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► **To cite this version:**

Hassen Fourati, Nouredine Manamanni, Lissan Afilal, Yves Handrich. Complementary Observer for Body Segments Motion Capturing by Inertial and Magnetic Sensors. *IEEE/ASME Transactions on Mechatronics*, 2014, 19 (1), pp.149-157. 10.1109/TMECH.2012.2225151 . hal-00690145v2

HAL Id: hal-00690145

<https://hal.science/hal-00690145v2>

Submitted on 1 Feb 2013

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Complementary Observer for Body Segments Motion Capturing by Inertial and Magnetic Sensors

Hassen Fourati, Nouredine Manamanni, *Member, IEEE*, Lissan Afilal, and Yves Handrich

Abstract—This paper presents a viable quaternion-based complementary observer (CO) that is designed for rigid body attitude estimation. We claim that this approach is an alternative one to overcome the limitations of the extended Kalman filter. The CO processes data from a small inertial/magnetic sensor module containing triaxial angular rate sensors, accelerometers, and magnetometers, without resorting to GPS data. The proposed algorithm incorporates a motion kinematic model and adopts a two-layer filter architecture. In the latter, the Levenberg Marquardt algorithm preprocesses acceleration and local magnetic field measurements, to produce what will be called *the system's output*. The system's output together with the angular rate measurements will become measurement signals for the CO. In this way, the overall CO design is greatly simplified. The efficiency of the CO is experimentally investigated through an industrial robot and a commercial IMU during human segment motion exercises. These results are promising for human motion applications, in particular future ambulatory monitoring.

Index Terms—Complementary observer (CO), inertial measurement unit (IMU), motion capture, quaternion, wearable MEMS sensors.

I. INTRODUCTION

MOTION tracking is the key technology in many fields such as stroke rehabilitation [1], gait analysis [2], human motion monitoring [3], and other applications that require continuous information about body attitude. The attitude acts as a tool for biologists to enable them to estimate the power transfer between the human body and the environment [4] in many applications such as stride analysis, physical labor, and rehabilitation [5]. A number of motion tracking systems have already been developed for human motion including mechanical, magnetic, optical, acoustic, and inertial/magnetic tracking systems [1].

Manuscript received March 14, 2012; revised June 12, 2012 and September 7, 2012; accepted October 7, 2012. Recommended by Technical Editor W.-J. Kim. This work was supported by the Alsace and Champagne-Ardenne Regions within the framework of the project "NaviMeles."

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Digital Object Identifier 10.1109/TMECH.2012.2225151

Mechanical tracking systems use an exoskeleton that is attached to the articulated structure intended to be tracked [6]. Goniometers within the skeletal linkages measure joint angles. These systems only track one single rigid body. Optical tracking systems using markers and vision are recently reported in the literature. Image-based systems determine position using multiple cameras to track predesignated points on moving objects within a working volume [7]. However, the use of markers is limited to small areas and the approach using these tools needs specific material (markers and cameras), which is relatively expensive or not user friendly.

Body tracking using inertial and magnetic sensors has recently attracted a strong interest. Inertial and magnetic tracking systems use a combination of inertial sensors (a set of accelerometers and gyroscopes) and magnetic sensors (magnetometer) [8]. There is no inherent latency associated with this sensing technology and all delays are due to data transmission and processing. Another benefit is its lack of necessary source, whereas electromagnetic, acoustic, and optic devices require emissions from a source to track objects. Due to the recent technological advances, inertial and magnetic sensors have become available with low-cost, small-size, and low-energy consumption [9]. This helped to build wrist-watch-sized, self-contained inertial/magnetic sensor modules. Consequently, human motion estimation can be tracked outdoor using smaller ambulatory measurement systems.

Inertial and magnetic sensors have varying advantages and drawbacks. Accelerometers theoretically measure the sum of linear acceleration and gravity [9]. In quasi-static situations, the linear acceleration can be neglected with respect to the gravity [10]. Measuring the gravity in the sensor coordinate frame using accelerometers allows the estimation of orientation relative to the horizontal plane. However, in a dynamic situation (free motion) the accelerometer measures linear acceleration and gravity. In this case, it is not easy to dissociate these two physical quantities, and thus, it becomes difficult to calculate the attitude accurately. Gyroscopes measure angular velocity and estimate a change in orientation. However, this solution would be prone to drifting over time due to the build-up of bias and drift errors [11]. Magnetometers are used to measure the local magnetic field vector in sensor coordinates and allow the determination of orientation relative to the vertical axis, which provides additional information regarding orientation [12]. The main problem with magnetometers is the influence of ferrous material. An in-depth analysis in [13] describes some experiments carried out to examine and characterize small-scale magnetic interference caused by typical objects and how this interference affects the accuracy of orientation estimates from

inertial/magnetic sensor modules. The only recommendation stated in [13] is to maintain a distance of approximately 1 m from the source of interference. Since each sensor shows some advantages and drawbacks, the key element of the study is how to combine these three data in order to improve the quality of the motion reconstruction.

Many studies of motion tracking using inertial and magnetic sensors have been performed using a number of fusion techniques in order to overcome the drawbacks that appear when we use each sensor individually. The orientation drift resulting from gyroscope bias [11] can be bounded by additional sensors. In [2] and [14], the authors describe an attitude estimation algorithm based on a three-axis gyroscope and three-axis accelerometer data. The proposed approaches used the cyclical nature of human gait and a Kalman filter (KF). The change in orientation calculated using gyroscopes is fused with the inclination measured by accelerometers to estimate an accurate orientation even with the presence of accelerations. In [15], an orientation tracking system is performed using nine-axis sensor modules containing three orthogonally mounted triads of angular rate sensors, accelerometers, and magnetometers. A linear KF was designed to process the sensor signals and to estimate the desired sensing variables of the gravity and magnetic field and further the orientation of the body segment. This nine-axis sensor module is also used in [16] to develop an extended Kalman filter (EKF) for real-time estimation of human limb segment orientation. This EKF presents some drawbacks such as the difficulty to guarantee the global convergence of the filter due to the linear approximation of its nonlinear process model. The work reported in [17] considered the subproblem of using integrated inertial and magnetic sensing to track the attitude. Experiments are performed to validate the developed KF only using small clinical motion and a small rotational perturbation. A wearable sensing system is developed in [18] for the tracking and monitoring of functional arm movement. A body-sensing module was designed, integrating an optical linear encoder and an accelerometer. The authors concluded that the best reconstructed human motion is still not perfect. Recently, the authors in [19] have described a method to determine the attitude of a human body based on micro inertial/magnetic sensor units. The approach employs a KF. A commercial inertial measurement unit (IMU) is used in [20] to measure the attitude, necessary for biofeedback problems.

Through the cited data fusion approaches, a multitude of attitude representation is used. Quaternion is introduced to improve computational efficiency and to avoid singularities. Due to the unconventional nature of quaternion kinematics, the filtering approaches have been designed in two different ways according to the formulation of the error vector and the update of state estimates. We can differentiate between the additive and multiplicative quaternion techniques [5], [16], [21], [22]. These two techniques are used in the calculation of error vector and in the update of state estimates. The developed EKF in [16] is based on a quaternion additive technique. This approach is easy to implement but is considered as localized approximation since it is valid for only small attitude changes. A complementary filter is developed in [5], [22] based on quaternion multiplicative

technique and can be applied for larger attitude maneuvers. It will also be used during this study.

In this paper, a complementary observer (CO) approach is developed for human segment motion estimation. In order to produce 3-D orientation estimates relative to an Earth-fixed reference frame, the CO uses input from a sensor module containing a triad of orthogonally mounted accelerometers, angular rate sensors, and magnetometers. Quaternion is used to represent the attitude. The algorithm incorporates a motion kinematic equation to model the human segment motion. It adopts a two-layer filter architecture in which the Levenberg Marquardt algorithm (LMA) preprocesses acceleration and local magnetic field measurements to produce what will be referred to as *the system's output*. The latter together with the angular rate data becomes measurements for the CO. In this way, the overall CO design is greatly simplified.

The primary contributions of this paper can be stated within the following points.

- 1) The experimental validation of the CO performance for human posture tracking applications.
- 2) The development of estimation approach without the intermediate step of gyro bias estimation/correction like previous attitude estimation approaches [21], [23].
- 3) The data fusion of inertial and magnetic measurements without resorting to GPS data.
- 4) The use of robust observer instead of EKF. EKF presents some drawbacks such as the difficulty in guaranteeing the global convergence of the filter due to the linear approximation of the nonlinear process model [24].
- 5) The development of an observer based on a quaternion multiplicative technique, rarely used in the literature although it is more convenient for the nature of quaternion algebra.

This paper is organized as follows. Section II details the materials used and methods: the problem position, the physical system, and the proposed CO for attitude estimation. Section III explains the experimental methodology deployed on a robot and a human subject. Section IV is devoted to the results interpretation and discussion. The final section provides some conclusions and future work.

II. MATERIALS AND METHODS

A. Physical System and Process Model

As stated previously, the objective of this paper is to design a CO for body motion estimation. Therefore, it is necessary to establish a process model representing body motion dynamics. We adopted the attitude kinematic differential equation (1) used in [25] to describe the motion of a rigid body

$$\dot{q}_r = \frac{1}{2} q_r \otimes \bar{\omega}_g \quad (1)$$

where $q_r = [q_{r,0} \quad q_{r,\text{vect}}^T]^T$ is the unit quaternion that denotes the mathematical representation of a rigid body attitude between two frames: 1) body-fixed frame $B (X_B, Y_B, Z_B)$ and 2) earth-fixed frame $N (X_N, Y_N, Z_N)$. The X_N -axis points true north. The Z_N -axis points toward the interior of the Earth,

perpendicular to the reference ellipsoid. The Y_N -axis completes the right-handed coordinate system, pointing east (NED: north, east, down).

The unit quaternion q_r is expressed as

$$q_r = q_{r,0} + q_{r,\text{vect}} = q_{r,0} + q_{r,1}i + q_{r,2}j + q_{r,3}k \quad (2)$$

where $q_{r,\text{vect}} = q_{r,1}i + q_{r,2}j + q_{r,3}k = [q_{r,1} \ q_{r,2} \ q_{r,3}]^T$ represents the imaginary vector and $q_{r,0}$ is the scalar element.

The quaternion product of $q_{ra} = [q_{ra0} \ q_{ra,\text{vect}}^T]^T$ and $q_{rb} = [q_{rb0} \ q_{rb,\text{vect}}^T]^T$ is defined as

$$q_{ra} \otimes q_{rb} = \begin{bmatrix} q_{ra0} & -q_{ra,\text{vect}}^T \\ q_{ra,\text{vect}} & I_{3 \times 3} q_{ra0} + [q_{ra,\text{vect}}^\times] \end{bmatrix} \begin{bmatrix} q_{rb0} \\ q_{rb,\text{vect}} \end{bmatrix} \quad (3)$$

where $I_{3 \times 3}$ is the identity matrix and $[q_{ra,\text{vect}}^\times]$ represents the skew-symmetric matrix defined as

$$[q_{ra,\text{vect}}^\times] = \begin{bmatrix} q_{ra,1} \\ q_{ra,2} \\ q_{ra,3} \end{bmatrix}^\times = \begin{bmatrix} 0 & -q_{ra,3} & q_{ra,2} \\ q_{ra,3} & 0 & -q_{ra,1} \\ -q_{ra,2} & q_{ra,1} & 0 \end{bmatrix}. \quad (4)$$

More details about quaternion can be founded in [26].

$\bar{\omega}_g = [0 \ \omega_g^T]^T$ is the quaternion representation of the angular velocity $\omega_g = [\omega_{gx} \ \omega_{gy} \ \omega_{gz}]^T$ expressed in B [27], [28]. ω_g is measured by a three-axis gyroscope and can be often corrupted with noises and bias $\Lambda(t)$ [24], [29]. The operator \otimes is defined in (3).

Equation (1) describes the time rate of attitude variation as a result of rigid body angular rates measured by the gyroscope. Including $\Lambda(t)$, (1) can be written as

$$\dot{q}_n = \frac{1}{2} \begin{bmatrix} -q_{n,\text{vect}}^T \\ I_{3 \times 3} q_{n,0} + [q_{n,\text{vect}}^\times] \end{bmatrix} (\omega_g + \Lambda(t)) \quad (5)$$

where $[q_{n,\text{vect}}^\times]$ is defined in (3) and $I_{3 \times 3}$ is the identity matrix of dimension 3.

The process model (Σ) is composed of (5) and the output y

$$(\Sigma) : \begin{cases} \dot{q}_n = \frac{1}{2} \begin{bmatrix} -q_{n,\text{vect}}^T \\ I_{3 \times 3} q_{n,0} + [q_{n,\text{vect}}^\times] \end{bmatrix} (\omega_g + \Lambda(t)) \\ y = q_n \end{cases} \quad (6)$$

where q_n is the mathematical representation of a rigid body real attitude expressed between two frames: 1) body-fixed frame B and 2) Earth-fixed frame N . It represents the attitude variation derived from the rigid body angular rates integration. Acceleration and local magnetic field measurements are used to produce what is called the *system's output* $q_m \in \mathbb{R}^{4 \times 1}$ based on the LMA.

B. Inertial/Magnetic Sensors for Motion Estimation: CO

CO design is described in Section II-A. The state vector is composed of four elements of quaternion. Fig. 1 shows the

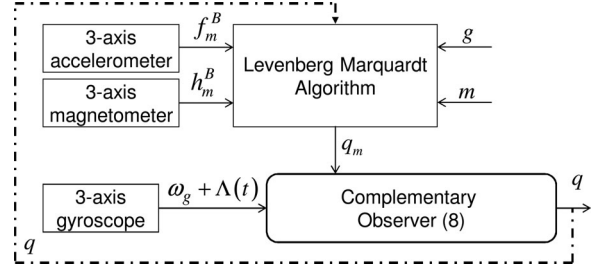


Fig. 1. Scheme of the estimation algorithm.

block diagram of the proposed approach. Acceleration and local magnetic field measurements are used as inputs to the LMA (see Section II-C) to produce *the system's output*. The system's output together with the angular rate measurements becomes inputs for the CO. Therefore, we can take advantages of the precision near to zero ($< 1^\circ$), given by the rate gyros integration in the short term ($\approx 1s$) and the reliable long-term accuracy provided by accelerometer / magnetometer [21].

The emphasis of the proposed observer is based on the multiplicative correction technique [26] written as follows:

$$q(k+1) = \delta q(k+1) \otimes q(k) \quad (7)$$

where $q(k+1)$ and $q(k)$ represent the post- and preupdate values of the quaternion estimates, respectively. Quaternion multiplication is used in (7) to correct and update the quaternion calculation. $\delta q(k+1)$ is the correction term which is a function of the error quaternion. This technique is more convenient for the transition between two quaternions in the case of large attitude motions. Finally, the CO can be designed from (5) and (7) as (8), shown at the bottom of the page, where q is the estimated quaternion by the three sensors and $[q_{\text{vect}}^\times]$ is defined in (4). δ_{K_1} represents the switching correction term and δ_{K_2} is the linear correction term. In order to perform the quaternion multiplication, each correction term should be converted into a quaternion. This conversion is obtained using the forced normalization method [30]. δ_{K_1} and δ_{K_2} are computed using

$$\delta_{K_1} = \frac{1}{\|\chi_1\|} \chi_1; \quad \delta_{K_2} = \frac{1}{\|\chi_2\|} \chi_2 \quad (9)$$

where

$$\chi_1 = K_1 \begin{bmatrix} 1 & \text{sat} \left[\frac{q_{e,\text{vect}}}{\rho} \right] \end{bmatrix}^T \quad \chi_2 = K_2 [1 \ q_{e,\text{vect}}]^T \quad (10)$$

with

$$K_1 = \begin{bmatrix} 1 & 0 & 0 & 0 \\ 0 & k_1 & 0 & 0 \\ 0 & 0 & k_2 & 0 \\ 0 & 0 & 0 & k_3 \end{bmatrix} \quad K_2 = \begin{bmatrix} 1 & 0 & 0 & 0 \\ 0 & k_4 & 0 & 0 \\ 0 & 0 & k_5 & 0 \\ 0 & 0 & 0 & k_6 \end{bmatrix}. \quad (11)$$

$$\begin{cases} \dot{q} = \begin{bmatrix} \dot{q}_0 \\ \dot{q}_1 \\ \dot{q}_2 \\ \dot{q}_3 \end{bmatrix} = \delta_{K_1} \otimes \delta_{K_2} \otimes \left(\frac{1}{2} \begin{bmatrix} -q_{\text{vect}}^T \\ I_{3 \times 3} q_0 + [q_{\text{vect}}^\times] \end{bmatrix} \left(\begin{bmatrix} \omega_{gx} \\ \omega_{gy} \\ \omega_{gz} \end{bmatrix} + \Lambda(t) \right) \right) \end{cases} \quad (8)$$

The term $q_{e,\text{vect}}$ in (10) represents the imaginary part of the error quaternion q_e . q_e measures the difference between the complementary estimated quaternion $\bar{q} = [q_0 \ -q_1 \ -q_2 \ -q_3]^T$ and the system's output q_m . q_e is expressed by

$$q_e = \bar{q} \otimes q_m = [q_{e0} \ q_{e,\text{vect}}]^T. \quad (12)$$

The scalar parts of χ_1 and χ_2 are chosen equal to unity since the incremental quaternion corresponds to a small rotation angle [26]. The saturation function in (10) is used to avoid the high-frequency chattering [31]. It can be written according to

$$\text{sat}[q_{e,\text{vect}}/\rho] = \begin{cases} 1, & q_{e,\text{vect}} \leq \rho \\ q_{e,\text{vect}}/\rho, & |q_{e,\text{vect}}| \leq |\rho| \\ -1, & q_{e,\text{vect}} \geq \rho. \end{cases} \quad (13)$$

The parameter ρ is the surface boundary layer. It determines the behavior in the vicinity of $q_{e,\text{vect}} = 0$. To preserve the unit quaternion norm, the estimated quaternion q in (8) can be normalized to avoid the divergence [30].

C. System's Output q_m Calculation: LMA

The LMA is used for a single-frame estimation of the system's output q_m . The algorithm was proposed to solve the Wahba problem [32] that involved the determination of rigid body attitude in reference to a fixed coordinate system based on a set of observation vectors known in a fixed frame and measured in a mobile frame. A minimum of two vectors are required to compute the optimal attitude represented by q_m .

In the present case, the three-axis magnetometer is a sensor that provides the Earth's magnetic field h in the body frame B . The theoretical output of a magnetometer can be written using the estimated quaternion such as [12]

$$h = [h_x \ h_y \ h_z]^T = M_N^B(q) m + \delta_h \quad (14)$$

where $M_N^B(q)$ is the rotation matrix expressed as

$$M_N^B(q) = \begin{bmatrix} 2q_0^2 + 2q_1^2 - 1 & 2(q_1q_2 + q_0q_3) & 2(q_1q_3 - q_0q_2) \\ 2(q_1q_2 - q_0q_3) & 2q_0^2 + 2q_2^2 - 1 & 2(q_0q_1 + q_2q_3) \\ 2(q_0q_2 + q_1q_3) & 2(q_2q_3 - q_0q_1) & 2q_0^2 + 2q_3^2 - 1 \end{bmatrix}. \quad (15)$$

$\delta_h \in \mathbb{R}^3$ is a vector of white noise and m represents the magnetic field vector measured in the Earth frame N as

$$m = [m_x \ 0 \ m_z]^T = [\|m\| \cos(\theta) \ 0 \ \|m\| \sin(\theta)]^T. \quad (16)$$

The theoretical model of the magnetic field nearest to reality considers this vector with an inclination angle of $\theta = 60^\circ$ and a vector norm of $\|m\| = 0.5$ Gauss [33]. The magnetic field is locally constant in the fixed frame N and can be represented by the vector m , which denotes *the first observation vector*.

The three-axis accelerometer measures the specific force f in the body frame B as follows [34]:

$$f = [f_x \ f_y \ f_z]^T = M_N^B(q) [a + g] + \delta_f \quad (17)$$

where $g = [0 \ 0 \ 9.81]^T$ is the gravity vector and $a = [a_x \ a_y \ a_z]$ is the linear acceleration of the body, expressed in the Earth frame N [35], [36]. The rotation matrix $M_N^B(q)$ is expressed in (15) and $\delta_f \in \mathbb{R}^3$ is a vector of white noise. In a quasi-static situation, the gravitational field g dominates the accelerometer measurements f ($\|a\|_2 \ll \|g\|_2$) [21]. In this case, the quantity g is also constant in the Earth frame N and could provide *the second observation vector*.

The LMA is an estimator that uses the Earth's magnetic field m and the gravity vector g as the observation vectors and the real measurements f_m^B and h_m^B to deduce the system's output q_m . It is based on the following steps.

- 1) Measure the readings f_m^B and h_m^B from the accelerometer and magnetometer, respectively.
- 2) Calculate $[0 \ f_m^N]^T = q \otimes [0 \ f_m^B]^T \otimes \bar{q}$ and do the same for $[0 \ h_m^N]^T$. f_m^N and h_m^N represent the estimated acceleration and magnetic field vectors in the Earth frame N .
- 3) Calculate the navigation errors $f_e = g - f_m^N$ and $h_e = m - h_m^N$ in order to form $z = [h_e \ f_e]^T$.
- 4) Calculate the Jacobian matrix $J = -2 \begin{bmatrix} [[h_m^N]^\times]^T & [[f_m^N]^\times]^T \end{bmatrix}$.
- 5) Calculate the pseudoinverse $O^* = [J^T J + \lambda I_{3 \times 3}]^{-1} J^T$. The constant λ is chosen small (between 10^{-1} and 10^{-2}) to ensure the nonsingularity of the minimization problem [37].
- 6) Calculate the quaternion error such as $q_{er} = \alpha O^* z$. α is a smooth parameter chosen between 0 and 1 [32].
- 7) Calculate q_m such as $q_m = q \otimes [1 \ q_{er}]^T$. q is estimated at each integration step by the CO.

D. Performance Analysis of the Designed Observer

A frequency analysis of inertial and magnetic sensor data shows that they have a complementary frequency spectrum [24]. The resulting structure of the proposed CO blends two frequency regions and is based on the complementary filtering theory [38].

- 1) The accelerometer and magnetometer data are characterized by a low-frequency region, where the attitude is usually more accurate.
- 2) The gyroscope data are located in the high-frequency region, where the integration of the angular velocity yields better attitude estimates.

To study the stability and performance of the CO, we established the block diagram of the linearized quaternion observer (see Fig. 2). This block diagram results from the (6), (8)–(13), and the system's output q_m issued from the LMA. We applied the Laplace transform (s is the Laplace operator) to q_m , \dot{q}_n , and q . The Laplace transformation (LT) of the quaternion q_m is $q_m(s)$ and $sq_{ng}(s)$ is the LT of \dot{q}_{ng} . Here, we suppose that $\dot{q}_{ng} = \dot{q}_n$ where q_{ng} is the quaternion obtained by integrating the three-axis gyroscope measurements using (6). In Fig. 2, the linearization led to approximate $\bar{q} \otimes q_m$ by $q_m - q$. More details on the linearization can be founded in [39].

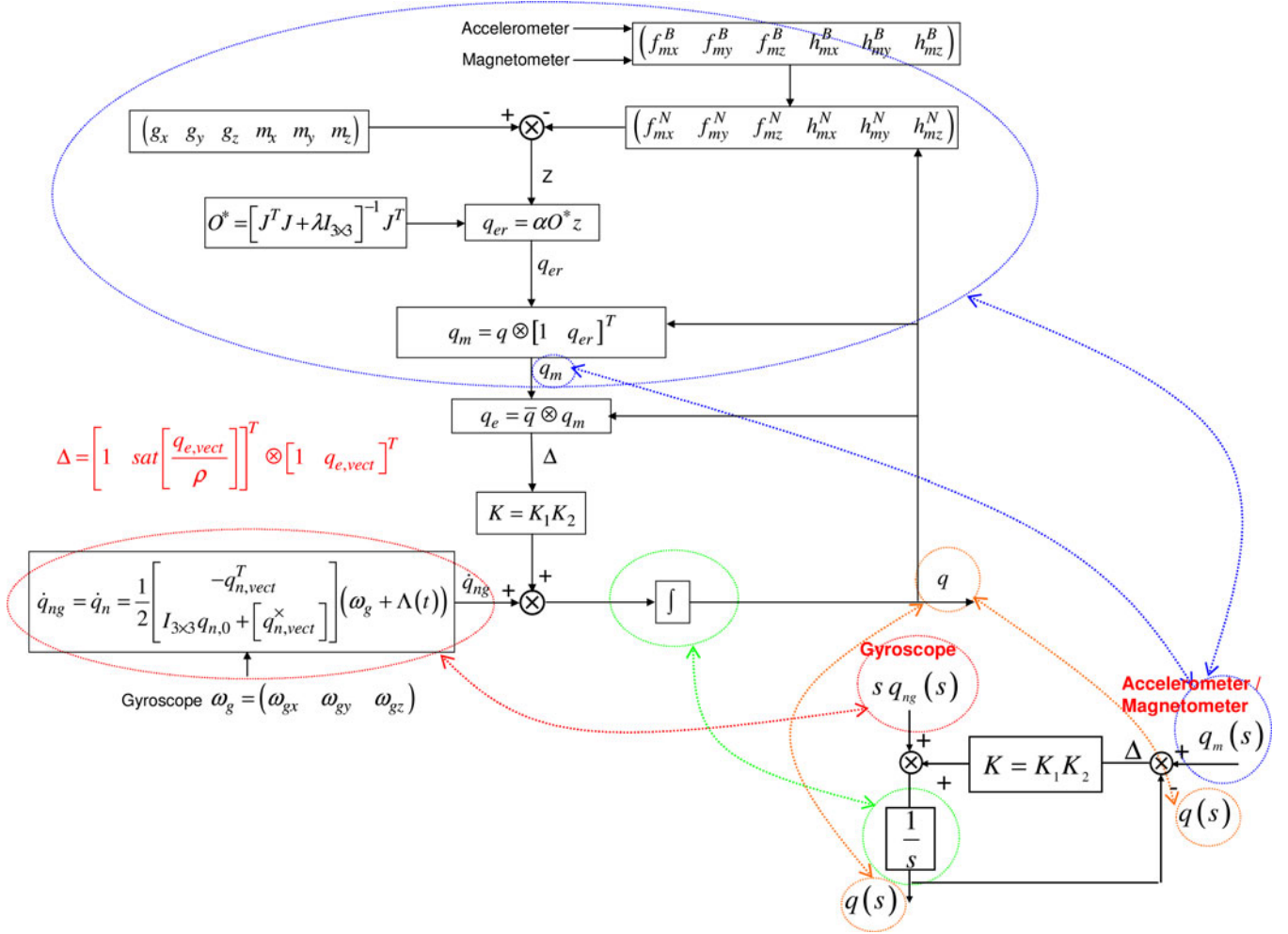


Fig. 2. Block diagram of the transform domain (Laplace) of the linearized quaternion observer.

From Fig. 2, the transfer function $F_1(s)$ based on the accelerometer and magnetometer inputs is given by

$$F_1(s) = \frac{q(s)}{q_m(s)} = \frac{Ks^{-1}}{1 + Ks^{-1}} = \frac{K}{s + K}. \quad (18)$$

Equation (18) has the form of a first-order low-pass filter. Therefore, the system's output $q_m(s)$ is low-pass filtered with respect to $q(s)$. So, the perturbation effects due to the high-frequency components (resulting from the linear acceleration a) of accelerometer signals are filtered from $q_m(s)$. The gain K can be written as $K = K_1 K_2$, where K_1 and K_2 are given in (11). Similarly, from Fig. 2, the transfer function based on the gyroscope inputs can be written as

$$F_2(s) = \frac{q(s)}{q_{ng}(s)} = \frac{1}{1 + Ks^{-1}} = \frac{s}{s + K}. \quad (19)$$

Equation (19) has the form of a first-order high-pass filter. The gyroscope measurements are high-pass filtered with respect to the output $q(s)$. Thus, the perturbations due to low-frequency components of gyroscope signal (the noises and biases) are filtered from $q_g(s)$.

The general requirement to prove the stability in the complementary filtering theory is that one of the transfer functions

complements the sum of the others [22]. Then, this requirement can be verified for the CO by the following [22]:

$$F_1(s) + F_2(s) = \frac{q(s)}{q_m(s)} + \frac{q(s)}{q_{ng}(s)} = \frac{K}{s + K} + \frac{s}{s + K} = 1. \quad (20)$$

The global transfer function of the CO is

$$q(s) = \left(\frac{K}{s + K} \right) q_m(s) + \left(\frac{s}{s + K} \right) q_{ng}(s). \quad (21)$$

III. EXPERIMENTAL METHODOLOGY

A. Experimental Tools for the Attitude Estimation: MTi and MTi-G

In order to evaluate the efficiency of the proposed CO under several conditions in real applications, a set of experimental data was collected using two IMUs: the *MTi* and the *MTi-G*. These sensor modules are designed by Xsens Technologies [40]. Sensing components for these units include a three-axis accelerometer, a three-axis magnetometer, and a three-axis gyroscope providing calibrated digital outputs at a rate of 100 Hz (3-D acceleration, 3-D angular rate, and 3-D local magnetic field). The *MTi-G* is a GPS-enhanced attitude and heading reference

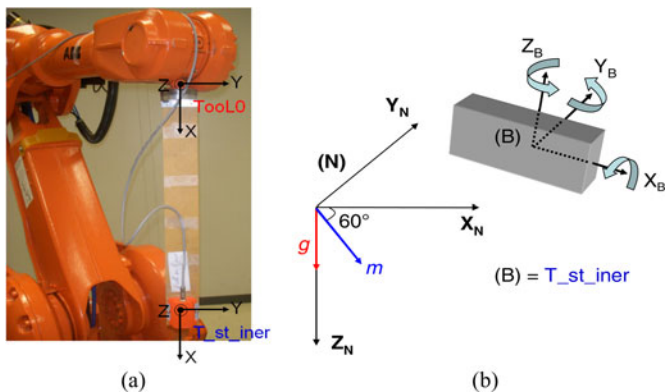


Fig. 3. (a) Experimental setup: the MTi is mounted on the robot for orientation tracking. (b) Coordinate system (B) of a rigid body represented in the Earth-fixed frame (N).

system (AHRS). The calibration procedure of the sensors is performed by the manufacturer. The *MTi* and the *MTi-G* measure the attitude based on an embedded EKF [40]. The EKF combines measurements from the inertial/magnetic sensors and a GPS module (the GPS is only for the *MTi-G*) to estimate the attitude in quaternion or rotation matrix representations. In the following experiments, the calibrated data from the *MTi* and the *MTi-G* are used as inputs to the CO.

B. Experimental Setup

The experimental setup used to evaluate the algorithm for the attitude estimation consists of three tests: 1) robot mounted tests; 2) improvement of the *MTi* shortcomings; and 3) human segments motion evaluation. The MATLAB computing program was used for all post-trial data processing and analysis.

1) *Robot Mounted Tests*: The experiments were carried out in the robotic laboratory of “PSA Peugeot Citroën” company based in Metz, France. The experimental setup used to evaluate the CO consists of an industrial robot IRB 2400 from the ABB Group [41] and the *MTi*. The robot is considered as the orientation reference since it offers an excellent motion control of around six axes and gives a high performance in the material handling with a position repeatability of 0.06 mm and 0.1° . The ranges of velocity rate used to move the robot are $[0-1 \text{ m/s}]$ and $[0-20^\circ/\text{s}]$.

Fig. 3(a) shows the complete setup. In order to mitigate possible magnetic effects generated by the steel construction of the robot, the *MTi* is mounted on a nonferrous extension below the terminal effector of the robot. The extension is made of a piece of Plexiglas board and is approximately 50 cm in length. Two aligned frames are considered in this experimental setup: *T_st_iner* and *ToolO* correspond to the *MTi* and the terminal effector of robot, respectively [see Fig. 3(b)].

The reference paths for the six degrees of freedom of the IRB robot are generated with a programmed model (a trajectory like a straight line). The simulated motion is recorded and used afterward to move the IRB robot. The measured *MTi* signals are used for the estimation of the *MTi*'s attitude with the proposed CO. The estimated quaternion is compared to the one obtained

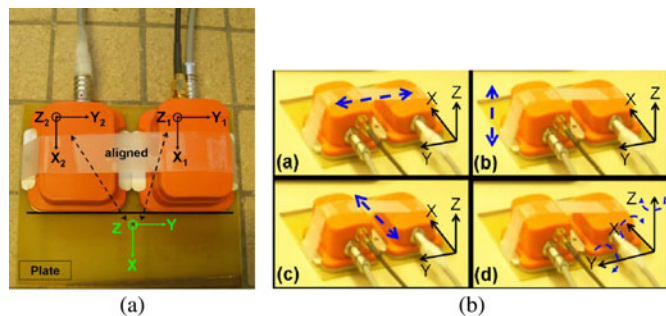


Fig. 4. (a) Whole system. The frame axes of a piece of Plexiglas board, *MTi*, and *MTi-G* are aligned (b) Motions performed during the evaluation test; the dashed arrow describes the direction of the motion.

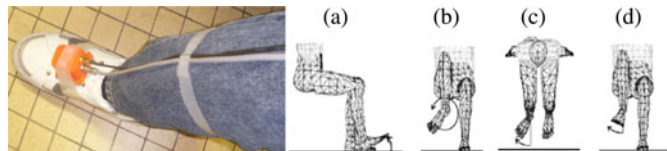


Fig. 5. Exercise 1 performed with the foot segment. (a) Toe rise foot. (b) Clockwise ankle rotation. (c) Lateral foot rotation. (d) Eversion.

from the incremental encoders of the IRB robot denoted as reference signals in the following. This motion is repeated four times by the robot to investigate the accuracy of the proposed filtering approach. During the experiment, we have chosen to increase the robot velocity at each test.

2) *Improvement of the MTi Shortcomings*: In this test, the goal is to elaborate an overall view of the attitude estimation using three methods: 1) the proposed CO; 2) the EKF in the *MTi*; and 3) the EKF in the *MTi-G*. The test procedure consists of attaching the *MTi* and the *MTi-G* to a piece of Plexiglas board using an adhesive strapping. Their frame axes are aligned as shown in Fig. 4(a). The subject took the overall package in his hand and is asked to perform the four motions outlined in Fig. 4(b) as follows: 1) move along the y -axis of the Plexiglas board in the level plane; 2) move along the z -axis of the Plexiglas board; 3) move along the x -axis of the Plexiglas board in the level plane; and 4) move around the x -axis, y -axis, and z -axis of the Plexiglas board (free motion).

3) *Human Segments Motion Evaluation*: Numerous experiments were conducted on a human subject to qualitatively evaluate the capability of the CO in the case of human body tracking. In each experiment, the *MTi-G* is attached to the human limb to be tracked using an elastic strapping. In most cases, this method appeared to keep the sensors fixed relative to the limb. The *MTi-G*'s signals are used for an estimation of the limb's attitude with the proposed CO. The estimated quaternion is compared to the one obtained from the *MTi-G*'s EKF. To validate the effectiveness of the CO, the experiments were chosen to cover a wide majority of the 3-D human motion. The subject was asked to perform four exercises on the following human segments: 1) the foot segment (see Fig. 5); 2) the lower leg segment (see Fig. 6); 3) the upper arm (see Fig. 7); and 4) the head (see Fig. 8).

In the first exercise, the subject performed the following four tasks with the foot segment for 80 s: 1) a toe rise foot; 2)

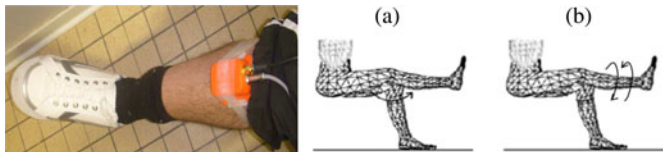


Fig. 6. Exercise 2 performed with the lower leg segment. (a) Knee extension. (b) Clockwise and anticlockwise rotation.

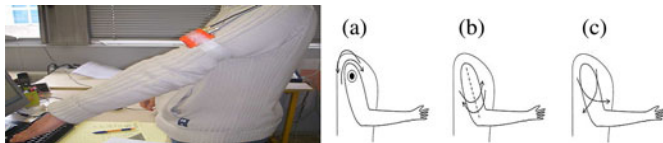


Fig. 7. Exercise 3 performed with the upper arm. (a) Clockwise and anticlockwise shoulder rotation. (b) Clockwise and anticlockwise rotation around the line defined along the upper arm segment. (c) Random motion.

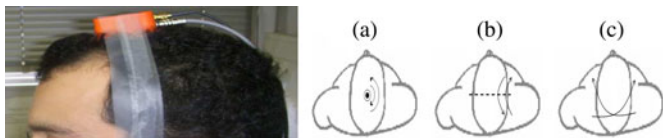


Fig. 8. Exercise 4 performed with the head. (a) Clockwise and anticlockwise neck rotation. (b) Clockwise and anticlockwise rotation around the dashed line lateral axis of the head. (c) Random motion.

a clockwise ankle rotation; 3) a lateral foot rotation; and 4) an eversion (see Fig. 5). In the second exercise, the subject carried out two tasks with the lower leg segment for 60 s: 1) an extension of the knee and 2) a rotation of the leg in clockwise and anticlockwise directions (see Fig. 6). The third exercise is done on the upper arm for 60 s. It consists of three tasks: 1) a shoulder rotation; 2) a rotation around the axis defined along the upper arm; and 3) a random motion (see Fig. 7). The exercises were completed by one performed on the head for 70 s. The subject did three tasks: 1) a rotation of the neck in clockwise and anticlockwise directions; 2) a rotation around the lateral axis of the head; and 3) a random motion (see Fig. 8). To ensure legal and personal conditions, the experiments conformed to the National Center for Scientific Research (CNRS) Standard of Ethics.

IV. RESULTS ANALYSIS AND DISCUSSION

A. Robot Tests: Results and Performance Analysis

Fig. 9(a) illustrates the evolution of the quaternion components in the case where the robot moves at 0.75 m/s. Fig. 9(a) obviously shows that the estimated quaternion by the CO is in good accordance with the reference signals calculated by the IRB robot. The observer is stable and estimates the truth attitude smoothly. Fig. 9(b) shows the quaternion estimation error between the CO and the reference (robot) after observer convergence. The obtained results illustrate that the quaternion estimation error remains in the interval that varies in ± 0.015 which proves the efficiency of the CO.

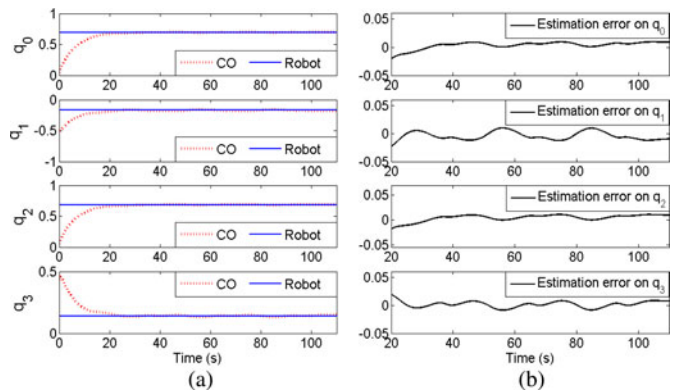


Fig. 9. (a) Comparison between quaternion components estimated by the CO and those given by the robot. (b) Quaternion estimation errors.

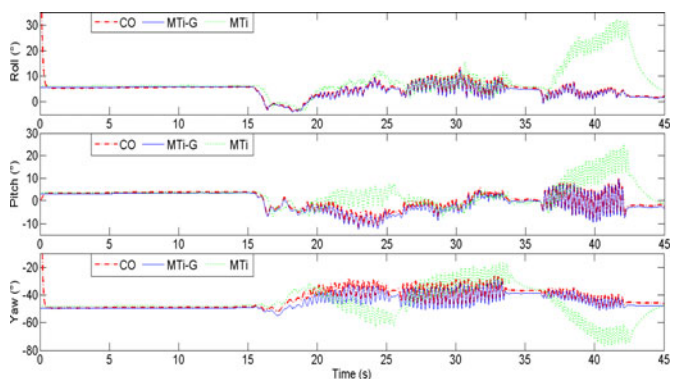


Fig. 10. Attitude estimation with the MTi, MTi-G, and CO.

B. Improvement of the MTi Shortcomings: Results and Performance Analysis

The CO is fed with inertial/magnetic data recorded from the *MTi-G* during the four motions. The estimated attitude is expressed with the Euler angle representation. Fig. 10 shows the time history evolution of the Euler angles obtained from the *MTi*, the *MTi-G*, and the CO. This figure shows the effectiveness of the estimates produced by the CO during the motion. One can notice that the estimation error of quaternion remains close to zero (between $[1^\circ, 3^\circ]$ for roll and pitch angles and between $[2^\circ, 5^\circ]$ for a yaw angle). This experiment shows the drawbacks of the *MTi*'s EKF during the intervals (17–34 s) and (37–42 s) where the estimation errors are between 10° and 20° .

C. Human Segments Motion Evaluation: Results and Performance Analysis

The estimated attitude by the CO is compared to the one obtained from the *MTi-G*'s EKF considered as reference in each exercise. The attitude is expressed also in the Euler angle representation using the mathematical transformation between the estimated quaternion and Euler angles given in [42]. Fig. 11(a) illustrates the obtained results from the exercise on the upper arm as an illustrative example. The time history evolution of the Euler angles obtained from the *MTi-G*'s EKF and the CO is shown in this figure. We obviously deduce a strong correlation between the attitudes obtained from the EKF and the CO,

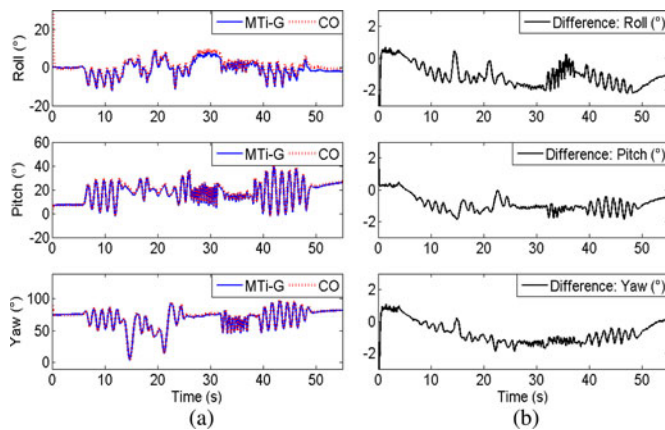


Fig. 11. (a) Euler angles estimated by the CO and those obtained by the MTi-G's EKF. (b) Estimation errors (exercise on the upper arm).

respectively. The convergence rate is fast and around 2 s. The Euler angle estimation errors are shown in Fig. 11(b), which provide an overview of the overall performance of the proposed approach in this paper. These errors are computed as the difference between the Euler angle estimates produced by the CO and the EKF, respectively. The performance consistency of the observer is illustrated in this figure since the estimation errors are around 2° for the roll and pitch angles and 3° for the yaw angle. This test shows the attitude estimation capability of the CO without resorting to the GPS data (GPS data are used only in the MTi-G's EKF). During this test, the observer was evaluated under a large range of movement and the estimation errors were very similar (between 2° and 4° for the roll and pitch angles, between 3° and 5° for the yaw angle).

V. CONCLUSION

This paper has presented the design and experimental results of a quaternion-based CO for human body motion tracking using inertial/magnetic sensor modules containing orthogonally mounted triads of accelerometers, gyroscopes, and magnetometers. The CO was designed with the goal of being able to produce highly accurate orientation estimates without resorting to GPS data. The observer design makes use of a simple kinematic motion equation to describe the system model. The CO design is further simplified by preprocessing accelerometer and magnetometer data using the LMA. The quaternion calculated by the LMA is provided as input to the CO along with the angular rate data. Some experiments were carried out on a robot and human body segments through sensor measurements provided by an IMU. The obtained results and estimation errors illustrate the performance of the proposed approach. Future work will focus on designing a low-cost, lightweight, and embedded prototype for this application.

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