizes these open-loop running trajectories as *provably asymptotically-stable orbits*. A key step is to design the feedback controller in such a way that the Poincaré map associated with a running gait can be computed on the basis of the models with impulse effects studied in [3] for the design of walking gaits. This leads to the notion of a hybrid zero dynamics (HZD) for running, and to the closed-form computation of the restricted Poincaré return map [12].

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Section III presents the models of the stance, flight, and impact phases of the studied robot. These are assembled into an overall hybrid model of running in Section IV. A qualitative discussion of the proposed control law for running is presented in Section V. The details of the control law are developed in Section VI. Like the model, the controller is hybrid, with both continuous-time and event-based actions taking place. In the stance phase, the control law is identical to the development in [3]. The control law in the flight phase consists of two parts: a continuous action similar to [3] plus a discrete or event-to-event action that adjusts the coefficients in the continuous portion of the controller in order to achieve a landing objective that assures the existence of a hybrid zero dynamics. The existence and stability properties of periodic orbits of the closed-loop hybrid model are studied in Section VII. The main contribution is the closed-form computation of the restricted Poincaré return map of the hybrid zero dynamics, because analysis of orbits in the closed-loop, multi-phase, hybrid model can then be performed on the basis of an easily computable scalar map. Section VIII begins an extensive simulation study of the control laws proposed in the paper as applied to the underactuated, planar bipedal robot, RABBIT [1]. The restricted Poincaré return map is computed for ten different running trajectories, varying from 0.5 m/s to 2.75 m/s, showing the ability to achieve stable running motions over a wide range of speeds. The trajectories for 1.5 m/s and 2.5 m/s are illustrated in detail under the assumption of a perfect model. The robustness to model imperfection is studied via simulation in Section IX. The systematic design of auxiliary event-based control action [4], [13] to accelerate the rate of convergence to a periodic orbit is illustrated in Section X. The paper is concluded in Section XI. Animations and other supporting material are available on the web [14].

II. RELATED WORK

In the early 1980’s, Raibert proposed an elegant conceptualization of running in terms of a one-legged, prismatic-kneed hopper [15], [16]. He decomposed his control actions into three parts—hopping height, foot touchdown angle, and body posture—and emphasized the role of symmetry in designing stable running motions. The remarkable success of Raibert’s control law motivated others to analytically characterize its stability [17], [18], and to further investigate the role of passive elements in achieving efficient running with a hopper [19]. Raibert’s control scheme has been augmented with leg-coordination logic to achieve running in prismatic-kneed bipeds and quadrupeds [15], [20].

Various attempts have been made to extend a Raibert-style controller to a robot with revolute joints. Building on the

ideas of Koditschek et al. [21]–[24], a spring-loaded inverted-pendulum model—essentially a prismatic-kneed hopper with a radially symmetric body and a massless leg—was approximately embedded into a one-legged, revolute-jointed robot with an ankle, knee, and hip [25], [26]. With this approach, the center of mass of the robot could be nicely controlled. However, there was no natural “posture principle” to specify the evolution of the remaining degrees of freedom of the robot; also, the ballistic phase was difficult to address, due to underactuation and the non-holonomic constraint arising from conservation of angular momentum. Various methods for controlling the attitude of bodies undergoing ballistic motion have been studied; see [27]–[31] and references therein.

In late 2003, both Iguana Robotics and Sony announced (separate) experimental demonstrations of running for bipedal robots with revolute knees. In early 2004, running was announced for another humanoid robot, HRP-2LR [32], and in December 2004, Honda’s robot, ASIMO, achieved running. The readers are invited to seek videos of these robots on the web. The controllers of the Sony and Honda robots are based on the ZMP, that of Iguana Robotics is based on central pattern generators (CPGs), and HRP-2LR uses “resolved momentum”. To our best knowledge, only two other bipeds with revolute knees have been designed to perform running—Johnnie in Munich, Germany [33], [34] and RABBIT in Grenoble, France [1], [35].

The computation of optimal running trajectories has been studied in [36]. Trajectory tracking for running was investigated on a simulation model of the Honda biped, ASIMO, [37]. A nominal trajectory was computed off-line, and then during the ballistic phase, an on-line trajectory modification was made to allow Raibert-like control of foot placement. A control strategy for running that does not rely on trajectory tracking was studied in [38], where a Raibert-like controller was used on a planar robot with a torso and two prismatic-kneed legs. The control action sought to excite natural passive solutions of the dynamics by restoring energy lost at touchdown.

III. MECHANICAL MODEL OF A BIPED RUNNER

A. The biped

The studied bipedal robot evolves with respect to a flat surface in the sagittal plane; see Fig. 1. The flat surface will be referred to as the ground. The robot is composed of five rigid links with mass, connected through ideal (i.e., rigid and frictionless) revolute joints to form a torso and two identical legs, with each leg articulated by a knee. Each leg end is terminated in a point so that, in particular, the robot does not have feet. An actuator is provided at the each knee and two actuators are provided at the hips, acting between the torso and the legs. There is no actuation at the leg ends. Hence, in single support, the robot is underactuated.

The robot is said to be in *flight phase* when there is no contact with the ground, and in *stance phase* when one leg end is in stationary contact with the ground (that is, the leg end is acting as an ideal pivot) and the other leg is free. For the stance phase, the leg in contact with the ground is called the *stance leg* and the other leg is the *swing leg*.

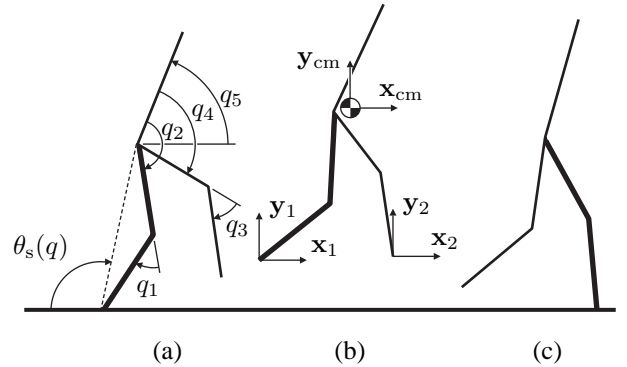


Fig. 1. Different phases of running with coordinate conventions labeled. The robot is shown (a) at the end of the stance phase; (b) during flight; and (c) at the beginning of the stance phase just after impact. To avoid clutter, the coordinate conventions have been spread out over the single support and flight phases even though they apply to all three phases. leg-1 is presented in bold. Angles are positive in the clockwise direction.

In the flight phase, the robot has seven degrees of freedom (DOF): a degree of freedom associated with the orientation of each link, plus two DOF associated with the horizontal and vertical displacement of the center of mass within the sagittal plane. The state vector of the dynamical model is thus 14-dimensional: there are seven configuration variables required to describe the position of the robot, plus the associated velocities. In the stance phase, the robot has only five DOF because the position of the center of mass is determined by the orientation of the five links (plus a horizontal, constant off-set of the stance leg end with respect to a world frame). The state vector of the dynamical model is thus 10-dimensional.

B. Lagrangian model for flight

A convenient choice of configuration variables is depicted in Fig. 1. The vector of body coordinates q_b consisting of the relative angles $(q_1, q_2, q_3, q_4)'$ describes the shape of the biped. The biped’s absolute orientation with respect to the world frame is given by q_5 . The biped’s absolute position is specified by the Cartesian coordinates of the center of mass, (x_{cm}, y_{cm}) . The vector of generalized coordinates is denoted as $q_f := (q_b', q_5, x_{cm}, y_{cm})'$.

The dynamic model is easily obtained with the method of Lagrange, which consists of first computing the kinetic energy and potential energy of each link, and then summing terms to compute the total kinetic energy, K_f , and the total potential energy, V_f [39]–[41]. The Lagrangian is defined as $\mathcal{L}_f = K_f - V_f$, and the dynamical model is determined from Lagrange’s equation

$$\frac{d}{dt} \frac{\partial \mathcal{L}_f}{\partial \dot{q}_f} - \frac{\partial \mathcal{L}_f}{\partial q_f} = \Gamma_f, \quad (1)$$

where Γ_f is the vector of generalized forces and torques applied to the robot.

In terms of the generalized coordinates of the robot, q_f , the total kinetic energy becomes

$$K_f = \frac{1}{2} \dot{q}_f' D_f(q_b) \dot{q}_f, \quad (2)$$

where

$$D_f = \begin{bmatrix} A(q_b) & 0_{5 \times 2} \\ 0_{2 \times 5} & mI_{2 \times 2} \end{bmatrix}, \quad (3)$$

m is the total mass of the robot, and A depends only on q_b because the total kinetic energy is invariant under rotations and translations of the body. The potential energy is

$$V_f = mg \mathbf{y}_{\text{cm}}. \quad (4)$$

The principle of virtual work yields that the external torques are

$$\Gamma_f = B_f u = \begin{bmatrix} I_{4 \times 4} \\ 0_{3 \times 4} \end{bmatrix} u, \quad (5)$$

where u is the vector of actuator torques applied at the four joints of the robot. Applying Lagrange's equation leads to a model of the form

$$D_f(q_b)\ddot{q}_f + C_f(q_b, \dot{q}_f)\dot{q}_f + G_f(q_f) = B_f u, \quad (6)$$

where D_f is the inertia matrix, the matrix C_f contains Coriolis and centrifugal terms, and G_f is the gravity vector. Introducing the state vector $x_f := (q_f', \dot{q}_f)'$, the Lagrangian model (6) is easily expressed as

$$\dot{x}_f = f_f(x_f) + g_f(x_f)u. \quad (7)$$

The state space is taken as $T\mathcal{Q}_f := \{x_f := (q_f', \dot{q}_f)' \mid q_f \in \mathcal{Q}_f, \dot{q}_f \in \mathbb{R}^7\}$, where, if \mathbb{S} denotes the circle, the configuration space \mathcal{Q}_f is a simply-connected, open subset of $\mathbb{S}^5 \times \mathbb{R}^2$ corresponding to physically reasonable configurations of the robot.

C. Lagrangian model for stance

For the stance phase, the generalized coordinates can be taken as $q := (q_b', q_5)' = (q_1, \dots, q_5)'$. Since the robot's legs are identical, in the stance phase, it will be assumed without loss of generality that leg-1 is in contact with the ground. Moreover, the Cartesian position of the stance leg end will be identified with the origin of the $(x-y)$ -axes of a world frame.

The position of the center of mass can be expressed in terms of q per

$$\begin{bmatrix} \mathbf{x}_{\text{cm}}(q) \\ \mathbf{y}_{\text{cm}}(q) \end{bmatrix} = \mathbf{f}_1(q), \quad (8)$$

where \mathbf{f}_1 is determined from the robot's geometric parameters (link lengths, masses, positions of the centers of mass). Hence

$$\dot{q}_f = \begin{bmatrix} I_{5 \times 5} \\ \frac{\partial \mathbf{f}_1}{\partial q} \end{bmatrix} \dot{q}. \quad (9)$$

Substituting (9) into (2) yields the kinetic energy of the stance phase,

$$K_s = \frac{1}{2} \dot{q}' D_s(q_b) \dot{q}, \quad (10)$$

with

$$D_s(q_b) = A(q_b) + m \frac{\partial \mathbf{f}_1(q)}{\partial q} \frac{\partial \mathbf{f}_1(q)}{\partial q}; \quad (11)$$

because the kinetic energy is invariant under rotations of the body, D_s depends only on q_b . The potential energy remains $V_s(q) = mg \mathbf{y}_{\text{cm}}(q)$. Lagrange's equation becomes

$$\frac{d}{dt} \frac{\partial \mathcal{L}_s}{\partial \dot{q}} - \frac{\partial \mathcal{L}_s}{\partial q} = \Gamma_s, \quad (12)$$

and the external torques are

$$\Gamma_s = B_s u = \begin{bmatrix} I_{4 \times 4} \\ 0_{1 \times 4} \end{bmatrix} u. \quad (13)$$

The dynamic model can therefore be written as

$$D_s(q_b)\ddot{q} + C_s(q_b, \dot{q})\dot{q} + G_s(q) = B_s u. \quad (14)$$

Introducing the state vector $x_s := (q', \dot{q})'$, the Lagrangian model (14) is easily expressed as

$$\dot{x}_s = f_s(x_s) + g_s(x_s)u. \quad (15)$$

The state space is taken as $T\mathcal{Q}_s := \{x_s := (q', \dot{q})' \mid q \in \mathcal{Q}_s, \dot{q} \in \mathbb{R}^5\}$, where the configuration space \mathcal{Q}_s is a simply-connected, open subset of \mathbb{S}^5 corresponding to physically reasonable configurations.

D. The impact model

The Cartesian position of the end of leg-2 can be expressed in terms of the Cartesian position of the center of mass and the robot's angular coordinates as

$$\begin{bmatrix} \mathbf{x}_2 \\ \mathbf{y}_2 \end{bmatrix} = \begin{bmatrix} \mathbf{x}_{\text{cm}} \\ \mathbf{y}_{\text{cm}} \end{bmatrix} - \mathbf{f}_2(q), \quad (16)$$

where \mathbf{f}_2 is determined from the robot's parameters (links lengths, masses, positions of the centers of mass); see (8). When leg-2 touches the ground at the end of a flight phase, an impact takes place. The impact model of [42], [43] is used, which represents the ground reaction forces at impact as impulses with intensity I_R . The impact is assumed inelastic, with the velocity of the contact leg end becoming zero instantaneously; furthermore, after impact, the contact leg end is assumed to act as an ideal pivot. This model yields that the robot's configuration q_f is unchanged during impact and there are instantaneous changes in the velocities.

The velocity vector *just before* impact is denoted by \dot{q}_f^- . After impact, with the assumption that the leg neither rebounds nor slides after impact, the robot is in stance phase. leg-2 acts as an ideal pivot and thus the linear velocity of the center of mass can be expressed in terms of the angular velocities *just after* impact, \dot{q}^+ , yielding

$$\begin{bmatrix} 0 \\ 0 \end{bmatrix} = \begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}^+ \\ \dot{\mathbf{y}}_{\text{cm}}^+ \end{bmatrix} - \frac{\partial \mathbf{f}_2(q)}{\partial q} \dot{q}^+. \quad (17)$$

The impact model of [42], [43] is expressed as

$$\begin{bmatrix} A(q_b) & 0_{5 \times 2} \\ 0_{2 \times 5} & mI_{2 \times 2} \end{bmatrix} \begin{pmatrix} \dot{q}^+ \\ \begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}^+ \\ \dot{\mathbf{y}}_{\text{cm}}^+ \end{bmatrix} - \dot{q}_f^- \end{pmatrix} = \begin{bmatrix} \frac{-\partial \mathbf{f}_2'(q)}{\partial q} \\ I_{2 \times 2} \end{bmatrix} I_R. \quad (18)$$

The vector I_R of the ground reaction impulses can be expressed using the last two lines of the matrix equation (18) in combination with (17):

$$I_R = m \left(\frac{\partial \mathbf{f}_2(q)}{\partial q} \dot{q}^+ - \begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}^- \\ \dot{\mathbf{y}}_{\text{cm}}^- \end{bmatrix} \right). \quad (19)$$

Substituting this into the first five lines of (18) and rearranging yields that the robot's angular velocity vector after impact is given by a linear expression with respect to the velocity before impact:

$$\dot{q}^+ = \left[A + m \frac{\partial \mathbf{f}_2'}{\partial q} \frac{\partial \mathbf{f}_2}{\partial q} \right]^{-1} \left[A \mid m \frac{\partial \mathbf{f}_2'}{\partial q} \right] \dot{q}_f^-, \quad (20)$$

which, for later use, is written as

$$\dot{q}^+ = \tilde{\Delta}(\dot{q}_f^-, \dot{q}_f^-). \quad (21)$$

Remark 1: In the case of running, since the robot has 7 DOF before impact and only 5 DOF after impact, for any post-impact velocity, there is a two-dimensional set of velocities in the flight phase that gets mapped onto that same vector. This is different from walking where, generically, the double support impact results in a one-to-one mapping between pre-impact and post-impact velocity vectors.

E. Some linear and angular momentum relationships

A few linear and angular momentum properties of the mechanical models are noted. Let σ_{cm} denote the angular momentum of the biped about its *center of mass*. In the flight phase, σ_{cm} can be computed by $\sigma_{\text{cm}} = \frac{\partial K_f}{\partial \dot{q}_5} = A_{55} \dot{q}$, where A_{55} is the fifth row of A . The fifth row of (1) yields conservation of σ_{cm} ,

$$\dot{\sigma}_{\text{cm}} = 0. \quad (22)$$

In addition, the last two rows of (1) correspond to Newton's second law in a central gravity field:

$$m \ddot{\mathbf{x}}_{\text{cm}} = 0 \quad \text{and} \quad m \ddot{\mathbf{y}}_{\text{cm}} = -mg. \quad (23)$$

Let σ_i denote the angular momentum of the biped about the *end of leg- i* , for $i = 1, 2$. The three angular momenta are related by

$$\sigma_i = \sigma_{\text{cm}} + m((\mathbf{y}_{\text{cm}} - \mathbf{y}_i) \dot{\mathbf{x}}_{\text{cm}} - (\mathbf{x}_{\text{cm}} - \mathbf{x}_i) \dot{\mathbf{y}}_{\text{cm}}). \quad (24)$$

This expression is valid in both the stance and flight phases. In the stance phase, σ_1 is determined by $\sigma_1 = \frac{\partial K_s}{\partial \dot{q}_5} = D_{s,5} \dot{q}$, where $D_{s,5}$ is the fifth row of D_s . The fifth row of (12) yields the angular momentum balance theorem:

$$\dot{\sigma}_1 = -\frac{\partial V_s}{\partial q_5} = mg \mathbf{x}_{\text{cm}}. \quad (25)$$

The impact model of [42], [43] yields conservation of angular momentum about the impact point. Indeed, because the only external impulsive force is applied at the impact point, the fifth row of (18) can be written as

$$\sigma_{\text{cm}}^+ - \sigma_{\text{cm}}^- = m [\mathbf{y}_{\text{cm}} - \mathbf{y}_2 \mid -\mathbf{x}_{\text{cm}} + \mathbf{x}_2] \begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}^+ - \dot{\mathbf{x}}_{\text{cm}}^- \\ \dot{\mathbf{y}}_{\text{cm}}^+ - \dot{\mathbf{y}}_{\text{cm}}^- \end{bmatrix}, \quad (26)$$

since $\frac{\partial \mathbf{f}_2}{\partial q_5} = [-\mathbf{x}_{\text{cm}}(q) \mid \mathbf{y}_{\text{cm}}(q)]'$. Using (24) results in

$$\sigma_2^+ = \sigma_2^-, \quad (27)$$

meaning the value of σ_2 is unchanged during the impact. Since the stance phase model assumes that the stance leg is leg-1, for later use, (27) is rewritten as

$$\sigma_1^{s+} = \sigma_2^{f-} \quad (28)$$

to reflect the swapping of the roles of the legs; see (30).

Remark 2: The notation s+ emphasizes that σ_1 is being evaluated at the *beginning of the stance phase* and the notation f- emphasizes that σ_2 is being evaluated at the *end of the flight phase*. If no confusion is possible, the notation + and - will be used. For example, the variable θ_s only makes sense in the stance phase, and hence θ_s^{s-} would be redundant. On the other hand, for a variable such as \mathbf{x}_{cm} , it is important to distinguish among $\mathbf{x}_{\text{cm}}^{s+}$, $\mathbf{x}_{\text{cm}}^{s-}$, $\mathbf{x}_{\text{cm}}^{f+}$, and $\mathbf{x}_{\text{cm}}^{f-}$.

Remark 3: The robot is assumed to advance from left to right that is, in the positive direction of the \mathbf{x} -coordinate. In this paper, angles are positive when measured in the clockwise direction so that the angular momenta about the stance leg end and the center of mass will be positive in the simulations. A more classical choice of measuring the angles in the trigonometric sense, that is, positive is counter clockwise, would lead to negative angular momenta for left-to-right movement of the robot. In this case, (24) would become $\sigma_i = \sigma_{\text{cm}} + m((\mathbf{x}_{\text{cm}} - \mathbf{x}_i) \dot{\mathbf{y}}_{\text{cm}} - (\mathbf{y}_{\text{cm}} - \mathbf{y}_i) \dot{\mathbf{x}}_{\text{cm}})$. In turn, certain equations derived from this one would have to be modified. For the convenience of the reader, the principal changes are noted in Appendix VI.

IV. HYBRID MODEL OF RUNNING

The overall biped robot model can be expressed as a nonlinear hybrid system containing two state manifolds (called "charts" in [44]):

$$\Sigma_f : \begin{cases} \mathcal{X}_f = TQ_f \\ \mathcal{F}_f : (\dot{x}_f) = f_f(x_f) + g_f(x_f)u \\ \mathcal{S}_f^s = \{x_f \in \mathcal{X}_f \mid H_f^s(x_f) = 0\} \\ \mathcal{T}_f^s : x_s^+ = \Delta_f^s(x_f^-) \end{cases} \quad (29)$$

$$\Sigma_s : \begin{cases} \mathcal{X}_s = TQ_s \\ \mathcal{F}_s : (\dot{x}_s) = f_s(x_s) + g_s(x_s)u \\ \mathcal{S}_s^f = \{x_s \in \mathcal{X}_s \mid H_s^f(x_s) = 0\} \\ \mathcal{T}_s^f : x_f^+ = \Delta_s^f(x_s^-) \end{cases}$$

where, for example, \mathcal{F}_f is the flow on state manifold \mathcal{X}_f , \mathcal{S}_f^s is the switching hyper-surface for transitions between \mathcal{X}_f and \mathcal{X}_s , and $\mathcal{T}_f^s : \mathcal{S}_f^s \rightarrow \mathcal{X}_s$ is the transition function applied when $x_f \in \mathcal{S}_f^s$.

The transition from flight phase to stance phase occurs when leg-2 impacts the ground. Hence, $H_f^s(x_f) = \mathbf{y}_2$; see Fig. 1. The ensuing initial value of the stance phase, x_s^+ , is determined from the impact model of Section III-D. A relabeling matrix R is applied to the angular coordinates to account for the impact

occurring on leg-2 while the stance model assumes leg-1 is in contact with the ground:

$$\Delta_f^s(x_f^-) = \begin{bmatrix} [R | 0_{5 \times 2}] q_f^- \\ R \tilde{\Delta}(x_f^-) \end{bmatrix}, \quad (30)$$

and

$$R = \begin{bmatrix} 0 & 0 & 1 & 0 & 0 \\ 0 & 0 & 0 & 1 & 0 \\ 1 & 0 & 0 & 0 & 0 \\ 0 & 1 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 & 1 \end{bmatrix}, \quad (31)$$

where (21) has been used.

The transition from stance phase to flight phase can be initiated by causing the acceleration of the stance leg end to become positive. If torque discontinuities¹ are allowed—as they are assumed to be in this paper—when to transition into the flight phase becomes a control decision. Here, in view of simplifying the analysis of periodic orbits in Section VII, the transition is assumed to occur at a pre-determined point in the stance phase. Hence, $H_s^f = \theta_s(q) - \theta_{s,0}^-$, where $\theta_s(q) := \frac{q_1}{2} + q_2 + q_5$ is the angle² of the hips with respect to end of the stance leg (see Fig. 1) and $\theta_{s,0}^-$ is a constant to be determined. The ensuing initial value of the flight phase, x_f^+ , is defined so as to achieve continuity in the position and velocity variables, using (8) and (9):

$$\Delta_s^f(x_s^-) = \begin{bmatrix} \begin{bmatrix} q^- \\ \mathbf{f}_1(q^-) \end{bmatrix} \\ \begin{bmatrix} q^- \\ \frac{\partial \mathbf{f}_1}{\partial q} \Big|_{q^-} \dot{q}^- \end{bmatrix} \end{bmatrix}. \quad (32)$$

Continuity of the torques is not imposed, and hence neither is continuity of the accelerations. It is assumed that the control law in the flight phase will be designed to achieve $\ddot{y}_1^{f+} > 0$; see [11].

The definition of a solution of the hybrid model is adopted from [44]. For the definitions of orbital stability in the sense of Lyapunov, attractivity, and orbital asymptotic stability in the sense of Lyapunov, see [2], [45], [46].

V. QUALITATIVE DISCUSSION OF CONTROL LAW DESIGN

This section describes in qualitative terms a control law design for planar bipedal running that is presented in detail in Section VI, analyzed in Section VII, and illustrated via simulations on RABBIT in Section VIII. A related discussion about control law design for walking is available in [1]. An important difference is that running has two continuous-time

¹This is a modeling decision. In practice, the torque is continuous due to actuator dynamics. It is assumed here that the actuator time constant is small enough that it need not be modeled. See the very end of Section IX-B for further discussion of this point.

²In RABBIT the femur and tibia are of equal length. In general, any function that is independent of the body coordinates and is monotonically increasing along the motion of the robot can be used. As long as the tibia and femur are approximately of the same length, $\theta_s(q) := \frac{q_1}{2} + q_2 + q_5$ is a reasonable choice.

phases, stance and flight, and discrete transitions between them. Moreover, the flight phase presents more complications for controller design than does the stance phase: it has two additional degrees of freedom and angular momentum about the center of mass is conserved.

A. Analytical tractability through invariance, attractivity, and configuration determinism at transitions

Above all, the control strategy is constructed to facilitate the stability analysis of the closed-loop hybrid system consisting of the robot, the running surface, and the feedback controller. If stability analysis can be rendered sufficiently simple, then it becomes possible to explore efficiently a large set of asymptotically-stable running gaits in order to find one that meets additional performance objectives, such as minimum energy consumption per distance traveled for a given average speed, or minimum peak-actuator power demand.

The controller will be hybrid, with continuous-time feedback signals applied in stance and flight phases, and discrete (or event-based) updates of controller parameters performed at transitions between phases. The design of the controller uses two principles that are ubiquitous in non-hybrid systems, namely invariance and attractivity, with the notion of invariance being extended to hybrid systems so as to include the discrete transitions as well as the continuous flow of the Lagrangian dynamics. Hybrid invariance will lead to the creation of a low-dimensional hybrid subsystem of the full-order closed-loop system. The low-order hybrid subsystem is called the Hybrid Zero Dynamics (HZD) of running. Attractivity will mean that trajectories of the full-order closed-loop system converge locally and sufficiently rapidly to those of the hybrid zero dynamics so that existence and stability of periodic running motions can be restricted to the study of the hybrid zero dynamics. The Poincaré return map for the hybrid zero dynamics will turn out to be one-dimensional. When transitions between phases in the hybrid zero dynamics occur at known configurations of the robot—this is called configuration determinism at transitions—it will turn out that the Poincaré return map can be computed in closed form, thereby rendering stability analysis of the closed-loop system tractable.

B. Desired geometry of the closed-loop system

The objective of the control law design is to achieve the internal structure of the closed-loop system that is depicted in Fig. 2. The vertical surfaces \mathcal{S}_s^f and \mathcal{S}_f^s represent the points in the state space where the transitions from stance to flight and from flight to stance occur, respectively; see (29). The horizontal surface on the left, \mathcal{Z}_s , is created by the stance-phase controller; it is designed to be invariant in the usual sense that if the system is initialized in \mathcal{Z}_s , the solution of the stance-phase differential equation remains in \mathcal{Z}_s until it intersects \mathcal{S}_s^f , at which time the flight phase is initiated. The family of horizontal surfaces on the right, \mathcal{Z}_{f,a_f} , (shown for two values of a_f , namely, $a_{f,1}$ and $a_{f,4}$) is created by the flight-phase controller; each surface is designed to be invariant in the usual sense that if the system is initialized in \mathcal{Z}_{f,a_f} , the

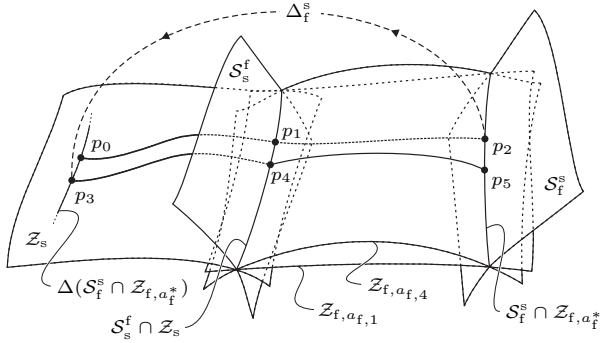


Fig. 2. Geometry of the closed-loop system that is achieved with the controller presented in this paper. Under the action of the feedback controller, the state of the system evolves on the low-dimensional surfaces \mathcal{Z}_s , \mathcal{Z}_{f,a_f} . Consider two strides of a running cycle initiated at the beginning of stance (at p_0). The robot's state evolves in \mathcal{Z}_s throughout stance until the state enters \mathcal{S}_s^f (at p_1), when two discrete events occur: the robot lifts off to begin the flight phase; and the controller selects a value of a_f that will be held constant throughout the flight phase. The value of the parameter, call it $a_{f,1}$, is determined as a function of the angular momentum about the stance leg end at transition into flight in such a way that the robot will land in a pre-determined configuration. During the flight phase, the state of the robot evolves in $\mathcal{Z}_{f,a_{f,1}}$ until it enters \mathcal{S}_f^s (at p_2). The impact mapping Δ_f^s is then applied and the next stance phase is begun (at p_3) and the process repeats. The robot will terminate the ensuing stance phase (at p_4) with an angular momentum that is different than what it had on the previous stride (at p_1). Due to conservation of angular momentum about the center of mass in the flight phase, the robot must evolve on a different surface in the flight phase this time in order to land in the same configuration it had on the previous landing. The required change in flight-phase evolution is accomplished by appropriate selection of $a_{f,4}$, which specifies $\mathcal{Z}_{f,a_{f,4}}$. By design, the rule for selecting a_f results in $\mathcal{S}_f^s \cap \mathcal{Z}_{f,a_{f,1}} = \mathcal{S}_f^s \cap \mathcal{Z}_{f,a_{f,4}}$. This common set has been denoted by $\mathcal{S}_f^s \cap \mathcal{Z}_{f,a_f^*}$, where a_f^* is the parameter value corresponding to a periodic orbit.

solution of the flight-phase differential equation remains in \mathcal{Z}_{f,a_f} until it intersects \mathcal{S}_f^s , at which time the stance phase is re-initialized through the impact map, Δ_f^s . The value of the parameter a_f is selected at each transition from stance to flight as a function of the angular momentum of the robot about the stance leg end, σ_1^{s-} , in such a way that evolution along the surface $\mathcal{Z}_{f,a_f(\sigma_1^{s-})}$ will cause the robot to land on \mathcal{Z}_s with a pre-determined configuration. This brings out a second form of invariance that is related to the hybrid nature of the system: the parameter a_f is selected so that the composition of the flight-phase dynamics with the impact map Δ_f^s maps $\mathcal{Z}_s \cap \mathcal{S}_f^s$ back to \mathcal{Z}_s . This is shown in Fig. 2 with the arrow looping from the end of the flight phase back to the initiation of the stance phase, and more succinctly in Fig. 3, where the composition of the flight-phase dynamics with the impact map is represented by a “generalized” impact map, Δ .

Not shown in either figure is the behavior of the robot off of the hybrid zero dynamics. This is where attractivity comes into play. In addition to creating the invariant surfaces, the feedback controller must also ensure that trajectories that start off of the surfaces converge to the surfaces. Both the creation of the invariant surfaces and their attractivity are accomplished with the use of virtual constraints.

C. Virtual constraints

Virtual constraints are holonomic constraints on the robot's configuration that are asymptotically imposed through feedback control. Their function is to coordinate the evolution

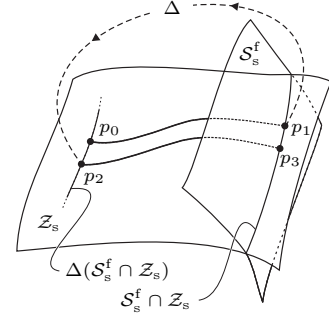


Fig. 3. Geometry of the closed-loop system when the flight phase and the impact map Δ_f^s are composed to form a generalized impact map Δ that maps $\mathcal{S}_f^s \cap \mathcal{Z}_s$ to \mathcal{Z}_s . This is analogous to the geometry that previous work has designed into walking. The analogy becomes exact if the robot's configuration at the initiation of the stance phase is the same for all points in $\mathcal{Z}_s \cap \mathcal{S}_f^s$.

of the various links throughout a stride—which is another way of saying that they reduce the degrees of freedom. It is the imposition of the constraints that creates the invariant surfaces. Since RABBIT has four independent actuators (two at the hips and two at the knees), four virtual constraints may be imposed in both the stance and flight phases. The virtual constraints used in this paper are constructed as follows. Consider a function pair $\{\theta(q), h_d(\theta)\}$, where $\theta: \mathcal{Q} \rightarrow \mathbb{R}$ is a scalar function of the configuration variables (in stance or flight) and $h_d: \mathbb{R} \rightarrow \mathbb{R}^4$ gives the desired configuration of the actuated joints. $\theta(q)$ should be selected to be monotonic along the planned motion and, for purposes of analysis, it is required that q_b and θ be independent, that is, $[q_b', \theta']$ is a valid change of coordinates on \mathcal{Q} . The virtual constraint is expressed in the form of an output $y = h(q) := q_b - h_d \circ \theta(q)$, and a controller is designed to asymptotically enforce the constraint by driving the output asymptotically to zero. Since the input and output have the same number of components, designing a controller to drive the output to zero is a standard problem [47]. The control action of driving the output to zero is what causes the trajectories of the system off of the invariant surface to converge to the surface.

VI. CONTROL LAW DEVELOPMENT

Separate state-variable control strategies are developed for the stance and flight phases of the running cycle. The controller for each phase is formulated as an output zeroing problem resulting in non-trivial zero dynamics [1], [3]. For the flight phase, it will be advantageous to allow the control to depend also on the initial value of the flight state or, equivalently, the final value of the stance state.

A. Stance phase control

As in [3, Sec. V], define the output

$$y_s = h_s(q) := q_b - h_{d,s} \circ \theta_s(q) \quad (33)$$

on (14), where the twice continuously differentiable function $h_{d,s}: \mathbb{R} \rightarrow \mathbb{R}^4$ encodes the stance-phase gait. It is assumed that the associated decoupling matrix³, $L_{g_s} L_{f_s} h_s$, is invertible,

³The control law is rewritten without the use of Lie derivatives in Appendix I.

$\Phi_s(q) := [h'_s, \theta_s]'$ is a global diffeomorphism on \mathcal{Q}_s ,

$$\mathcal{Z}_s := \{x_s \in T\mathcal{Q}_s \mid h_s(x_s) = 0, L_{f_s} h_s(x_s) = 0\} \quad (34)$$

is an embedded two-dimensional submanifold of $T\mathcal{Q}_s$, and $\mathcal{S}_s^f \cap \mathcal{Z}_s$ is an embedded one-dimensional submanifold of $T\mathcal{Q}_s$; see Fig. 2.

The feedback control is chosen to be continuous and to render \mathcal{Z}_s invariant under the closed-loop dynamics as well as attractive in finite time (the exact hypotheses from [3] are given in Appendix II):

$$u_s(x_s) = (L_{g_s} L_{f_s} h_s(x_s))^{-1} (v(h_s(x_s), L_{f_s} h_s(x_s)) - L_{f_s}^2 h_s(x_s)), \quad (35)$$

where v renders the origin of

$$\frac{d^2 y_s}{dt^2} = v \quad (36)$$

globally asymptotically stable with finite convergence time. The closed-loop system is denoted

$$f_{cl,s}(x_s) := f_s(x_s) + g_s(x_s)u_s(x_s). \quad (37)$$

The feedback control

$$u_s^*(x_s) = -(L_{g_s} L_{f_s} h_s(x_s))^{-1} L_{f_s}^2 h_s(x_s) \quad (38)$$

renders \mathcal{Z}_s invariant under the stance-phase dynamics; that is, for every $z \in \mathcal{Z}_s$,

$$f_{zero}(z) := f_s(z) + g_s(z)u_s^*(z) \in T_z \mathcal{Z}_s. \quad (39)$$

\mathcal{Z}_s is called the *stance-phase zero dynamics manifold* and $\dot{z} = f_{zero}(z)$ is called the *stance-phase zero dynamics*. Following the development in [1], [3], (θ_s, σ_1) is a valid set of local coordinates for \mathcal{Z}_s and in these coordinates the zero dynamics has the form

$$\begin{aligned} \dot{\theta}_s &= \frac{1}{\mathbf{I}(\theta_s)} \sigma_1, \\ \dot{\sigma}_1 &= mg \mathbf{x}_{cm}(\theta_s), \end{aligned} \quad (40)$$

where $\mathbf{I}(\theta_s)$ plays the role of an inertia. Moreover, in these coordinates, $\mathcal{S}_s^f \cap \mathcal{Z}_s$ is given by

$$\{(q_0^{s-}, \dot{q}^{s-}) \mid q_0^{s-} = \Phi_s^{-1}(0, \theta_{s,0}^-), \dot{q}^{s-} = \dot{q}_0^{s-} \sigma_1^{s-}, \sigma_1^{s-} \in \mathbb{R}\}, \quad (41)$$

where

$$\dot{q}_0^{s-} = \left[\begin{array}{c} \frac{\partial h_s}{\partial q} \\ A_5 \end{array} \right]^{-1} \Big|_{q_0^{s-}} \left[\begin{array}{c} 0_{4 \times 1} \\ 1 \end{array} \right]. \quad (42)$$

For later use in computing a Poincaré return map on the zero dynamics, it is noted that (40) has Lagrangian [3, Eq. (59)] $\mathcal{L}_{s,zero} := K_{s,zero} - V_{s,zero}$, where

$$K_{s,zero} := \frac{1}{2} (\sigma_1)^2 \quad (43)$$

$$V_{s,zero}(\theta_s) := - \int_{\theta_s^+}^{\theta_s} \mathbf{I}(\xi) mg \mathbf{x}_{cm}(\xi) d\xi; \quad (44)$$

the choice of the lower limit θ_s^+ is arbitrary and will be made later. Also for later use, define

$$\left[\begin{array}{c} \lambda_x(q_0^{s-}) \\ \lambda_y(q_0^{s-}) \end{array} \right] := \frac{\partial \mathbf{f}_1(q_0^{s-})}{\partial q} \dot{q}_0^{s-}, \quad (45)$$

so that

$$\left[\begin{array}{c} \dot{\mathbf{x}}_{cm}^{s-} \\ \dot{\mathbf{y}}_{cm}^{s-} \end{array} \right] \Big|_{\mathcal{S}_s^f \cap \mathcal{Z}_s} = \left[\begin{array}{c} \lambda_x(q_0^{s-}) \\ \lambda_y(q_0^{s-}) \end{array} \right] \sigma_1^{s-}. \quad (46)$$

B. Flight phase control

The overall goal of the flight-phase controller is to land the robot in a favorable manner for continuing with the stance phase. It will turn out that a particularly interesting objective is the following: if the robot enters the flight phase from the stance-phase zero dynamics manifold, \mathcal{Z}_s , control the robot so that it lands on \mathcal{Z}_s in a *fixed* configuration. The analytical motivation for this objective will be made clear in Section VII. The feasibility of landing in a fixed configuration will be illustrated in Section VIII with a feedback controller that depends on x_f and the final value of the state of the preceding stance phase. To realize such a controller as a state-variable feedback, the flight-state vector is augmented with dummy variables, $\dot{a}_f = 0$, whose values can be set at the transition from stance to flight. Let $a_f \in \mathcal{A} := \mathbb{R}^p$, $p \in \mathbb{N}$.

In other regards, paralleling the development of the stance phase controller⁴, define the output

$$y_f = h_f(q_f, a_f) := q_b - h_{d,f}(\mathbf{x}_{cm}, a_f), \quad (47)$$

where $h_{d,f}$ is at least twice differentiable. Then the following can be easily shown: for any value of a_f ,

- 1) the decoupling matrix, $L_{g_f} L_{f_f} h_f$, is everywhere invertible;
- 2) $\Phi_f := [h'_f, q_5, \mathbf{x}_{cm}, \mathbf{y}_{cm}]'$ is a global diffeomorphism on \mathcal{Q}_f ;
- 3) the flight-phase zero-dynamics manifold

$$\mathcal{Z}_{f,a_f} := \{x_f \in T\mathcal{Q}_f \mid h_f(x_f, a_f) = 0, L_{f_f} h_f(x_f, a_f) = 0\} \quad (48)$$

is a six-dimensional embedded submanifold of $T\mathcal{Q}_f$;

- 4) $\mathcal{S}_f^f \cap \mathcal{Z}_{f,a_f}$ is a five-dimensional embedded submanifold of $T\mathcal{Q}_f$;
- 5) $(q_5, \mathbf{x}_{cm}, \mathbf{y}_{cm}, \sigma_{cm}, \dot{\mathbf{x}}_{cm}, \dot{\mathbf{y}}_{cm})$ is a set of global coordinates for \mathcal{Z}_{f,a_f} ; and
- 6) the flight-phase zero dynamics has the form

$$\dot{q}_5 = \kappa_{1,f}(\sigma_{cm}, \mathbf{x}_{cm}, \dot{\mathbf{x}}_{cm}, a_f) \quad (49)$$

$$\dot{\sigma}_{cm} = 0 \quad (50)$$

$$\ddot{\mathbf{x}}_{cm} = 0 \quad (51)$$

$$\ddot{\mathbf{y}}_{cm} = -g, \quad (52)$$

where (49) arises from evaluating

$$\dot{q}_5 = \frac{\sigma_{cm}}{A_{55}(q_b)} - \sum_{i=1}^4 \frac{A_{5i}(q_b)}{A_{55}(q_b)} \dot{q}_i \quad (53)$$

on \mathcal{Z}_{f,a_f} . Note that in Fig. 2, only a two-dimensional projection of \mathcal{Z}_{f,a_f} could be shown and $\mathcal{S}_f^f \cap \mathcal{Z}_{f,a_f}$ was represented as a one-dimensional projection.

⁴As in the stance phase, the control law for the flight phase is rewritten without the use of Lie derivatives in Appendix I.

The feedback controller is defined as

$$u_f(x_f, a_f) := -(L_{g_f} L_{f_f} h_f(x_f, a_f))^{-1} \left(K_p h_f(x_f, a_f) + K_d L_{f_f} h_f(x_f, a_f) + L_{f_f}^2 h_f(x_f, a_f) \right), \quad (54)$$

where $\ddot{y}_f + K_d \dot{y}_f + K_p y_f = 0$ is exponentially stable. Let $\bar{x}_f := (x'_f, a'_f)'$ and denote the right-hand side of (7) and the trivial parameter dynamics $\dot{a}_f = 0$ in closed loop with (54) by

$$f_{cl,f}(\bar{x}_f) := \begin{bmatrix} f_f(x_f) + g_f(x_f) u_f(\bar{x}_f) \\ 0 \end{bmatrix}. \quad (55)$$

C. Closed-loop hybrid model

The closed-loop hybrid model is

$$\Sigma_{f,cl} : \begin{cases} \bar{\mathcal{X}}_f = T\mathcal{Q}_f \times \mathcal{A} \\ \bar{\mathcal{F}}_{cl,f} : (\bar{x}_f) = f_{cl,f}(\bar{x}_f) \\ \bar{\mathcal{S}}_f^s = \{(x_f, a_f) \in \bar{\mathcal{X}}_f \mid H_f^s(x_f) = 0\} \\ \bar{T}_f^s : x_s^+ = \bar{\Delta}_f^s(\bar{x}_f^-) := \Delta_f^s(x_f^-) \end{cases} \quad (56)$$

$$\Sigma_{s,cl} : \begin{cases} \mathcal{X}_s = T\mathcal{Q}_s \\ \mathcal{F}_{cl,s} : (\dot{x}_s) = f_{cl,s}(x_s) \\ \mathcal{S}_s^f = \{x_s \in \mathcal{X}_s \mid H_s^f(x_s) = 0\} \\ \bar{T}_f^f : x_f^+ = \Delta_s^f(x_s^-), a_f^+ = w_s^f(x_s^-), \end{cases}$$

where the parameter update law w_s^f is at least continuously differentiable. The internal geometry of the closed-loop system is shown in Fig. 2.

VII. EXISTENCE AND STABILITY OF PERIODIC ORBITS

This section of the paper could equally well be titled, ‘‘Main Results’’, as it is here that the internal geometric structure of the closed-loop system (56) is exploited to obtain a low-dimensional, closed-form characterization of asymptotically stable, periodic, running motions. Section VII-A develops the Poincaré section and the Poincaré return map that will be used for analyzing periodic orbits of (56). The analytical results based on restriction dynamics—that is, the hybrid zero dynamics of Fig. 3—are developed in Section VII-B. The relation to stability in the full-order model is treated in Section VII-C.

A. Definition of the Poincaré return map

A periodic running motion corresponds to a periodic orbit in the closed-loop model. The Poincaré return map is a well known tool for determining the existence of periodic orbits and their stability properties; for its use in hybrid systems, see [2], [45], [48], [49]. A fixed point of the Poincaré return map is equivalent to a periodic orbit.

Since running consists of two phases, the Poincaré return map is naturally viewed as the composition of two maps: $P := P_s \circ P_f$, as in Fig. 4, where $P_s : \bar{\mathcal{S}}_f^s \rightarrow \mathcal{S}_s^f$ follows a solution of the closed-loop model from the impact event at the end of flight to just before the end of stance, and $P_f : \mathcal{S}_s^f \rightarrow \bar{\mathcal{S}}_f^s$ follows a solution from the end-of-stance event to just before the end of flight. Since not every landing will result in the

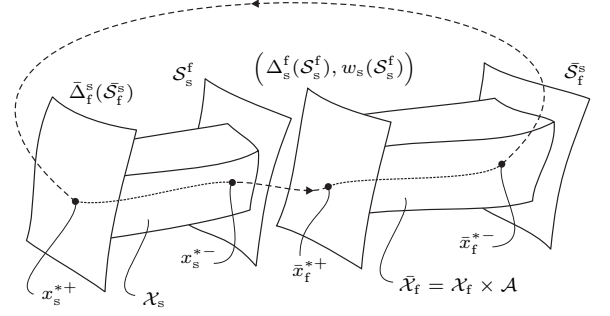


Fig. 4. Poincaré maps for the closed-loop system. Conceptually, $P_s : \bar{\mathcal{S}}_f^s \rightarrow \mathcal{S}_s^f$, and is determined by following the flow of the closed-loop stance model from the impact at the end of flight up to, and not including, the transition from stance to flight. Similarly, $P_f : \mathcal{S}_s^f \rightarrow \bar{\mathcal{S}}_f^s$, and is determined by following the flow of the closed-loop flight model augmented with the trivial parameter dynamics ($\dot{a}_f = 0$) from the transition at the end of stance up to, and not including, the impact event at end of flight. The Poincaré map is $P : \mathcal{S}_s^f \rightarrow \bar{\mathcal{S}}_f^s$, where $P := P_s \circ P_f$. A periodic orbit corresponds to a fixed point of P , namely, $x_s^{*-} = P(x_s^{*-})$.

completion of a stance phase (for example, the robot may not have sufficient speed), P_s is only a partial map. And, since not every transition out of stance results in a successful flight phase followed by a successful stance phase, P_f is a partial map. Hence, the domains where these maps are well defined must be identified, which will give rise to the subsets $\tilde{\mathcal{S}}_s^f \subset \mathcal{S}_s^f$ and $\tilde{\mathcal{S}}_f^s \subset \bar{\mathcal{S}}_f^s$.

Following [2], define the *stance-time-to-impact function*⁵, $T_{I,s} : x_0 \in \mathcal{X}_s \rightarrow \mathbb{R} \cup \{\infty\}$, by

$$T_{I,s} := \begin{cases} \inf\{t \geq 0 \mid \varphi_{cl,s}(t, x_0) \in \mathcal{S}_s^f\} & \text{if } \exists t \text{ such that} \\ & \varphi_{cl,s}(t, x_0) \in \mathcal{S}_s^f \\ \infty & \text{otherwise,} \end{cases} \quad (57)$$

where $\varphi_{cl,s}(t, x_0)$ is an integral curve of (37) corresponding to $\varphi_{cl,s}(0, x_0) = x_0$. From [2, Lemma 3], $T_{I,s}$ is continuous at points x_0 where $0 < T_{I,s}(x_0) < \infty$ and the intersection with \mathcal{S}_s^f is transversal⁶. Hence, $\tilde{\mathcal{X}}_s := \{x_s \in \mathcal{X}_s \mid 0 < T_{I,s}(x_s) < \infty, L_{f_{cl,s}} H_s^f(\varphi_{cl,s}(T_{I,s}(x_s), x_s)) \neq 0\}$ is open, and consequently, $\tilde{\mathcal{S}}_f^s := \bar{\Delta}_f^s^{-1}(\tilde{\mathcal{X}}_s)$ is an open subset of $\bar{\mathcal{S}}_f^s$. It follows that the *generalized Poincaré stance map* $P_s : \tilde{\mathcal{S}}_f^s \rightarrow \mathcal{S}_s^f$ defined by

$$P_s(\bar{x}_f) := \varphi_{cl,s}(T_{I,s}(\bar{\Delta}_f^s(\bar{x}_f)), \bar{\Delta}_f^s(\bar{x}_f)), \quad (58)$$

is well defined and continuous (the terminology of a *generalized-Poincaré map* follows Appendix D of [50]).

Similarly, the *generalized Poincaré flight map* $P_f : \tilde{\mathcal{S}}_s^f \rightarrow \bar{\mathcal{S}}_f^s$, is defined by

$$P_f(x_s) := \varphi_{cl,f}(T_{I,f}(\Delta_s^f(x_s), w_s^f(x_s)), \Delta_s^f(x_s), w_s^f(x_s)), \quad (59)$$

⁵Flows from one surface to another are sometimes called impact maps or impact functions. $T_{I,s}$ could also be called the *time-to-flight function*.

⁶Transversality guarantees nonzero vertical leg end velocity at impact; that is, the foot does not just scuff the ground.

where, $T_{I,f} : \bar{x}_0 \in \bar{\mathcal{X}}_f \rightarrow \mathbb{R} \cup \{\infty\}$ by

$$T_{I,f} := \begin{cases} \inf\{t \geq 0 \mid \varphi_{\text{cl},f}(t, \bar{x}_0) \in \tilde{\mathcal{S}}_f^s\} & \text{if } \exists t \text{ such that} \\ & \varphi_{\text{cl},f}(t, \bar{x}_0) \in \tilde{\mathcal{S}}_f^s \\ \infty & \text{otherwise,} \end{cases} \quad (60)$$

and $\tilde{\mathcal{S}}_f^s = \{x_s \in \mathcal{S}_s^f \mid 0 < T_{I,f}(\Delta_s^f(x_s), w_s^f(x_s)) < \infty, L_{f,\text{cl},f} H_f^s(\varphi_{\text{cl},f}(T_{I,f}(\Delta_s^f(x_s), w_s^f(x_s)), \Delta_s^f(x_s), w_s^f(x_s))) \neq 0\}$. In [50, Appendix D], it is proved that P_f is continuously differentiable.

The Poincaré return map $P : \tilde{\mathcal{S}}_f^s \rightarrow \mathcal{S}_s^f$ for (56) is defined by

$$P := P_s \circ P_f. \quad (61)$$

B. Analysis of the Poincaré return map

Theorem 1 (Connecting running to walking): Let P be as (61), and let $\mathcal{S} := \mathcal{S}_s^f$, and $\tilde{\mathcal{S}} := \tilde{\mathcal{S}}_f^s$. Then P is also the Poincaré return map for the system with impulse effects

$$\Sigma_{\text{cl}} : \begin{cases} \dot{x}_s(t) = f_{\text{cl},s}(x_s(t)) & x_s^-(t) \notin \mathcal{S} \\ x_s^+(t) = \Delta(x_s^-(t)) & x_s^-(t) \in \mathcal{S}, \end{cases} \quad (62)$$

where $\Delta := \bar{\Delta}_f^s \circ P_f$.

Proof: This follows from the construction of the Poincaré return map in [2, Eq. (14)]. ■

This observation is important because models of the form (62)—called systems with impulse effects [51], [52]—have been studied in the context of walking gaits [3], [4]. The association of running with walking may indicate how results developed for walking, such as closed-form stability analysis on the basis of a restricted Poincaré map, may be extended to running. One motivation for pursuing this association is the fact that the control law designs of [3], [4] have been validated experimentally [8], [9]. In this section and the next, several results along this line of reasoning are developed and illustrated on an asymptotically stable running gait.

Suppose that $\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s$, where \mathcal{Z}_s is the stance-phase zero dynamics manifold. Then, from [3], (62) has a hybrid zero dynamics, which may be called the *hybrid zero dynamics of running*:

$$\begin{aligned} \dot{z} &= f_{\text{zero}}(z) & z^- \notin \mathcal{S} \cap \mathcal{Z}_s \\ z^+ &= \Delta_{\text{zero}}(z^-) & z^- \in \mathcal{S} \cap \mathcal{Z}_s, \end{aligned} \quad (63)$$

where the restricted impact map is $\Delta_{\text{zero}} := \Delta|_{\tilde{\mathcal{S}} \cap \mathcal{Z}_s}$ and f_{zero} is given by (39). The key properties in walking gaits that led to a rich analytic theory were *mathcal{Z}_s*-invariance, $\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s$, and what one may call *configuration determinism at transition*: $\pi \circ \Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s)$ consists of a single point, where $\pi : T\mathcal{Q}_s \rightarrow \mathcal{Q}_s$ is the canonical projection. How to achieve these conditions for $\Delta = \bar{\Delta}_f^s \circ P_f$ through design of the flight-phase controller will be detailed in Section VIII.

Let q_0^{s-} be as defined in (41) and define $q_0^{s+} := \pi \circ \Delta(q_0^{s-}, *)$. Use (8) to define the positions of the center of mass at the beginning of the stance phase, $(\mathbf{x}_{\text{cm}}^{s+}, \mathbf{y}_{\text{cm}}^{s+})$, and at the end of the stance phase, $(\mathbf{x}_{\text{cm}}^{s-}, \mathbf{y}_{\text{cm}}^{s-})$. In the following, it is assumed that the center of mass is behind the stance leg at the beginning of the stance phase, and thus, $\mathbf{x}_{\text{cm}}^{s+} < 0$.

Theorem 2 (Characterization of restricted impact map): Suppose that $\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s$ and $\pi \circ \Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) = \{q_0^{s+}\}$. In the coordinates (θ_s, σ_1) for \mathcal{Z}_s , the restricted impact map is given by

$$\Delta_{\text{zero}}(\theta_{s,0}^-, \sigma_1^{s-}) = \begin{bmatrix} \theta_{s,0}^+ \\ \delta(\sigma_1^{s-}) \end{bmatrix}, \quad (64)$$

where

$$\begin{aligned} \theta_{s,0}^+ &= \theta_s(q_0^{s+}) \\ \delta(\sigma_1^{s-}) &= \chi \sigma_1^{s-} - \sqrt{(\beta \sigma_1^{s-})^2 + \alpha}, \end{aligned} \quad (65)$$

and

$$\begin{aligned} \alpha &= -2m^2 g(\mathbf{x}_{\text{cm}}^{s+})^2 (\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-}) \\ \beta &= m \mathbf{x}_{\text{cm}}^{s+} \lambda_y(q_0^{s-}) \end{aligned} \quad (66)$$

$\chi = 1 + m \mathbf{x}_{\text{cm}}^{s-} \lambda_y(q_0^{s-}) + m(\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-}) \lambda_x(q_0^{s-})$. The proof is given in Appendix III.

Remark 4:

- 1) $\chi < 0$ would imply a sign change in the angular momentum at impact, which would be incompatible with the definition of running as forward motion with a flight phase.
- 2) If $\mathbf{x}_{\text{cm}}^{s+} < 0$ is not assumed, the general expression for (65) is $\delta(\sigma_1^{s-}) = \chi \sigma_1^{s-} + \text{sgn}(\mathbf{x}_{\text{cm}}^{s+}) \sqrt{(\beta \sigma_1^{s-})^2 + \alpha}$, where $\text{sgn}(x)$ is the sign of x .
- 3) When $\alpha = 0$, that is, the center of mass has the same height at the beginning and end of the stance phase, $\delta(\sigma_1^{s-}) = (\chi - |\beta|) \sigma_1^{s-}$ is linear, exactly as in walking.
- 4) In terms of the coordinates $(\theta_{s,0}^-, \zeta := \frac{1}{2}(\sigma_1^{s-})^2)$ for $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$, where the (generalized) kinetic energy of the stance-phase zero dynamics is used instead of the angular momentum, the second entry in (64) becomes

$$\begin{aligned} \delta_e(\zeta) &= (\chi^2 + \beta^2) \zeta \\ &\quad - \chi \sqrt{2\alpha\zeta + (2\beta\zeta)^2} + \frac{\alpha}{2}. \end{aligned} \quad (67)$$

- 5) Implicit in the construction of $\tilde{\mathcal{S}}$ is the condition $2\alpha\zeta + (2\beta\zeta)^2 \geq 0$. Also a part of the construction of $\tilde{\mathcal{S}}$ is the condition that $T_{I,f}$ is a positive real number; under the assumptions made on Δ , this is equivalent to checking that $\mathbf{y}_{\text{cm}}^{s+} > \mathbf{y}_{\text{cm}}^{s-}$ and $\lambda_y(q_0^{s-}) < 0$ do not simultaneously occur.

Let $P : \tilde{\mathcal{S}} \rightarrow \mathcal{S}$ be the Poincaré return map for (62), and hence, also for (56), and suppose that $\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s$, as in Fig. 3. Then $P(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{S} \cap \mathcal{Z}_s$, and the restriction map $\rho : \tilde{\mathcal{S}} \cap \mathcal{Z}_s \rightarrow \mathcal{S} \cap \mathcal{Z}_s$ by

$$\rho := P|_{\tilde{\mathcal{S}} \cap \mathcal{Z}_s} \quad (68)$$

is well defined. The restricted Poincaré return map ρ is important because it is scalar and, by [2, Theorem 2] (see [3, Sec. IV]), asymptotically stable fixed points of it correspond to asymptotically stable periodic orbits of the hybrid model (62), and hence, to asymptotically stable running gaits.

Theorem 3 (Closed-form for ρ): Suppose that $\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s$ and $\pi \circ \Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) = \{q_0^{s+}\}$. Let $(\theta_{s,0}^-, \sigma_1^{s-}) \in \tilde{\mathcal{S}} \cap \mathcal{Z}_s$, and set $\zeta := \frac{1}{2}(\sigma_1^{s-})^2$. Then

$$\rho(\zeta) = \delta_e(\zeta) - V_{s,\text{zero}}(\theta_{s,0}^-), \quad (69)$$

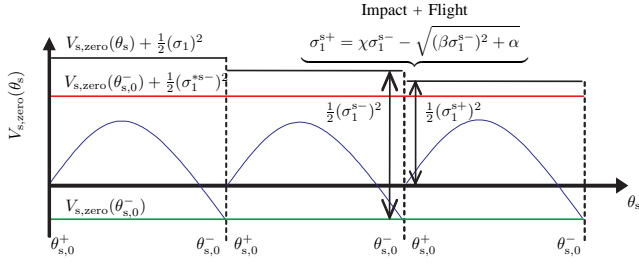


Fig. 5. The stance phase zero dynamics is Lagrangian, and thus throughout the stance phase, the corresponding total energy $V_{s,zero}(\theta_s) + \frac{1}{2}(\sigma_1)^2$ is constant. Over the impact plus flight phase, the change in total energy depends on the angular momentum through $\delta(\sigma_1^{s-})$ and the potential energy through $V_{s,zero}(\theta_{s,0}^-)$. The total energy corresponding to the periodic orbit is $V_{s,zero}(\theta_{s,0}^-) + \frac{1}{2}(\sigma_1^{s-})^2$.

with domain of definition

$$\mathcal{D}_\rho := \left\{ \zeta > 0 \mid \delta_e(\zeta) - V_{s,zero}^{\max} > 0, \right. \\ \left. 2\alpha\zeta + (2\beta\zeta)^2 \geq 0 \right\}, \quad (70)$$

where δ_e is defined in (67), and

$$V_{s,zero}^{\max} := \max_{\theta_{s,0}^+ \leq \theta_s \leq \theta_{s,0}^-} V_{s,zero}(\theta_s). \quad (71)$$

Moreover, the first derivative of the restricted Poincaré return map is

$$\frac{d\rho}{d\zeta}(\zeta) = \frac{d\delta_e}{d\zeta}(\zeta) = (\chi^2 + \beta^2) - \chi \frac{\alpha + 4\beta^2\zeta}{\sqrt{2\alpha\zeta + (2\beta\zeta)^2}}. \quad (72)$$

The proof is given in Appendix IV.

Remark 5: 1) Computing a fixed point of (69) is easily reduced to solving a quadratic equation. If its discriminant Υ is non-negative, where

$$\Upsilon := 4\chi^2 \left(\chi^2 \alpha^2 \right. \\ \left. + (-2V_{s,zero}(\theta_{s,0}^-) + \alpha) (-\alpha\chi^2 + \alpha - 2\beta^2 V_{s,zero}(\theta_{s,0}^-)) \right), \quad (73)$$

the fixed point can be explicitly calculated as

$$\zeta^* = \frac{(\chi^2 + \beta^2 - 1)(2V_{s,zero}(\theta_{s,0}^-) - \alpha) + 2\chi^2\alpha - \sqrt{\Upsilon}}{2((\chi + \beta)^2 - 1)((\chi - \beta)^2 - 1)}. \quad (74)$$

2) As in walking [1, Fig. 14], the restricted Poincaré map can be interpreted in terms of energy transfer; see Fig. 5.

The following two corollaries are immediate.

Corollary 1 (Exponentially stable fixed points):

Suppose that $\zeta^* \in \mathcal{D}_\rho$ is a fixed point of ρ . Then it is exponentially stable if, and only if,

$$\mu := (\chi^2 + \beta^2) - \chi \frac{\alpha + 4\beta^2\zeta^*}{\sqrt{2\alpha\zeta^* + (2\beta\zeta^*)^2}} \quad (75)$$

satisfies $|\mu| < 1$.

Corollary 2 (Qualitative analysis of ρ): The following statements are true:

- (a) $\lim_{\zeta \searrow 0} \frac{d\rho}{d\zeta}(\zeta) = -\infty$, for $\chi > 0$ and $\alpha \geq 0$;

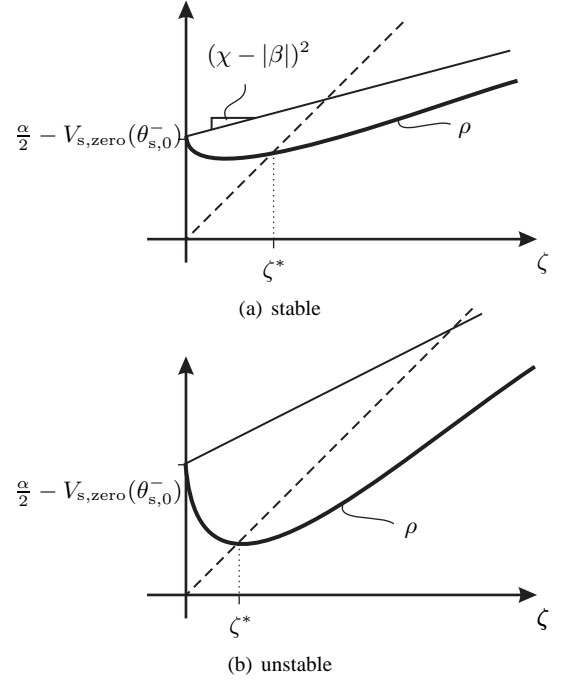


Fig. 6. Qualitatively different Poincaré maps that may occur in running. The dashed line is the identity map and the bold line is a sketch of the restricted Poincaré return map. In (a), the fixed point is exponentially stable because the intersection with the identity line occurs with a positive slope less than 1.0. In (b), the fixed point is unstable because the intersection with the identity line occurs with a negative slope less than -1.0 .

- (b) $\lim_{\zeta \searrow \frac{-\alpha}{2\beta^2}} \frac{d\rho}{d\zeta}(\zeta) = -\infty$, for $\chi > 0$ and $\alpha < 0$;
- (c) $\lim_{\zeta \rightarrow \infty} \frac{d\rho}{d\zeta}(\zeta) = (\chi - |\beta|)^2$;
- (d) $\frac{d^2\rho}{d\zeta^2}(\zeta) = \chi \frac{\alpha^2}{(2\alpha\zeta + 4\beta^2\zeta^2)^{3/2}}$ does not change sign.

Fig. 6 provides a graphical depiction of ρ for $\chi > 0$, $\alpha \geq 0$, and $\frac{\alpha}{2} - V_{s,zero}(\theta_{s,0}^-) > 0$. Similar figures could be drawn for other cases. The next result shows that these qualitative features of the Poincaré return map lead to a large region of attraction for an exponentially stable fixed-point.

Theorem 4 (Non-local convergence in the HZD):

Consider $\rho: \mathcal{D}_\rho \rightarrow \mathbb{R}$, and suppose that

- 1) $(\chi - |\beta|)^2 < 1$,
- 2) $\chi > 0$,
- 3) and there exists $\zeta^* \in \mathcal{D}_\rho$ such that $\rho(\zeta^*) = \zeta^*$ and $\frac{d\rho}{d\zeta}(\zeta^*) > 0$.

Then, the following statements are true:

- (a) ζ^* is the unique fixed-point of ρ ;
- (b) the set

$$\tilde{\mathcal{D}}_\rho = \left\{ \zeta \in \mathcal{D}_\rho \mid \frac{d\rho}{d\zeta}(\zeta) > 0 \right\} \quad (76)$$

is unbounded and connected; and

- (c) ζ^* is locally exponentially stable and every solution of $\zeta(k+1) = \rho(\zeta(k))$ initialized in $\tilde{\mathcal{D}}_\rho$ converges monotonically to ζ^* .

The proof is given in Appendix V. This result shows that once the motion of the robot has settled to the hybrid zero dynamics, the domain of attraction of the periodic orbit is quite large. The analysis in Theorem 4 has not accounted for

	Femur	Tibia	Torso
Length (m)	0.4	0.4	0.625
Mass (kg)	6.8	3.2	17.0
Inertia about COM (kg-m ²)	0.47	0.20	1.33

TABLE I
EXPERIMENTALLY MEASURED PARAMETERS FOR RABBIT.

the peak torque of the actuators and the allowed friction cone at the support leg end. This theorem should thus be viewed as stating that such physical considerations will determine the limits on the region of attraction, and that the semi-global convergence of the control loop *per se* is not the key limiting factor.

For all of the examples worked by the authors, if an exponentially stable fixed point was found, hypotheses (1), (2) and (3) of Theorem 4 have always held as well. In particular, μ was always greater than 0.4 and \tilde{D}_ρ equalled D_ρ , that is, the Poincaré map was always strictly increasing on the region of interest. In the case of Raibert's hopper, the Poincaré map was unimodal—and thus not strictly increasing on the domain of interest [17]. Nevertheless, semi-global stability was established using a more powerful analysis method due to Singer [53] and Guckenheimer [54].

C. Stability in the full-order model

Under the assumptions made in Section VI-A on the stance-phase controller, Theorem 2 of [2] guarantees that asymptotically stable orbits in the hybrid zero dynamics (63) are also asymptotically stable in the full-order hybrid model (62) (for a discussion of exponential stability, see Remark 1 in [13]). By Theorem 1, the Poincaré return maps of (62) and (56) are identical. Hence, once asymptotic stability of an orbit has been proven in (62), it follows that the orbit is also asymptotically stable for (56). Putting all of this together, proving the existence and stability of an orbit in the hybrid zero dynamics of running leads to the desired conclusions in the state space of the closed-loop hybrid model (56).

VIII. ILLUSTRATION ON RABBIT

The analytical results of Section VII make it straightforward to determine if a control law of the kind specified in Section VI leads to the existence of a stable periodic orbit. However, proposing specific values for the output functions so that the evolution of the robot is energetically efficient, while respecting actuator limits, the friction cone at the contact point of the leg end, and lift-off at the beginning of the flight phase, is nearly impossible to do by intuition. Here, the feedback designs will be based on optimization.

Using the method proposed in [11], time-trajectories of (29), corresponding to average running speeds varying from 0.5 m/s to 2.75 m/s and parameter values given in Tab. I, were determined for RABBIT (see [1] for details on the planar, bipedal robot, RABBIT). The running trajectories satisfy $\dot{y}_1 > 0$ at the beginning of the flight phase, the duration of the flight phase

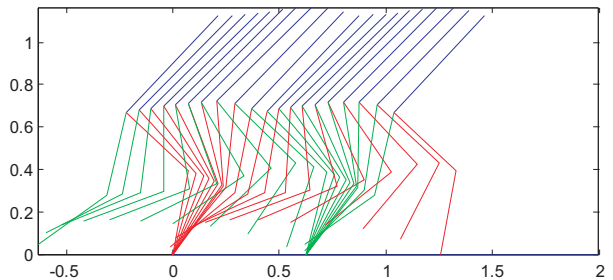


Fig. 7. Stick diagram for a running trajectory with average speed 1.5 m/s.

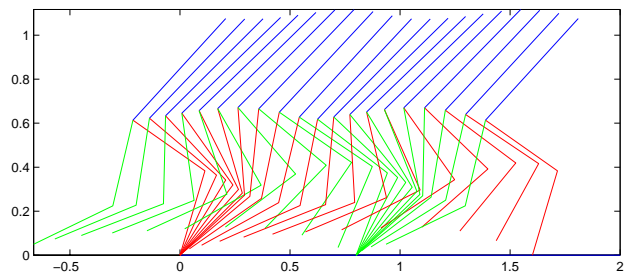


Fig. 8. Stick diagram for a running trajectory with average speed 2.5 m/s.

is at least 25% of the duration of a stride, and the required coefficient of friction is less than 2/3. Stick-figure diagrams corresponding to the running motions of 1.5 m/s and 2.5 m/s are given in Figs. 7 and 8.

Denote by \mathcal{O} the path traced out in the state spaces of the hybrid model of the robot by any one of these running trajectories. It was checked that \mathcal{O} intersects \mathcal{S}_s^f and \mathcal{S}_f^s exactly once; define x_f^{*-} and x_s^{*-} as the initial and final configurations on the periodic orbit, respectively. The goal is to design a time-invariant state-feedback controller *à la* Section VI that has \mathcal{O} as its asymptotically-stable periodic orbit. Recall that designing the controller is equivalent to specifying the output functions in (33) and (47) and the parameter update-law in (56).

A. Stance Phase Controller Design

On the basis of x_f^{*-} and x_s^{*-} , the values of q_0^{s+} (the initial configuration in stance on the periodic orbit), q_0^{s-} (final configuration in stance on the periodic orbit), \dot{q}_0^{s+} (normalized initial velocity in stance on the periodic orbit; see (42)), and \dot{q}_0^{s-} (normalized final velocity in stance on the periodic orbit⁷) are easily deduced, which in turn give the initial and final values of θ_s on the periodic orbit, $\theta_{s,0}^+$ and $\theta_{s,0}^-$.

As in [5], an output $y_s = h_s(q) := q_b - h_{d,s} \circ \theta_s(q)$ is designed so that it satisfies the boundary condition and vanishes (nearly) along the stance phase of the periodic orbit, and thus the orbit will be an integral curve of the stance-phase zero dynamics. For this, the function $h_{d,s}$ was selected to be a fourth-order polynomial in θ_s . The design method in [11] that is used to compute the periodic orbit essentially guarantees that the technical conditions of Section VI are satisfied for h_s ; nevertheless, the conditions were formally verified. Once h_s is known, so is \mathcal{Z}_s , and, by construction, $\mathcal{O} \cap TQ_s \subset \mathcal{Z}_s$.

⁷In (42), replace evaluation at q_0^{s-} with q_0^{s+} .

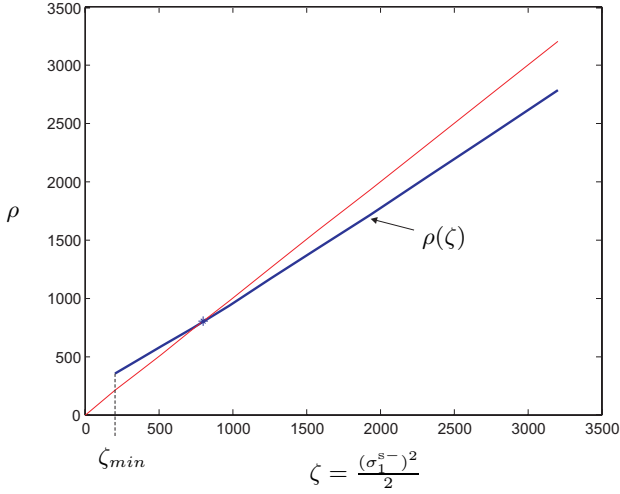


Fig. 9. Running at 1.5 m/s. The restricted Poincaré map (bold) associated with the closed-loop system. The fixed point occurs where the graph of ρ intersects the graph of the identity map (thin line).

B. Stability of the periodic orbits

The data required to determine the restricted Poincaré map ρ in Theorem 3 and Theorem 4 can be computed directly from $h_{d,s}$. This was carried out for each of the running trajectories studied in this paper. The numerical values are summarized in Tab. II. In each case, $\mu < 1$ and hence if a flight-phase controller can be determined to meet the conditions of Theorem 3, the corresponding orbit will be asymptotically stable. Note that slower running speeds yield smaller values of μ . So, for fast running, the convergence toward the periodic orbit will be slow. A plot of the restricted Poincaré map is provided in Fig. 9 for the trajectory corresponding to an average speed of 1.5 m/s.

C. Flight Phase Controller Design

The flight phase controller, $y_f = h_f(q_f, a_f) := q_b - h_{d,f}(\mathbf{x}_{cm}, a_f)$, $a_f = w_s^f(x_s^-)$, is to be designed so that trajectories of the closed-loop system that takeoff from the stance-phase zero dynamics manifold, \mathcal{Z}_s , land on \mathcal{Z}_s ; moreover, the landing configuration should be independent of the robot's takeoff velocity from \mathcal{Z}_s . Since from Section VIII-A the initial stance-phase configuration of the robot on the periodic orbit is equal to q_0^{s+} , these two conditions become

$$\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s \quad (77)$$

$$\pi \circ \Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) = q_0^{s+}, \quad (78)$$

where, as before, $\pi : T\mathcal{Q}_s \rightarrow \mathcal{Q}_s$ is the canonical projection. The design of the controller can now be broken down into several steps. First, (77) and (78) will be translated from boundary conditions on configuration and velocity at the *beginning* of the (next) stance phase, into boundary conditions at the *end* of the (current) flight phase. This will result in control objectives for the configuration and velocity of the body coordinates and for the overall orientation of the robot at landing. In a second step, because the body coordinates q_b are directly actuated, it is straightforward to design a family of

functions $h_{d,f}(\mathbf{x}_{cm}, a_f)$ that achieve the boundary conditions on the body-coordinate configuration and velocity, once the flight duration is determined from the ballistic motion of the robot's center of mass. The final step is more difficult because it is indirect: adjust the evolution of the body coordinates as a function of the takeoff velocity so as to achieve a desired orientation q_5 of the robot at landing.

To begin the first step, observe that because (q_0^{s+}, \dot{q}) is in $\pi^{-1}(q_0^{s+}) \cap \mathcal{Z}_s$ if, and only if, $\dot{q} = \dot{q}_0^{s+} \sigma_1^{s+}$ for some $\sigma_1^{s+} \in \mathbb{R}$, and (q_0^{s-}, \dot{q}) is in $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$ if, and only if, $\dot{q} = \dot{q}_0^{s-} \sigma_1^{s-}$ for some $\sigma_1^{s-} \in \mathbb{R}$, conditions (77) and (78) are equivalent to

$$\forall \sigma_1^{s-}, \exists \sigma_1^{s+} \text{ s.t. } \Delta(q_0^{s-}, \dot{q}_0^{s-} \sigma_1^{s-}) = (q_0^{s+}, \dot{q}_0^{s+} \sigma_1^{s+}). \quad (79)$$

From Theorem 2, it follows that $\sigma_1^{s+} = \delta(\sigma_1^{s-})$, and hence (79) is equivalent to

$$\Delta(q_0^{s-}, \dot{q}_0^{s-} \sigma_1^{s-}) = (q_0^{s+}, \dot{q}_0^{s+} \delta(\sigma_1^{s-})), \quad (80)$$

which gives specific boundary conditions, just *after* impact, to be met by the design of the flight phase controller. In particular, recalling that $q = (q'_b, q_5)'$, it is seen that (80) places constraints on the body configuration variables and their derivatives, and on the overall orientation of the robot, q_5 , while the constraint on \dot{q}_5 is equivalent to $\sigma_1^{s+} = \delta(\sigma_1^{s-})$, if the other constraints are met.

For the purpose of computation, it is convenient to transform (80) to conditions in the flight-phase state space, $T\mathcal{Q}_f$, instead of the stance-phase state space, $T\mathcal{Q}_s$. This is done as follows: the boundary conditions (80) specify the height of the center of mass at impact, and from this information, the flight time, t_f , is computed for any initial condition in $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$; see (100) in the Appendix. Using (101) and (46), the velocity of the center of mass can be expressed as a function of σ_1^{s-} ,

$$\begin{bmatrix} \dot{\mathbf{x}}_{cm}^{f-} \\ \dot{\mathbf{y}}_{cm}^{f-} \end{bmatrix} = \begin{bmatrix} \lambda_x(q_0^{s-}) \sigma_1^{s-} \\ -\sqrt{(\lambda_y(q_0^{s-}) \sigma_1^{s-})^2 - 2g(\mathbf{y}_{cm}^{s+} - \mathbf{y}_{cm}^{s-})} \end{bmatrix}. \quad (81)$$

The impact model (20), can be rewritten to define the angular velocity at the end of flight satisfying (80):

$$\dot{q}^{f-} = A^{-1} \left(A + m \frac{\partial \mathbf{f}_2'}{\partial q} \frac{\partial \mathbf{f}_2}{\partial q} \right) R^{-1} \dot{q}_0^{s+} \delta(\sigma_1^{s-}) + mA^{-1} \frac{\partial \mathbf{f}_2'}{\partial q} \begin{bmatrix} \dot{\mathbf{x}}_{cm}^{f-} \\ \dot{\mathbf{y}}_{cm}^{f-} \end{bmatrix}. \quad (82)$$

These last two equations define a function $\dot{q}_0(q_0^{s+}, \sigma_1^{s-})$ such that (80) is equivalent to

$$\begin{aligned} q_0^{f-} &= R^{-1} q_0^{s+} \\ \dot{q}^{f-} &= \dot{q}_0(q_0^{s+}, \sigma_1^{s-}). \end{aligned} \quad (83)$$

In summary, the objective of the flight-phase controller is to meet the boundary conditions given in (83). Meeting these two conditions will assure that invariance of \mathcal{Z}_s under the composition of the flight phase and impact model is achieved, (77), and that configuration determinism at transition, (78), is also met; see Figs. 2 and 3.

Average velocity	$V_{\text{zero}}(\theta_s^-)$	$V_{s,\text{zero}}^{\text{max}}$	$\mathbf{x}_{\text{cm}}^{s+}$ (cm)	$\mathbf{y}_{\text{cm}}^{s+}$ (cm)	$\mathbf{x}_{\text{cm}}^{s-}$ (cm)	$\mathbf{y}_{\text{cm}}^{s-}$ (cm)	$\lambda_x(q_0^{s-})$ (10^{-2})	$\lambda_y(q_0^{s-})$ (10^{-3})
0.50 m/s	-66	21	-6.8	62.4	14.0	69.8	3.74	5.4
0.75 m/s	-114	36	-8.8	62.1	18.4	68.8	3.83	3.3
1.00 m/s	-168	54	-10.9	61.5	22.6	67.5	3.95	2.3
1.25 m/s	-219	74	-12.9	60.5	26.4	65.7	4.09	2.0
1.50 m/s	-258	100	-15.1	59.3	29.6	63.8	4.27	2.3
1.75 m/s	-274	134	-17.7	58.1	32.3	61.7	4.48	3.0
2.00 m/s	-285	167	-20.1	56.7	34.6	59.7	4.69	3.3
2.25 m/s	-306	123	-17.5	55.6	34.0	59.1	4.78	3.9
2.50 m/s	-309	81	-14.4	54.7	32.5	59.0	4.85	5.0
2.75 m/s	-260	70	-13.2	55.2	29.8	58.6	4.91	5.0

Average velocity	α	β (10^{-2})	χ	ζ_{min}	ζ^*	μ	$\chi - \beta $
0.50 m/s	9.12	-1.37	0.926	53	151	0.695	0.832
0.75 m/s	14.26	-1.07	0.926	88	275	0.708	0.838
1.00 m/s	19.04	-0.92	0.931	125	434	0.729	0.850
1.25 m/s	23.34	-0.96	0.940	164	615	0.754	0.866
1.50 m/s	27.33	-1.29	0.955	206	801	0.785	0.887
1.75 m/s	30.84	-1.99	0.976	253	982	0.826	0.914
2.00 m/s	32.77	-2.47	0.990	294	1162	0.856	0.932
2.25 m/s	29.56	-2.52	0.986	231	1327	0.859	0.922
2.50 m/s	23.69	-2.66	0.984	161	1503	0.870	0.916
2.75 m/s	15.91	-2.45	0.994	127	1729	0.908	0.940

TABLE II

STABILITY ANALYSIS OF VARIOUS RUNNING MOTIONS. IF $\zeta > \zeta_{\text{min}}$, THEN $\zeta \in \mathcal{D}_p$.

The design of $h_{d,f}$ can now be given in two more steps. First, define⁸

$$\tau(\mathbf{x}_{\text{cm}}, \sigma_1^{s-}) = \frac{\mathbf{x}_{\text{cm}} - \mathbf{x}_{\text{cm}}^{f+}}{t_f \dot{\mathbf{x}}_{\text{cm}}^{f+}} = \frac{\mathbf{x}_{\text{cm}} - \mathbf{x}_{\text{cm}}^{f+}}{t_f \lambda_x(q_0^{s-}) \sigma_1^{s-}}; \quad (84)$$

the real-valued function τ varies between 0 and 1 and can be used to parameterize trajectories from $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$ to $\pi^{-1}(q_0^{s+}) \cap \mathcal{Z}_s$ in a neighborhood of the periodic orbit. Choose a function $\text{fcn}(a_1, \dots, a_5) : [0, 1] \rightarrow \mathbb{R}^4$ such that

$$\begin{aligned} \text{fcn}(a_1, \dots, a_5)(0) &= a_1 \\ \frac{d\text{fcn}}{d\tau}(a_1, \dots, a_5)(0) &= a_2 \\ \text{fcn}(a_1, \dots, a_5)(1) &= a_3 \\ \frac{d\text{fcn}}{d\tau}(a_1, \dots, a_5)(1) &= a_4, \end{aligned} \quad (85)$$

and there exist a_1^*, \dots, a_5^* for which $q_b - \text{fcn}(a_1^*, \dots, a_5^*)(\tau)$ (nearly) vanishes on \mathcal{O} . Here, this was accomplished with a fourth order polynomial. Off of the orbit, use (85) to solve for a_1, \dots, a_4 as functions of σ_1^{s-} so that $q_b(\tau) = \text{fcn}(a_1, \dots, a_5)(\tau)$ satisfies the constraints on the body coordinates imposed by (83). Specifically, set $a_1 = (q_0^{s-})_b$, $a_3 = (R^{-1}q_0^{s+})_b$, $a_2 = (\dot{q}_0^{s-} \sigma_1^{s-})_b$, and $a_4 = (\dot{q}_0(q_0^{s+}, \sigma_1^{s-}))_b$. Define

$$h_{d,f}(\mathbf{x}_{\text{cm}}, \sigma_1^{s-}, a_5) := \text{fcn}(a_1, \dots, a_5)(\tau) \quad (86)$$

with $a_i(\sigma_1^{s-})$, $i = 1, \dots, 4$ and $\tau(\mathbf{x}_{\text{cm}}, \sigma_1^{s-})$ as determined above. Define $q_5(0) = (q_0^{s-})_5$ and $q_{5,d} = (R^{-1}q_0^{s+})_5$.

In the final step, the goal is to select a_5 as a function of σ_1^{s-} so that the q_5 -component—the overall orientation of

the robot—satisfies the landing constraint. This is done as follows. The output (86) satisfies all of the conditions of Section VI, and hence the evolution of q_5 in the flight-phase zero dynamics is given by $\dot{q}_5 = \kappa_{1,f}(\sigma_{\text{cm}}, \mathbf{x}_{\text{cm}}, \dot{\mathbf{x}}_{\text{cm}}, \sigma_1^{s-}, a_5)$. In the flight phase, σ_{cm} and $\dot{\mathbf{x}}_{\text{cm}}$ are constant and can be substituted by their values from $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$. In addition, $\mathbf{x}_{\text{cm}}(t) = \mathbf{x}_{\text{cm}}^{s-} + t \lambda_x(q_0^{s-}) \sigma_1^{s-}$. Hence, $\dot{q}_5 = \tilde{\kappa}_{1,f}(t, \sigma_1^{s-}, a_5)$. Letting σ_1^{*s-} denote the value of σ_1^{s-} on the orbit, \mathcal{O} , $q_{5,d} = q_5(0) + \int_0^{t_f} \tilde{\kappa}_{1,f}(t, \sigma_1^{*s-}, a_5^*) dt$ is satisfied because, by construction of the output, the orbit corresponds to an integral curve of the flight-phase zero dynamics. Finally, it is verified (numerically) that

$$\frac{\partial}{\partial a_5} \left(q_{5,d} - q_5(0) - \int_0^{t_f} \tilde{\kappa}_{1,f}(t, \sigma_1^{*s-}, a_5) dt \right) \Big|_{a_5=a_5^*} \neq 0, \quad (87)$$

and thus by the implicit function theorem, there exists an open subset about σ_1^{*s-} and a differentiable function \tilde{w}_s^f such that $\tilde{w}_s^f(\sigma_1^{*s-}) = a_5^*$ and

$$q_{5,d} = q_5(0) + \int_0^{t_f} \tilde{\kappa}_{1,f}(t, \sigma_1^{s-}, \tilde{w}_s^f(\sigma_1^{s-})) dt. \quad (88)$$

Since (88) is scalar while a_5 has four components, there exist an infinite number of solutions for \tilde{w}_s^f . Hence, a numerical optimization was performed to find, for each point in a neighborhood of σ_1^{*s-} , a value of a_5 that steers q_5 to $q_{5,d}$, while minimizing⁹ $\|a_5 - a_5^*\|$. The flight-phase control design is completed by formally defining $h_{d,f}(q_f, a_f)$, $a_f := (\sigma_1^{s-}, a_5^*)'$, and $w_s^f(x_s^-) := (\sigma_1^{s-}, \tilde{w}_s^f(\sigma_1^{s-}))'$.

⁹Other criteria could be used, such as minimization of the torques in the flight phase. This latter criterion requires the computation of the torques via the dynamic model, and hence is costly in calculation time.

⁸Note that $\mathbf{x}_{\text{cm}}^{f+} = \mathbf{x}_{\text{cm}}^{s-}$.

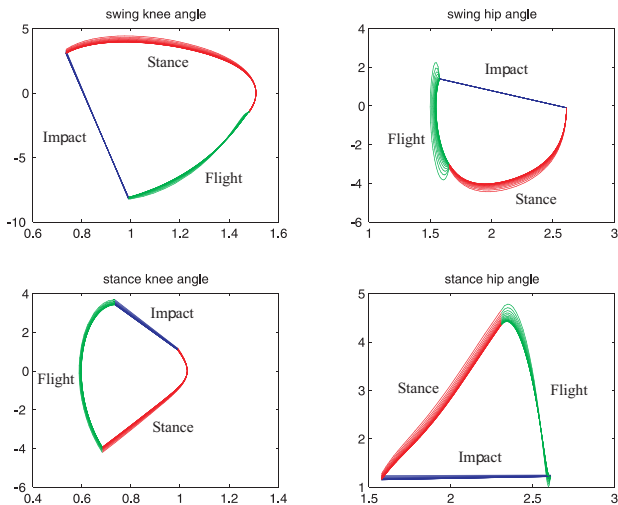


Fig. 10. Running at 1.5 m/s. The four graphs depict the relative joint angles in radians (x-axis) versus their velocities in radians per second (y-axis) in the stance, flight and impact phases: the swing knee angle (knee of leg-2), the swing hip angle (hip of leg-2), the stance knee angle (knee of leg-1) and the stance hip angle (hip of leg-1). At impact, the roles of the limbs are exchanged; as a consequence the configuration angles change at impact; see (30). Notice that the robot has the same configuration at each transition between phases. The plots indicate that a limit cycle is achieved.

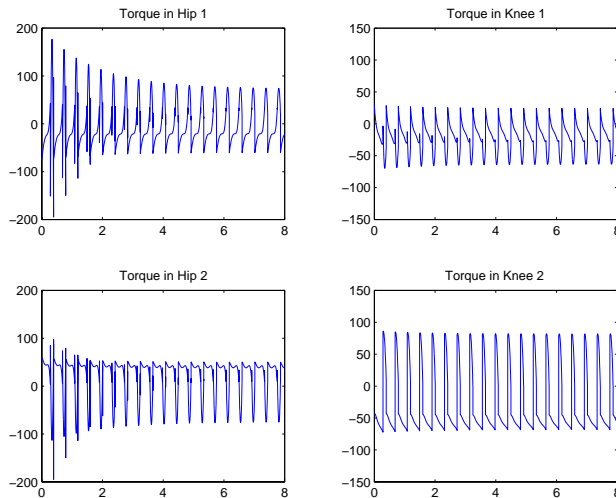


Fig. 12. Running at 1.5 m/s. The four graphs depict the joint torques in Newton-meters (y-axis) versus time in seconds (x-axis) in the stance and flight phases. Upon convergence to the periodic orbit, the achieved torques are very close to their optimal values. The torque is higher in the flight phase away from the periodic orbit, especially in the hips.

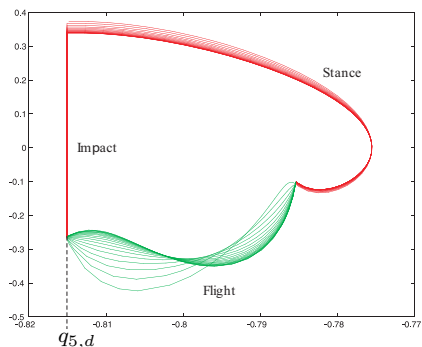


Fig. 11. Running at 1.5 m/s. The graph depicts torso angle in radians (x-axis) versus its velocity in radians per second (y-axis) in the stance and flight phases. Notice that the flight-phase controller has regulated the torso angle to its desired value of $q_{5,d}$ at impact. The plot indicates that a limit cycle is achieved.

D. Simulation without modeling error

The control law developed above has been simulated on a model of RABBIT for the various running motions. Assuming no modeling error and initializing the closed-loop system off of the periodic orbit—with the initial velocity 10% higher than the value on the periodic orbit—the simulation data presented in Figs. 10 to 17 are obtained for the running motions of 1.5 m/s and 2.5 m/s.

For a running speed of 1.5 m/s (resp., 2.5 m/s) Figs. 10 and 11 (resp., Figs. 14 and 15) show the phase-plane evolution of the configuration variables. The convergence to the periodic orbit is clear. By the design of the controller, the stance-phase evolution of the configuration variables does not change stride-to-stride; only the velocities change. In the flight phase, (most notably, for the hips and the torso when running at 1.5 m/s), the path traced out is modified so that the robot lands in the desired state.

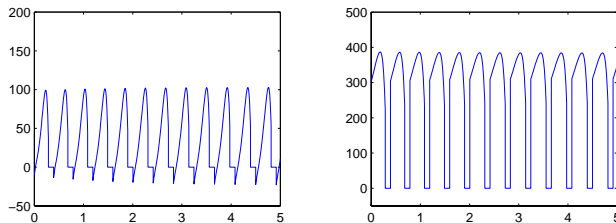


Fig. 13. Running at 1.5 m/s. The left graph depicts leg-1 (stance leg) horizontal force in Newtons (y-axis) versus time in seconds (x-axis) in the stance and flight phases. The right graph depicts vertical force (y-axis) versus time (x-axis) in the stance and flight phases. The impulsive forces existing during impact are not presented.

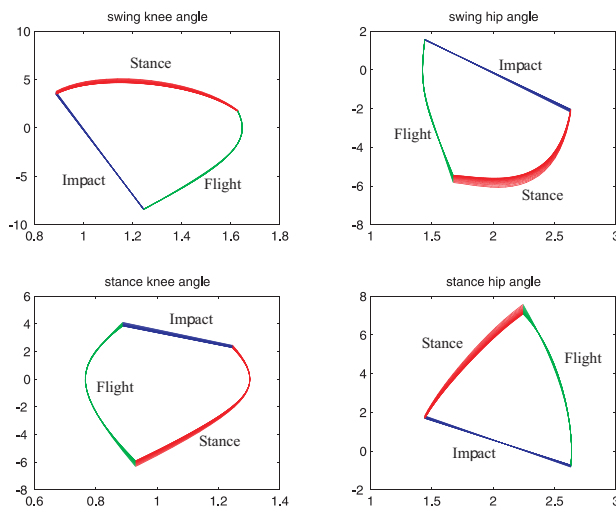


Fig. 14. Running at 2.5 m/s. The four graphs depict the relative joint angles in radians (x-axis) versus their velocities in radians per second (y-axis) in the stance, flight and impact phases: the swing knee angle (knee of leg-2), the swing hip angle (hip of leg-2), the stance knee angle (knee of leg-1) and the stance hip angle (hip of leg-1). At impact, the roles of the limbs are exchanged; as a consequence the configuration angles change at impact; see (30). Notice that the robot has the same configuration at each transition between phases. The plots indicate that a limit cycle is achieved.

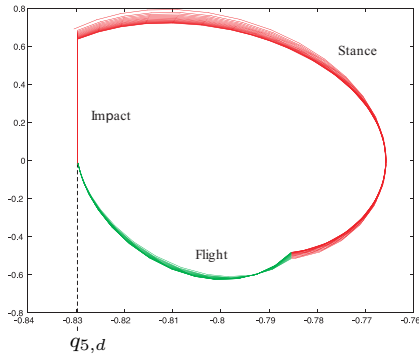


Fig. 15. Running at 2.5 m/s. The graph depicts torso angle in radians (x-axis) versus its velocity in radians per second (y-axis) in the stance and flight phases. Notice that the flight-phase controller has regulated the torso angle to its desired value of $q_{5,d}$ at impact. The plot indicates that a limit cycle is achieved.

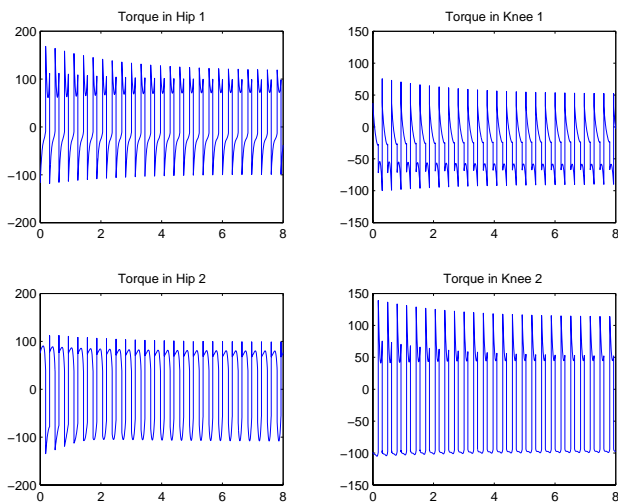


Fig. 16. Running at 2.5 m/s. The four graphs depict the joint torques in Newton-meters (y-axis) versus time in seconds (x-axis) in the stance and flight phases. Upon convergence to the periodic orbit, the achieved torques are very close to their optimal values. The torque is higher in the flight phase away from the periodic orbit, especially in the hips.

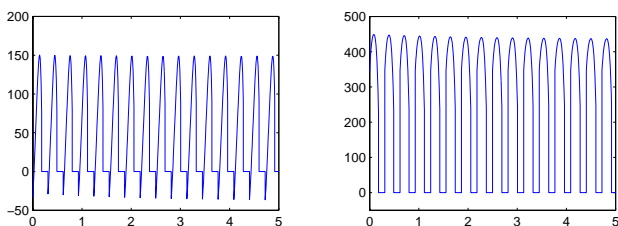


Fig. 17. Running at 2.5 m/s. The left graph depicts leg-1 (stance leg) horizontal force in Newtons (y-axis) versus time in seconds (x-axis) in the stance and flight phases. The right graph depicts vertical force (y-axis) versus time (x-axis) in the stance and flight phases. The impulsive forces existing during impact are not presented.

Figs. 12 and 16 depict the torques for running at 1.5 m/s and 2.5 m/s, respectively. As the motion converges to the periodic orbit, the torques correspond to their optimal values, and hence are within the capabilities of the actuators. Off of the periodic orbit, the torques are significantly higher in the flight phase. For the slower 1.5 m/s-orbit, the torque increase occurs principally in the hips. For the faster 2.5 m/s-orbit, the torque increase is more evenly divided among the four actuators and is smaller in magnitude; the corresponding modification to the path in the flight phase is also smaller; see Figs. 14 and 15.

The reaction forces on leg-1 are provided in Figs. 13 and 17. These graphs show the alternating phases of single support and flight. The robot will not slip for a coefficient of friction greater than 0.5. The vertical force during the single support phase is very close to the weight of the robot (from Tab. I, its mass is 37 kg).

IX. A PARTIAL ROBUSTNESS EVALUATION

The purpose of this section is to show that the proposed control strategy may still yield an attractive limit cycle even if the hypotheses made in the modeling of the robot, the control law's construction, and the analysis and simulation of the closed-loop system are not met exactly. The model of Section III assumed a rigid contact between the leg end and the ground. Here, a compliant contact model will be used [5]. This has several consequences. First of all, the seven DOF model of Section III-B will be used in the stance phase, with the position of the leg end with respect to the ground evolving freely as a function of the reaction forces provided by the compliant contact model. Secondly, the robot will enter the flight phase when the reaction forces at the leg end go to zero. Finally, the impact forces at touch down will be computed by the compliant model as well. In addition to these changes, parameter error will be introduced in the robot model.

A. Compliant contact model

In the experimental platform of RABBIT [1], the contact between the ends of the robot's legs and the ground is compliant and the ends of the legs may slip. A model that more closely reflects these points is summarized here. A more detailed discussion is available in [5] and the references therein.

The dynamic model consists of the full 7-DOF model of the biped (6) with the computation of the forces acting on the leg end being given by

$$\begin{aligned} F_n &= -\lambda^a |\mathbf{z}|^n \dot{\mathbf{z}} - \lambda^b |\mathbf{z}|^n \operatorname{sgn}(\dot{\mathbf{z}}) \sqrt{|\dot{\mathbf{z}}|} + k |\mathbf{z}|^n \\ F_t &= (\vartheta^a d + \vartheta^b \dot{d} + \vartheta^c \mathbf{v} + \vartheta^d \operatorname{sgn}(\dot{\mathbf{v}}) \sqrt{|\mathbf{v}|}) |F_n| \quad (89) \\ \dot{d} &= \mathbf{v} - |\mathbf{v}| \frac{\vartheta^a}{\vartheta^e} d, \end{aligned}$$

where $\mathbf{z} \leq 0$ is the penetration depth (if $\mathbf{z} \leq 0$, the leg is in contact with the ground, if $\mathbf{z} > 0$, the leg is not in contact with the ground and the contact forces equal zero) and \mathbf{v} is the relative velocity of the end of the leg with respect to the ground. This model supposes that the interface between the two contacting surfaces is a contact between bristles; the

Parameter	Value	Parameter	Value
λ^a	9×10^6	ϑ^a	260
λ^b	0.3	ϑ^b	0.6
ϑ^c	0.18	n	1.5
ϑ^d	0.3	k	25×10^5
		ϑ^e	0.285

TABLE III
COMPLIANT CONTACT MODEL PARAMETERS.

average deflection d of the bristles is an internal state used to model dynamic friction. The numerical values used in the simulation, given in Tab. III, were adjusted for a nominal penetration of approximately 3 mm and to avoid rebound of the leg during the stance phase.

Together, the models (6) and (89) describe the robot's evolution in all phases of motion: flight, stance and impact. The robot's dynamics are then described by ordinary (non-hybrid) differential equations over the entire stride, even during the impact, which will now have a non-zero duration. With this model, contact forces at the leg end are continuous, which means in particular that they will not experience an instantaneous jump to zero at the transition from stance to flight as supposed in the development of the control law.

B. Simulation with modeling error

In addition to the structural change in the contact model, parametric modeling error is included. A deviation of $\pm 20\%$ in the masses and inertias was introduced between the robot's design model and the simulation model; symmetry of the two legs was preserved. It is important to note that one consequence of parametric error is that there will be an error in the state of the robot at landing: because the flight-phase controller does not correspond to the simulation model, it will not correctly account of the conservation of angular momentum. Finally, saturation was introduced on the torques (± 150 Nm) to take into account the limitations of the actuators of RABBIT.

Despite all of the differences between the simulation model and the model used to design the controller, the feedback controller illustrated in Section VIII is able to induce a stable running motion. This is shown in Figs. 18 to 21 for a nominal speed of 1.5 m/s. In the simulations, the controller was switched from the stance phase to flight phase when $\theta_s(q)$ attained θ_s^- , and it was switched from flight phase to stance phase when the penetration of the leg end into the compliant surface exceeded 2 mm. Due to the differences in the design and simulation models, the limit cycle does not correspond exactly to the theoretical prediction. The value of ζ^* calculated from the simulation data and the model parameters is 829, whereas the value predicted with the rigid model and perfectly known parameters was 801 (see Tab. II). The average running speed was calculated to be 1.54 m/s, compared to the previous value of 1.50 m/s.

Figs. 18 and 19 show the evolution of the configuration variables in the phase plane; the convergence to a limit cycle

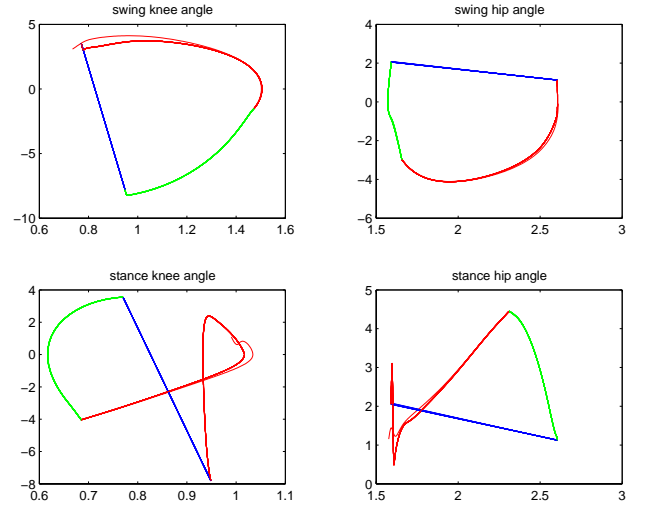


Fig. 18. Running at 1.5 m/s with the compliant contact model and parametric modeling error. The four graphs depict the relative joint angles in radians (x-axis) versus their velocities in radians per second (y-axis) in the stance, flight and impact phases: the swing knee angle (knee of leg-2), the swing hip angle (hip of leg-2), the stance knee angle (knee of leg-1) and the stance hip angle (hip of leg-1). At impact, the roles of the limbs are exchanged. Notice the abrupt change in the velocities at impact, especially in the stance leg. The plots indicate that a limit cycle is achieved.

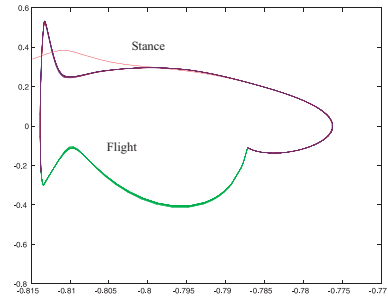


Fig. 19. Running at 1.5 m/s with the compliant contact model and parametric modeling error. The graph depicts torso angle in radians (x-axis) versus its velocity in radians per second (y-axis) in the stance and flight phases. Notice that the flight-phase controller has approximately regulated the torso angle to its desired value of $q_{5,d}$ at impact. The plot indicates that a limit cycle is achieved.

is clear. At touchdown, the roles of the legs are swapped, as when the rigid contact model was used. At the beginning of the stance phase, the impact causes an abrupt change in the robot's velocities. At the moment of contact, the robot's velocities still correspond to their values from the flight phase. The control law sees this as a large set-point error and consequently applies a large torque, resulting in saturation; see Fig. 20. Once past the impact, the evolution of the relative angles is quite close to what was predicted with the rigid impact model; see Fig. 10 and Fig. 11. The perturbations during the flight phase are small because the initial condition of the simulation lies on the periodic orbit corresponding to the rigid contact model and no parametric modeling error.

The reaction forces on leg-1 are provided in Fig. 21. These graphs show the alternating phases of single support and flight. Except during impact, which is no longer instantaneous, the forces are close to the values predicted by the earlier

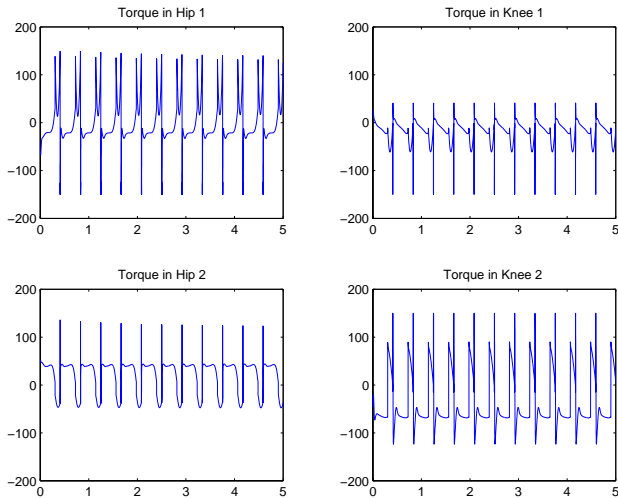


Fig. 20. Running at 1.5 m/s with the compliant contact model and parametric modeling error. The four graphs depict the joint torques in Newton-meters (y-axis) versus time in seconds (x-axis) in the stance and flight phases. The torques are limited to ± 150 Nm. Upon convergence to the periodic orbit, the achieved torques are close to their optimal values. Prior to convergence, note the larger torques in the beginning of the stance phase due to a combination of modeling error and landing in the wrong state.

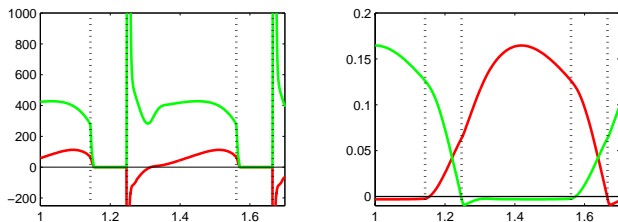


Fig. 21. Running at 1.5 m/s with the compliant contact model and parametric modeling error. The left graph depicts leg-1 (stance leg) horizontal and vertical force components in Newtons (y-axis) versus time in seconds (x-axis) in the stance and flight phases. Large forces occur at touchdown; the maximal vertical force is close to 8000 N and the maximal horizontal force is close to -4000 N with the compliant contact model. The vertical lines show the instant of transition between the control law phases. The right graph depicts vertical position of the leg end in meters (y-axis) versus time in seconds (x-axis) in the stance and flight phases. Notice that the flight control law induces the stance leg to lift off quickly and the reaction forces to go to zero.

simulation; see Fig. 13. The penetration of the stance leg end stabilizes at approximately 3 mm. These two plots show clearly the very rapid liftoff of the stance leg to initiate the flight phase. Consequently, for the purposes of modeling, feedback design, and analysis, it is as reasonable to suppose an instantaneous transition to the flight phase as it is to suppose an instantaneous impact.

X. ADDITIONAL EVENT-BASED CONTROL FOR RUNNING

Each of the feedback designs illustrated in Section VIII resulted in a nominally exponentially stable running motion. Indeed, this has been the case for all of the periodic orbits computed by the authors using the techniques in [11]. From Tab. II, it is seen that the rate of convergence to the periodic orbit decreases as the average running speed increases (that is, μ becomes closer to 1.0). The aim of this section is to illustrate how an additional event-based-control action studied

in [4], [8], [13] can be profitably used to increase the rate of convergence to the periodic orbit. It will also be shown that the additional feedback action can be used to reduce the magnitude of the torques that are used in the flight phase to attain the desired landing state.

Remark 6: In Section IX-B, it was seen that modeling error alters the average running speed. As in [8], event-based control could also be used to attenuate the effects of modeling error on average running speed. In addition, it could be used to stabilize a periodic orbit that was nominally unstable under the feedback designs proposed so far.

A. Deciding what to control

Based on [15], it is natural to conjecture that modification of the target landing configuration stride-to-stride can be used to ameliorate the rate of convergence to the orbit and the peak torques in the flight phase. In particular, the horizontal distance between the center of mass and the stance leg has a strong effect¹⁰ on μ . This suggests modifying the landing configuration in the direction $[0, 0, 1, 0, 0]'$. On the other hand, the action of modifying the flight trajectory to obtain the correct orientation of the torso at landing is what leads to the higher torques. This suggests modifying the landing configuration in the direction $[0, 0, 0, 0, 1]'$.

B. Implementing stride-to-stride updates of landing configuration

Let q_0^{f-} denote the nominal landing configuration for one of the running motions of Section VIII; see (83). Set the desired landing configuration at the k th stride to be

$$q_{0,d}^{f-}(k) = q_0^{f-} + [0, 0, w_1(k), 0, w_2(k)]', \quad (90)$$

where the real scalars $w_1(k)$ and $w_2(k)$ are to be updated at the end of each stance phase. Through the impact map (30), a change in the desired landing configuration needs to be accompanied by a corresponding change in the desired initial stance configuration. Both of these changes entail stride-to-stride parameter updates to the stance and flight controllers of Section VI. As a result, the restricted Poincaré map is now a function of $w_1(k)$ and $w_2(k)$ and can be viewed as a discrete-time control system

$$\zeta(k+1) = \rho(\zeta(k), w_1(k), w_2(k)) \quad (91)$$

with state space $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$ and inputs $(w_1, w_2)' \in \mathbb{R}^2$; see [13] for details. Linearizing (91) about the nominal fixed-point ζ^* corresponding to $w_1 = 0$ and $w_2 = 0$ results in

$$\delta\zeta(k+1) = \mu\delta\zeta(k) + b_1\delta w_1(k) + b_2\delta w_2(k). \quad (92)$$

The value of μ is determined from Corollary 1; the sensitivities b_1 and b_2 are more easily determined numerically through a simulation of the model.

Linear state variable feedback $\delta w_1(k) = k_1\delta\zeta(k)$, $\delta w_2(k) = k_2\delta\zeta(k)$ can then be used to tradeoff peak torques

¹⁰When the heights of the center of mass at the beginning and end of the stance phase are the same, $\mu = (\chi - |\beta|)$, which is a function only of the horizontal position of the center of mass with respect to the stance leg end; see (66).

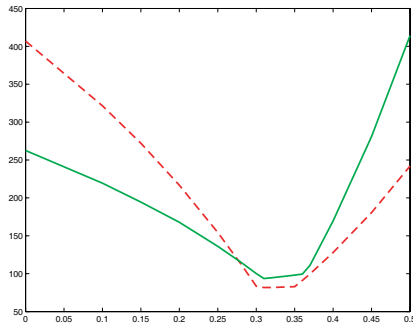


Fig. 22. A one-parameter search to minimize peak torque. Let $k_2 = ak_1$. The graph depicts the maximal torque in Newton-meters (y -axis) versus the parameter a (x -axis) for an initial velocity of the robot equal to $\pm 10\%$ of its value on the periodic orbit (the solid line corresponds to $+10\%$ and the dashed line corresponds to -10%). The best choice of parameter a is $0.3 < a < 0.35$ to minimize the peak torque.

and the rate of convergence to the fixed point. For the running motion with average speed of 1.5 m/s, it was arbitrarily decided to place the closed-loop eigenvalue at $\mu_d = 2/3$. A one-parameter search was then performed to minimize the torques in the flight phase when the velocity upon entering the flight phase differed from the value on the periodic orbit by $\pm 10\%$, subject to $\mu + k_1b_1 + k_2b_2 = 2/3$; see Fig. 22. This resulted in $k_1 = 7.8 \times 10^{-5}$ and $k_2 = 2.6 \times 10^{-5}$. It is important to note that transient performance has been optimized subject to a stability constraint.

C. Simulation results

Assuming no modeling error and initializing the closed-loop system off of the periodic orbit—with the initial velocity 10% higher than its value on the periodic orbit—yields the simulation data presented in Figs. 23 through 25. The landing configuration is being modified at each stride. The orientation of the support hip and the torso vary slightly stride-to-stride under the event-based feedback. The deviation in the flight phase trajectory—compare Figs. 23 and 24 to Figs. 10 and 11—is clearly much less under the event-based control action. Consequently, the torques during the flight phase are noticeably reduced; see Fig. 25.

The evolution of ζ from stride-to-stride over the course of the simulation is presented in Fig. 26. The desired convergence rate has been achieved.

The evolution of the event-based-control action, $w_1 + w_2$, is presented in Fig. 27. The induced variation in the landing configuration is rather small. Despite this, there are significant improvements in the rate of convergence to the periodic orbit and the reduction in peak torque. This sensitivity points to potential problems when implementation is pursued on the actual mechanism. It may be necessary to develop other methods to avoid unnecessarily high torques in the flight phase.

XI. CONCLUSIONS

A time-invariant feedback control strategy has been developed for a bipedal runner. The control strategy is hybrid: it

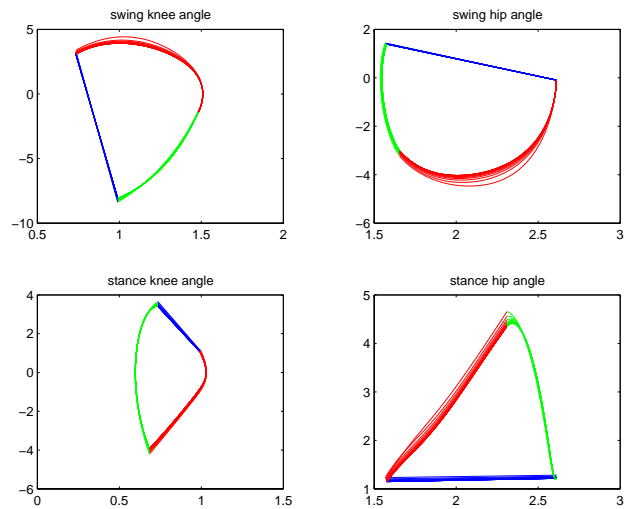


Fig. 23. Running at 1.5 m/s with event-based control of the landing configuration. The four graphs depict the relative joint angles in radians (x -axis) versus their velocities in radians per second (y -axis) in the stance, flight and impact phases: the swing knee angle (knee of leg-2), the swing hip angle (hip of leg-2), the stance knee angle (knee of leg-1) and the stance hip angle (hip of leg-1). At impact, the roles of the limbs are exchanged; as a consequence the configuration angles change at impact; see (30). Notice that the robot no longer has the same configuration at each transition between phases. The plots indicate that a limit cycle is achieved.

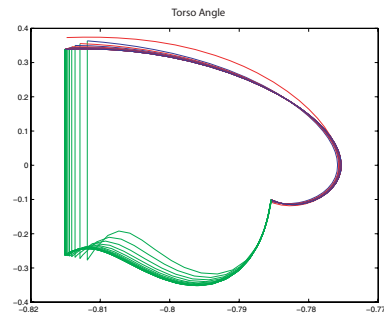


Fig. 24. Running at 1.5 m/s with event-based control of the landing configuration. The graph depicts torso angle in radians (x -axis) versus its velocity in radians per second (y -axis) in the stance and flight phases. Notice that the torso angle at the end of the flight phase varies stride-to-stride. The plot indicates that a limit cycle is achieved.

acts both continuously during a stride and discretely stride-to-stride. The flight-phase portion of the control strategy was designed so as to create a generalized impact map whose properties are similar to those of the impact maps that occur in models of walking. This led to the deliberate design of a hybrid zero dynamics of running, that is, a low-dimensional, invariant, sub-dynamic of the closed-loop hybrid system. Asymptotically stable orbits of the hybrid zero dynamics are asymptotically stabilizable orbits of the full-order hybrid model. Using the idea of a restricted Poincaré return map—which is the Poincaré return map associated with the hybrid zero dynamics—an explicit criterion for the existence of periodic orbits was given, as well as an explicit characterization of their stability properties. With this theoretical tool, stability analysis involves the straightforward computation of a scalar map.

The principal results were illustrated on a five-link, four-actuator planar biped with revolute joints. Periodic trajectories

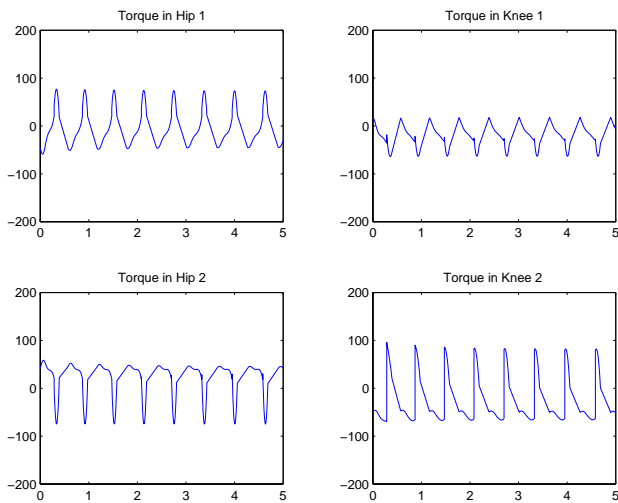


Fig. 25. Running at 1.5 m/s with event-based control of the landing configuration. The four graphs depict the joint torques in Newton-meters (y-axis) versus time in seconds (x-axis) in the stance and flight phases. Modifying the landing configuration stride-to-stride has resulted in much smaller torques when the robot is off of the periodic orbit.

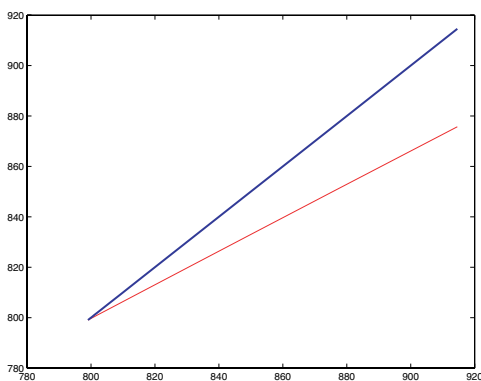


Fig. 26. Running at 1.5 m/s with event-based control of the landing configuration. The graph's thick line depicts the value of ζ at step $k+1$ (y-axis) versus its value at step k (x-axis) as obtained directly from the simulation. The desired modification in the slope of the Poincaré map has been obtained without changing the fixed point: slope ≈ 0.66 , $\zeta^* \approx 800$. The thin line is the identity map. The fixed point is at the intersection of the two lines.

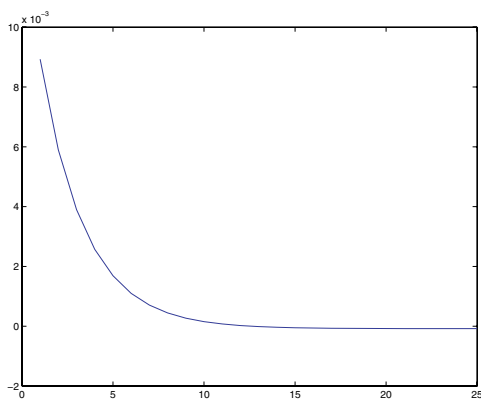


Fig. 27. Running at 1.5 m/s with event-based control of the landing configuration. The graph depicts w_1 (y-axis) versus step number (x-axis) as obtained in the simulation. Note that $w_2 = \frac{k_2}{k_1} w_1$.

computed in [11] were interpreted as desired periodic orbits in the state space of the robot model. For each trajectory, a time-invariant feedback controller was designed to realize the corresponding orbit as an attractive solution of the hybrid zero dynamics, without, it is very important to note, the use of trajectory tracking. Very roughly speaking, the controller is “clocked” to events on the orbit and not to time. Hence, when perturbed away from the orbit, the robot’s links regain “synchrony” with respect to the robot’s position on the orbit and not with respect to time. In this sense, the work here is philosophically similar to [15]–[18], [55] and diametrically opposed to most other work in the legged-locomotion literature.

The performance of the controller was simulated on slow and fast running motions. Robustness of the controller to model imperfections was demonstrated through simulations where the rigid impact model was replaced by a compliant contact model, and where the parameters of the robot model did not correspond to those used to design the controller. The event-based control method of [4], [13] was used to accelerate the rate of convergence to a periodic orbit and to improve the transient performance in the flight phase.

This work has only partially explored the benefits of approaching the feedback control of running with tools previously developed for walking. On the basis of [4], it seems likely that event-based control can be profitably used in running to provide additional regulation of average running speed and rejection of perturbations. These extensions may involve event-based control actions in the stance-phase controller, parallel to what was done here for the flight-phase controller.

While this paper was in review, experimental implementation of a running controller was initiated [56].

ACKNOWLEDGMENTS

The work of E.R. Westervelt and J.W. Grizzle was supported by NSF grants CMS-0408348 and ECS-0322395, respectively.

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APPENDIX I

STANCE AND FLIGHT PHASE CONTROL LAWS

The control laws are rewritten here in a form where their dependence on terms in the models becomes more explicit.

A. Stance phase control

The control law is defined so as to achieve input-output linearization with respect to v , $\frac{d^2 y_s}{dt^2} = v(y_s, \dot{y}_s)$, where $v(y_s, \dot{y}_s)$ is subsequently chosen to globally asymptotically stabilize the origin with finite-time convergence [2]. Input-output linearization is equivalent to

$$\ddot{q}_b = \frac{\partial h_{d,s}}{\partial \theta_s} \ddot{\theta}_s + \frac{\partial^2 h_{d,s}}{\partial \theta_s^2} \dot{\theta}_s^2 + v. \quad (93)$$

The robot's configuration is a linear combination of q_b and θ_s ,

$$q = \mathbf{T} \begin{bmatrix} q_b \\ \theta_s \end{bmatrix}, \quad (94)$$

where \mathbf{T} is a constant invertible matrix, and hence the control law is easily expressed directly in terms of the dynamic model. Indeed, the required torques satisfy

$$D_s(q_b) \mathbf{T} \begin{bmatrix} \frac{\partial h_{d,s}}{\partial \theta_s} \ddot{\theta}_s + \frac{\partial^2 h_{d,s}}{\partial \theta_s^2} \dot{\theta}_s^2 + v \\ \ddot{\theta}_s \end{bmatrix} + C_s(q_b, \dot{q}) \dot{q} + G_s(q) = \begin{bmatrix} I_{4 \times 4} \\ 0 \end{bmatrix} u. \quad (95)$$

The fifth row of the model yields $\ddot{\theta}_s$. Substituting this expression in the above yields the required feedback controller

$$u_s = -D_{s,1:4}(q_b) \mathbf{T} \begin{bmatrix} \frac{\partial h_{d,s}}{\partial \theta_s} \\ 1 \end{bmatrix} \left(D_{s,5}(q_b) \mathbf{T} \begin{bmatrix} \frac{\partial h_{d,s}}{\partial \theta_s} \\ 1 \end{bmatrix} \right)^{-1} \left(D_{s,5}(q_b) \mathbf{T} \begin{bmatrix} \frac{\partial^2 h_{d,s}}{\partial \theta_s^2} \dot{\theta}_s^2 + v \\ 0 \end{bmatrix} + C_{s,5}(q_b, \dot{q}) \dot{q} + G_{s,5}(q) \right) + D_{s,1:4}(q_b) \mathbf{T} \begin{bmatrix} \frac{\partial^2 h_{d,s}}{\partial \theta_s^2} \dot{\theta}_s^2 + v \\ 0 \end{bmatrix} + C_{s,1:4}(q_b, \dot{q}) \dot{q} + G_{s,1:4}(q), \quad (96)$$

where the subscript “1 : 4” designates the sub-matrix composed of the first four rows and the subscript “5” designates the fifth row only.

B. Flight phase control

The control law is defined so as to achieve input-output linearization, $\frac{d^2 y_f}{dt^2} = v(y_f, \dot{y}_f)$, where $v(y_f, \dot{y}_f) = -K_p y_f - K_d \dot{y}_f$ exponentially stabilizes the double integrator. Input-output linearization is equivalent to

$$\ddot{q}_b = \frac{\partial^2 h_{d,f}(x_{cm}, a_f)}{\partial x_{cm}^2} \ddot{x}_{cm} + v, \quad (97)$$

because a_f is constant during the flight phase and $\ddot{x}_{cm} = 0$. The control law is easily determined using the first five rows of the dynamic model:

$$u_f = (A_{1:4,1:4} - A_{1:4,5} A_{5,5}^{-1} A_{5,1:4}) \left(\frac{\partial^2 h_{d,f}}{\partial x_{cm}^2} \ddot{x}_{cm} + v \right) + C_{f,1:4} - A_{1:4,5} A_{5,5}^{-1} C_{f,5}, \quad (98)$$

where the subscript “1 : 4, 1 : 4” designates the square sub-matrix composed of the first four rows and columns, the

subscript “1 : 4, 5” denotes the first four rows of the fifth column, the subscript “5, 1 : 4” denotes the first four columns of the fifth row and the subscript “5” designates the fifth row only.

APPENDIX II

HYPOTHESES ON THE FINITE-TIME CONTROLLER IN THE STANCE PHASE

Let $v(y, \dot{y})$ be any feedback controller on (36) satisfying conditions CH2–CH5 of [2], that is, for $N = 4$ the dimension of y :

Controller Hypotheses: for the closed-loop chain of double integrators, $\ddot{y} = v(y, \dot{y})$,

CH2) solutions globally exist on \mathbb{R}^{2N-2} , and are unique;

CH3) solutions depend continuously on the initial conditions;

CH4) the origin is globally asymptotically stable, and convergence is achieved in finite time;

CH5) the settling time function¹¹, $T_{\text{set}} : \mathbb{R}^{2N-2} \rightarrow \mathbb{R}$ by

$$T_{\text{set}}(y_0, \dot{y}_0) := \inf \{ t > 0 \mid (y(t), \dot{y}(t)) = (0, 0), (y(0), \dot{y}(0)) = (y_0, \dot{y}_0) \}$$

depends continuously on the initial condition, (y_0, \dot{y}_0) .

Hypotheses CH2–CH4 correspond to the definition of finite-time stability [57], [58]; CH5 is also needed, but is not implied by CH2–CH4 [59]. These requirements rule out traditional sliding mode control, with its well-known discontinuous action.

APPENDIX III

PROOF OF THEOREM 2

By (41), points in $\tilde{\mathcal{S}} \cap \mathcal{Z}_s$ are parameterized by $(q_0^{s-}, \dot{q}_0^{s-}, \sigma_1^{s-})$. The position of the center of mass $(x_{cm}^{s-}, y_{cm}^{s-})$ is obtained by evaluating (8) at q_0^{s-} and its velocity is obtained from (46), $(\dot{x}_{cm}^{s-}, \dot{y}_{cm}^{s-}) = (\lambda_x(q_0^{s-})\sigma_1^{s-}, \lambda_y(q_0^{s-})\sigma_1^{s-})$. The angular momentum about the center of mass can be determined from (24) to be

$$\sigma_{cm}^{s-} = \sigma_1^{s-} - m (y_{cm}^{s-} \lambda_x(q_0^{s-})\sigma_1^{s-} - x_{cm}^{s-} \lambda_y(q_0^{s-})\sigma_1^{s-}). \quad (99)$$

Since the transition map from the stance phase to the flight phase preserves positions and velocities, (99) is also the angular momentum at the *beginning* of the flight phase, σ_{cm}^{f+} , and because angular momentum is conserved during ballistic motion, (99) is also the value of the angular momentum at the *end* of the flight phase, σ_{cm}^{f-} . From the hypotheses $\Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s) \subset \mathcal{Z}_s$ and $\pi \circ \Delta(\tilde{\mathcal{S}} \cap \mathcal{Z}_s)$ is a single point, the position of the center of mass at the end of the flight phase is known and equal to the position of the center of mass at the beginning of the subsequent stance phase, $(x_{cm}^{s+}, y_{cm}^{s+})$. From this, the flight time, t_f , can be computed

$$t_f = \frac{\dot{y}_{cm}^{s-}}{g} + \frac{\sqrt{(\dot{y}_{cm}^{s-})^2 - 2g(y_{cm}^{s+} - y_{cm}^{s-})}}{g}, \quad (100)$$

¹¹That is, the time it takes for a solution initialized at (y_0, \dot{y}_0) to converge to the origin. The terminology is taken from [57].

and from (23), the velocity of the center of mass at the end of the flight phase is determined

$$\begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}(t_f) \\ \dot{\mathbf{y}}_{\text{cm}}(t_f) \end{bmatrix} = \begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}^{s-} \\ -\sqrt{(\dot{\mathbf{y}}_{\text{cm}}^{s-})^2 - 2g(\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-})} \end{bmatrix}. \quad (101)$$

Equations (99), (101), and (24) allow the angular momentum about the contact point at the end of the flight phase, σ_2^{f-} , to be evaluated, and then (28) allows the evaluation of the angular momentum about the stance leg at the beginning of the subsequent stance phase. This yields

$$\begin{aligned} \sigma_1^{s+} = & \sigma_1^{s-} - m(\mathbf{y}_{\text{cm}}^{s-} \lambda_x(q_0^{s-}) \sigma_1^{s-} - \mathbf{x}_{\text{cm}}^{s-} \lambda_y(q_0^{s-}) \sigma_1^{s-}) + \\ & m \left(\mathbf{y}_{\text{cm}}^{s+} \lambda_x(q_0^{s-}) \sigma_1^{s-} + \right. \\ & \left. \mathbf{x}_{\text{cm}}^{s+} \sqrt{(\lambda_y(q_0^{s-}) \sigma_1^{s-})^2 - 2g(\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-})} \right), \quad (102) \end{aligned}$$

which, after simplification, completes the proof.

APPENDIX IV PROOF OF THEOREM 3

From [3, Sec. IV], in the coordinates $(\theta_s, K_{s,\text{zero}} = \frac{1}{2}(\sigma_1)^2)$ for Z_s , the stance-phase zero dynamics can be integrated as

$$K_{s,\text{zero}}(\theta_s) = K_{s,\text{zero}}(\theta_s^+) - V_{s,\text{zero}}(\theta_s). \quad (103)$$

Evaluating the above at θ_s^- and applying (67) yields the restricted Poincaré map

$$\rho(\zeta) = \delta_e(\zeta) - V_{s,\text{zero}}(\theta_s^-), \quad (104)$$

where $\zeta = \frac{1}{2}(\sigma_1^{s-})^2$. The domain of ρ follows from [3, Thm. 3].

Remark 7: The integration of the stance phase zero dynamics can also be expressed as

$$K_{s,\text{zero}}(\theta_s) + V_{s,\text{zero}}(\theta_s) = K_{s,\text{zero}}(\theta_s^+),$$

for $\theta_s^+ \leq \theta_s < \theta_s^-$, which is conservation of total ‘‘pseudo-energy’’ during the stance phase; see also Fig. 5.

APPENDIX V PROOF OF THEOREM 4

By (d) of Corollary 2, the hypotheses imply that ρ is strictly convex, and by (c), $\frac{d\rho}{d\zeta} \leq (\chi - |\beta|)^2 < 1$. Hence, the graph of ρ can have at most one intersection with the graph of the identity function, which implies that there can exist at most one fixed point. Since δ_e and ρ differ by a constant, their derivatives are equal and Corollary 2 applies equally to δ_e . Therefore, δ_e is strictly increasing on \tilde{D}_ρ , and therefore, if $\tilde{\zeta} \in \tilde{D}_\rho$, then $\zeta \in \tilde{D}_\rho$ for all $\zeta > \tilde{\zeta}$. It follows that \tilde{D}_ρ is unbounded and connected. By Corollary 1, ζ^* is exponentially stable.

Let $\zeta \in \tilde{D}_\rho$ be such that $\zeta < \zeta^*$. Then, since ρ is strictly increasing on \tilde{D}_ρ , $\rho(\zeta) < \rho(\zeta^*) = \zeta^*$. Hence, $\rho^{(k)}(\zeta)$ is a strictly increasing sequence bounded from above, and therefore has a limit. By continuity of ρ , this limit is a fixed point of ρ , and by uniqueness of the fixed point,

$\lim_{k \rightarrow \infty} \rho^{(k)}(\zeta) = \zeta^*$. Similarly, let $\zeta \in \tilde{D}_\rho$ be such that $\zeta > \zeta^*$. Then $\zeta^* = \rho(\zeta^*) < \rho(\zeta)$, and similar reasoning shows that $\lim_{k \rightarrow \infty} \rho^{(k)}(\zeta) = \zeta^*$, with the convergence being monotonic.

APPENDIX VI EFFECTS OF MEASURING ANGLES IN COUNTER-CLOCKWISE DIRECTION

If angles are positive when measured in the counter-clockwise direction, the following equations are modified. Equation (24) becomes

$$\sigma_i = \sigma_{\text{cm}} + m((\mathbf{x}_{\text{cm}} - \mathbf{x}_i)\dot{\mathbf{y}}_{\text{cm}} - (\mathbf{y}_{\text{cm}} - \mathbf{y}_i)\dot{\mathbf{x}}_{\text{cm}}). \quad (105)$$

The fifth row of (18) is

$$\sigma_{\text{cm}}^+ - \sigma_{\text{cm}}^- = -m[\mathbf{y}_{\text{cm}} - \mathbf{y}_2 | -\mathbf{x}_{\text{cm}} + \mathbf{x}_2] \begin{bmatrix} \dot{\mathbf{x}}_{\text{cm}}^+ - \dot{\mathbf{x}}_{\text{cm}}^- \\ \dot{\mathbf{y}}_{\text{cm}}^+ - \dot{\mathbf{y}}_{\text{cm}}^- \end{bmatrix}, \quad (106)$$

since $\frac{\partial f_2}{\partial q_5} = [\mathbf{y}_{\text{cm}}(q) - \mathbf{y}_2(q) | -\mathbf{x}_{\text{cm}}(q) + \mathbf{x}_2(q)]'$, and using (105) results in (27). Consequently, Appendix III is modified as follows. The angular momentum about the center of mass can be determined from (105) to be

$$\sigma_{\text{cm}}^{s-} = \sigma_1^{s-} + m(\mathbf{y}_{\text{cm}}^{s-} \lambda_x(q_0^{s-}) \sigma_1^{s-} - \mathbf{x}_{\text{cm}}^{s-} \lambda_y(q_0^{s-}) \sigma_1^{s-}). \quad (107)$$

This yields

$$\begin{aligned} \sigma_1^{s+} = & \sigma_1^{s-} + m(\mathbf{y}_{\text{cm}}^{s-} \lambda_x(q_0^{s-}) \sigma_1^{s-} - \mathbf{x}_{\text{cm}}^{s-} \lambda_y(q_0^{s-}) \sigma_1^{s-}) - \\ & m \left(\mathbf{y}_{\text{cm}}^{s+} \lambda_x(q_0^{s-}) \sigma_1^{s-} + \right. \\ & \left. \mathbf{x}_{\text{cm}}^{s+} \sqrt{(\lambda_y(q_0^{s-}) \sigma_1^{s-})^2 - 2g(\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-})} \right). \quad (108) \end{aligned}$$

Thus

$$\delta(\sigma_1^{s-}) = \chi \sigma_1^{s-} + \sqrt{(\beta \sigma_1^{s-})^2 + \alpha}, \quad (109)$$

with

$$\begin{aligned} \alpha &= -2m^2g(\mathbf{x}_{\text{cm}}^{s+})^2(\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-}) \\ \beta &= m \mathbf{x}_{\text{cm}}^{s+} \lambda_y(q_0^{s-}) \\ \chi &= 1 - m \mathbf{x}_{\text{cm}}^{s-} \lambda_y(q_0^{s-}) - m(\mathbf{y}_{\text{cm}}^{s+} - \mathbf{y}_{\text{cm}}^{s-}) \lambda_x(q_0^{s-}). \end{aligned} \quad (110)$$

and equations (65) and (66) must be changed accordingly. Since the sign of the angular momentum is different, the signs of λ_x and λ_y are changed. The numerical evaluation of α and χ gives the same value whether the positive direction is defined clockwise or counter-clockwise. The sign of β is changed, but it does not play any role. The main difference is the sign before the square root; thus the evolution of the absolute value of the angular momentum given by (65) for positive angular momentum and by (109) for negative angular momentum is the same. The remainder of the analysis is based on the evolution of $\zeta = \frac{1}{2}(\sigma_1^{s-})^2$, and since the sign of the angular momentum does not affect ζ , the analysis and formulas are valid for any choice of positive direction.

APPENDIX VII
NOMENCLATURE

The subscript “f” corresponds to the flight phase and the subscript “s” corresponds to the stance phase. The superscripts “+” and “-” define the beginning and end of a phase respectively. When they are applied to a variable that is only defined for a single phase, such as θ_s , then there is no ambiguity, as in θ_s^{s+} . For a variable such as \mathbf{x}_{cm} , which is used in both flight and stance phases, the notation \mathbf{x}_{cm}^{s+} , \mathbf{x}_{cm}^{s-} , \mathbf{x}_{cm}^{f+} , and \mathbf{x}_{cm}^{f-} is used. The superscript “*” occasionally is used to denote the value of a variable on a periodic orbit; an exception is u^* , which is a feedback control rendering a zero dynamics manifold invariant.

Symbol	Meaning	Defined
a_f	parameter in flight-phase virtual constraints; is up-dated each step	Sec. VI-B
$A(q_b)$	part of inertia matrix in flight phase, $A_i : i - th$ row of A , $A_{ij} = A(i, j)$	Eq. (3)
\mathcal{A}	space where a_f takes values $\mathcal{A} := \mathbb{R}^p$, p number of parameters	Sec. VI-B
α	parameter defining the function δ	Eq. (66)
b_1, b_2	sensitivities of restricted Poincaré map w.r.t. variations w_1 and w_2	Sec. X
B_f, B_s	transforms u into Γ_f or Γ_s	Eqs. (5), (13)
β	parameter defining the function δ	Eq. (66)
C_f, C_s	coriolis and centrifugal terms	Secs. III-B, III-C
χ	parameter defining the function δ	Eq. (66)
D_f, D_s	inertia matrices	Secs. III-B, III-C
\mathcal{D}_ρ	domain of definition of ρ	Eq. (70)
$\tilde{\mathcal{D}}_\rho$	domain where ρ has positive slope	Eq. (76)
δ	function that defines the change of angular momentum σ_1 in the restricted impact map	Eq. (65)
δ_e	same as δ , but expressed in terms of energy instead of momentum	Eq. (67)
Δ	generalized impact map	Thm. 1
Δ_s^f, Δ_f^s	impact maps determining new initial conditions at transitions	Sec. IV
Δ_{zero}	restricted impact map	Eqs. (63), (64)
$\tilde{\Delta}$	change in velocity at impact	Sec. III-D
$\tilde{\Delta}_f^s$	Δ_f^s extended to include the parameters	Sec. VI-C

Symbol	Meaning	Defined
$f_{cl,f}, f_{cl,s}$	closed-loop system	Secs. VI-A, VI-B
\mathbf{f}_i	vector function defining the Cartesian coordinates of the mass center with respect to the Cartesian coordinates of leg i	Eqs. (8), (16)
f_f, f_s	drift vector fields in flight and stance phase models	Eqs. (7), (15)
$f_{zero}(z)$	stance-phase zero dynamics	Sec. VI-A
fcn	conveniently parameterized function used in the flight-phase controller, $h_{d,f}(\mathbf{x}_{cm}, a_f) = \text{fcn}(a_1, \dots, a_5)(\tau)$	Sec. VIII-C
F_n, F_t	normal and tangential components of the ground reaction force	Sec. IX-A
$\mathcal{F}_f, \mathcal{F}_s$	open-loop flow on state manifold	Sec. IV
$\tilde{\mathcal{F}}_{cl,f}, \tilde{\mathcal{F}}_{cl,s}$	closed-loop flow on state manifold	Sec. VI-C
g	gravity constant	
g_f, g_s	control vector fields in flight and stance phase models	Eqs. (7), (15)
G_f, G_s	gravity vector	Sec. III-B
Γ_f, Γ_s	vector of generalized forces and torques	Secs. III-B, III-C
h_f, h_s	output functions in flight and stance phases	Secs. VI-A, VI-B
$h_{d,f}, h_{d,s}$	part output function defining a virtual constraint	Sec. VI-A, VI-B
H_f^s, H_s^f	switching condition from “subscript” to “superscript”	Sec. IV
\mathbf{I}	pseudo inertia, defines the zero dynamics	Eq. (40)
I_R	intensity of the ground reaction impulses at impact	Sec. III-D
k_1, k_2	gains of the stride-to-stride controller	Sec. X
K_d, K_p	derivative and proportional gains of the flight-phase controller	Sec. VI-B
K_f, K_s	total kinetic energy	Secs. III-B, III-C
$K_{s,zero}$	kinetic energy for the zero dynamics in stance phase	Eq. (44)
$\kappa_{1,f}$	part of zero dynamics in flight phase	Eqs. (49), (50)
leg-1	supporting leg in stance	Fig. 1
leg-2	swing leg in stance, leg that impacts at end of flight	Fig. 1
$\mathcal{L}_f, \mathcal{L}_s$	Lagrangian	Secs. III-B, III-C
$\mathcal{L}_{s,zero}$	Lagrangian for the stance phase zero dynamics	Sec. VI-A
λ_x, λ_y	normalized velocity of the center of mass	Eqs. (45), (46)
m	total mass of the biped	Sec. III-B
μ	slope of the restricted Poincaré return map at its fixed point	Eq. (75)
μ_d	desired value of μ	Sec. X
\mathbb{N}	natural numbers	Sec. VI-B
\mathcal{O}	periodic orbit viewed as a subset of the state space	Sec. VIII
P	Poincaré map	Eq. (61)
P_f, P_s	generalized Poincaré map for a single phase	Sec. VII-A

Symbol	Meaning	Defined
π	canonical projection that picks off the configuration variable	Sec. VII-B
$\varphi_{cl,f}(t, x_0), \varphi_{cl,s}(t, x_0)$	integral curve of closed-loop dynamics with initial state x_0	Sec. VII-A
Φ_f, Φ_s	coordinate change on configuration space	Secs. VI-A, VI-B
q_b	body coordinates, $q_b = (q_1, q_2, q_3, q_4)'$	Fig. 1
q_f, q	generalized coordinates in the flight $q_f := (q'_b, q_5, x_{cm}, y_{cm})'$ and stance $q := (q'_b, q_5)'$ phases	Sec. III-B
q_0^{f-}	reference configuration at the end of flight; equals $R^{-1}q_0^{s+}$	Eq. (83)
q_0^{s-}	reference configuration at the end of stance	Eq. (41)
q_0^{s+}	reference configuration at the beginning of stance	Secs. VII-A, VIII-A
q_5	angle of torso referenced to world frame, and hence biped's absolute orientation	Fig. 1
\dot{q}_0^{s-}	normalized reference velocity at the end of stance	Eq. (42)
\dot{q}_0^{s+}	normalized reference velocity at the beginning of stance	Sec. VIII-A
\dot{q}_0	function determining the velocity before impact as function of velocity after impact	Eq. (83)
$\mathcal{Q}_f, \mathcal{Q}_s$	configuration space	Secs. III-B, III-C
R	relabeling matrix to take into account the change of leg number	Eq. (31)
ρ	restricted Poincaré return map	Eq. (68)
\mathbb{S}	topological circle	
\mathcal{S}	switching hyper-surface, also called an impact hyper-surface, $\mathcal{S} := \mathcal{S}_s^f$	Thm. 1
$\mathcal{S}_s^f, \mathcal{S}_f^s$	switching hyper-surface	Sec. IV
$\tilde{\mathcal{S}}_f^s, \tilde{\mathcal{S}}_s^f, \tilde{\mathcal{S}}$	domains where associated Poincaré maps are well defined	Sec. VII-A, VII-B
$\tilde{\mathcal{S}}_f^s$	switching hyper-surface in the extended closed-loop model that includes the parameters as states	Sec. VI-C
σ_{cm}	angular momentum of the biped about its center of mass	Sec. III-E
σ_i	angular momentum of the biped about the end of leg- i	Sec. III-E
σ_1^{s-}	value of σ_1^{s-} on a periodic orbit	Sec. VIII-C
Σ_f, Σ_s	flight and stance portions of open-loop hybrid model	Eq. (29)
$\Sigma_{f,cl}, \Sigma_{s,cl}$	flight and stance portions of closed-loop hybrid model	Eq. (56)
t_f	duration of flight	Eq. (100)
$T_{I,f}, T_{I,s}$	time-to-impact function for one phase	Eq. (57)
$T\mathcal{Q}_f, T\mathcal{Q}_s$	state space, tangent spaces of \mathcal{Q}_f and \mathcal{Q}_s	Secs. III-B, III-C
\mathbf{T}	coordinate transformation matrix	App. I
$\mathcal{T}_s^f, \mathcal{T}_f^s$	transition function from "subscript" to "superscript"	Sec. IV
$\bar{\mathcal{T}}_f^s, \bar{\mathcal{T}}_s^f$	transition function from "subscript" to "superscript"	Sec. VI-C

Symbol	Meaning	Defined
τ	scaled time for flight phase, varies from 0 to 1	Eq. (84)
θ_s	in stance phase, angle of the hips with respect the stance leg end	Fig. 1
θ_s^-	value of θ_s just before transition to flight phase	Sec. IV
θ_s^+	value of θ just after transition from flight phase	Eq. (65)
$\theta_{s,0}^-$	desired value of θ_s at the transition from stance to flight; a constant	Sec. IV
$\theta_{s,0}^+$	desired value of θ at the beginning of stance; a constant	Eq. (65)
u	vector of actuator torques	Sec. III-B
u_s^*	feedback control that renders stance-phase zero dynamic manifold invariant	Eq. (38)
v	control after partial feedback linearization	Eq. (35)
\mathbf{v}	relative velocity of the end of the leg with respect to the ground	Sec. IX-A
V_f, V_s	total potential energy	Secs. III-B, III-C
$V_{s,zero}$	potential energy for the stance phase zero dynamics	Eq. (43)
$V_{s,zero}^{max}$	maximal value of the potential energy for the zero dynamics during stance phase	Eq. (71)
w_s^f	transition function for the flight-phase parameters a_f	Sec. VI-C
w_1, w_2	functions defining the landing configuration	Sec. X
x_f, x_s	state variables for flight and stance, $(q'_f, \dot{q}'_f)'$ and $(q', \dot{q}')'$	Secs. III-B, III-C
x_f^{*-}, x_s^{*-}	state on intersection of periodic orbit and switching hyper-surface	Sec. VIII
\bar{x}_f	extended state in flight phase, includes the parameters, $\bar{x}_f := (x'_f, a'_f)'$	Sec. VI-B
x_{cm}	horizontal Cartesian coordinate of the center of mass	Fig. 1
$\mathbf{x}_1, \mathbf{x}_2$	horizontal Cartesian coordinates of the leg ends	Fig. 1
$\mathcal{X}_f, \mathcal{X}_s$	state space	Sec. IV
$\bar{\mathcal{X}}_f$	state space including the parameters	Sec. VI-C
y_f, y_s	output used to define the desired coordination of the limbs	Secs. VI-A, VI-B
y_{cm}	vertical Cartesian coordinate of the center of mass	Fig. 1
y_1, y_2	vertical Cartesian coordinates of leg end	Fig. 1
z	state variable for hybrid zero dynamics	Eq. (39)
\mathbf{z}	penetration depth of leg end in compliant ground model	Eq. (89)
$\mathcal{Z}_{f,a_f}, \mathcal{Z}_s$	zero dynamics manifold	Secs. VI-A, VI-B
ζ	kinetic energy for the stance-phase zero dynamics just before transition to flight, $\zeta := \frac{1}{2}(\sigma_1^{s-})^2$	Sec. VII-B
ζ^*	fixed point of restricted Poincaré map	Eq. (74)